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Owner's Manual

For the

Cargo Hook Swing Suspension System

with

Talon LC Hydraulic Cargo Hook

On the

Airbus Helicopters AS350 Series

System Part Numbers 200-280-02, 200-280-03

Owner's Manual Number 120-104-02
Revision 29
05/09/14

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Record of Revisions

Revision	*Date*	*Page(s)*	*Reason for Revision*
21	07/30/10	Section 1 through 4	Replaced P/N 212-014-00 with 212-014-01 and updated hydraulic fluid filling instructions to use new kit. Updated warnings, cautions and notes section to safety label section. Updated safety label format through out document.
22	08/09/10	Section 1, 2 & 4, & 6-2	Replaced warnings, cautions and notes section with safety labels sections. Updated safety label format to current format throughout document. Added fuel drain guard to kit and associated instructions. Updated weight for fixed provisions to included fuel drain guard kit.
23	03/28/11	2-8, 2-9, 6-10 & 6-12	Added Fuel Drain Warning Placard to System Parts Numbers section under swing hook/frame assembly. Added P/N 291-523-00 as an optional clamp half that is available.
24	04/20/11	1-6	Added Half Clamp Pad (EC P/N 350A-41-1099-20) to the table of Eurocopter required part numbers.
25	09/09/11	1-7, 2-5, 2-6	Updated electrical system interface information to include Eurocopter mod. # 073475.
26	12/19/11	6-14	Replaced Cup Seal (P/N 556-038-00) with Quad Ring (P/N 556-097-00) inside Slave Cylinder assembly.
27	02/19/13	1-7, 2-5, 2-6	Updated electrical schematic to reflect a/c side wiring interface of post mods 07-4280 and 07-3450. Added note regarding post-mod. 07-4280.
28	06/06/13	1-7, 1-8, 2-34, & 6-11	Replaced Bumper P/N 290-839-01 & 290-839-00 with P/N 290-839-02.
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Section 1
General Information

Introduction

The P/N 200-280-02 and 200-280-03 Cargo Hook Swing Suspension System kits consist of fixed provisions and removable provisions. The fixed provisions are permanently installed on the aircraft while the removable provisions are easily removed when not required on the helicopter's mission. The 200-280-02 kit is intended for installation on Airbus Helicopters AS350B, AS350BA, AS350D, AS350B1, and AS350B2 model helicopters.

The P/N 200-280-03 Cargo Hook Swing Suspension System kit is the same as the –02 kit except is includes a different release lever assembly and it includes a friction knob for the collective. It is intended for installation on the AS350B3 model.

Newer Cargo Hook Swing Suspension System kits (shipped after August 2010) include a fuel drain guard which is compatible with fuel tanks on AS350 B2 and B3 helicopters and on AS350B, B1, BA, and D models that have been retrofitted with the B-2 style dual fuel pump type tank. The Airbus Helicopters part number for this tank is 350A55-1015-0251. The fuel drain guard is intended for helicopters with their fuel drain levers previously modified per AD 2005-03-08. See Theory of Operation section for description of fuel drain guard.

Safety Labels

The following definitions apply to safety labels used in this manual.

Indicates a hazardous situation which, if not avoided, will result in death or serious injury.

Indicates a hazardous situation which, if not avoided, could result in death or serious injury.

Indicates a hazardous situation which, if not avoided, could result in minor or moderate injury.

Draws the reader's attention to important or unusual information not directly related to safety.

Used to address practices not related to personal injury.

Specifications

Table 1.1 Suspension System Specifications

Design load	3,086 lbs. (1400 kg.)
Design ultimate strength	11,574 lbs. (5250 kg.)
Unit weight - Fixed Provisions	5.5 lbs. (2.5 kg.)
Unit weight - Removable Provisions	30 lbs. (13.6 kg.)

Table 1.2 P/N 528-028-00 Cargo Hook Specifications

Design load	3,500 lbs. (1,587 kg.)
Design ultimate strength	13,125 lbs. (5,952 kg.)
Electrical release capacity	8,750 lbs. (3,970 kg.)
Mechanical release capacity	8,750 lbs. (3,970 kg.)
Force required for mechanical release at 3,500 lb.	12 lbs max. @ Master Cylinder
Electrical requirements	22-32 VDC 6.9 – 10 amps
Minimum release load	0 pounds
Unit weight	3.0 pounds (1.35 kg.)
Mating electrical connector	PC05A8-2S

Load capacities given are for the equipment described only. Loading limits for your particular helicopter model still apply. Consult your flight manual.

Inspection

Inspect the kit items for evidence of damage, corrosion and security of lock wire and fasteners. If damage is evident, do not use the items until they are repaired.

Bill of Materials

The following items are included with the 200-280-02 Swing Suspension System, the 210-202-02 fixed provisions kit, and the 210-201-02 removable provisions kit. If shortages are found contact the company from whom the system was purchased. See Table 1.4 for kit P/N 200-280-03.

Table 1.3 Onboard Systems Bill of Materials – 200-280-02

Part No.	Description	200-280-02	210-202-02	210-201-02
120-104-02	Owner's Manual	1	1	1
121-012-02	RFMS	1	1	1
122-015-00	Cargo Hook Service Manual	1	-	1
123-011-02	ICA	1	1	1
210-095-00	C-39 Indicator	1	1	-
210-095-04**	C-39 Indicator, NVG Lights	-	-	-
215-165-00	AS 350 Multiple Decal Sheet	1	1	-
215-166-00	Max Hook Load 3086 Decal	1	-	1
215-168-00	Max Hook Load 2557 Decal	1	-	1
232-137-01	Shackle Assembly	4	4	-
232-140-01	Fwd Attach Cable Assembly	2	-	2
232-141-01	Aft Attach Cable Assembly	2	-	2
232-145-02	Swing Frame Assembly	1	-	1
232-165-00	Master Cylinder Assembly	1	1	-
270-106-02	LW Internal Harness	1	1	-
270-108-00	Release Internal Harness	1	1	-
270-125-00	Ground Strap, Fixed	1	1	
290-772-00	Indicator Mount bracket	1	1	-
290-884-00	Connector Bracket	1	1	-
290-783-00	Relay Bracket	1	1	-
290-888-00*	Retainer	1	1	-
290-889-01*^	Guard	1	1	-
290-893-00*	Bracket	1	1	-
445-005-00	Relay	1	1	-
500-065-00	Grommet Edging	1	1	-
505-014-00	Grommet	1	1	-
510-029-00	Nut	8	8	-
510-042-00	Washer	2	2	-
510-062-00	Washer	8	8	-
510-095-00	Washer	3	3	-
510-102-00	Nut	2	2	-
510-277-00	Screw	2	2	-
510-278-00	Washer	2	2	-
510-279-00	Nut	2	2	-
510-453-00	Bolt	2	2	-
510-457-00	Screw	4	4	-
510-475-00	Screw	3	3	-
510-481-00	Screw	8	8	-
510-486-00	Rivet	3	3	-
510-526-00*	Cotter Pin	2	2	-
512-003-00	Ty-wrap	5	5	-
512-005-00	Adel Clamp	4	4	-
610-024-00*	Seal	1	1	-

** Fuel drain guard component, included with kits shipped after August 2010.*

^Fuel drain guard P/N 290-889-01 supersedes fuel drain guard P/N 290-889-00.

*** The 210-095-04 Indicator is equipped with NVG compatible lights. It is an optional indicator that can be ordered in place of the 210-095-00 Indicator.*

Bill of Materials continued

The following items are included with the 200-280-03 Swing Suspension System and the 210-202-03 fixed provisions kit. The 200-280-03 system uses the 210-201-02 removable provisions kit (see Table 1.3). If shortages are found contact the company from whom the system was purchased.

Table 1.4 Onboard Systems Bill of Materials – 200-280-03

Part No.	Description	200-280-03	210-202-03
120-104-02	Owner's Manual	1	1
121-012-02	RFMS	1	1
122-015-00	Cargo Hook Service Manual	1	-
123-011-02	ICA	1	1
210-095-00	C-39 Indicator	1	1
210-095-04**	C-39 Indicator, NVG Lights	1*	1*
210-201-02	Removable Provisions Kit	1	-
215-165-00	AS 350 Multiple Decal Sheet	1	1
215-166-00	Max Hook Load 3086 Decal	1	-
215-168-00	Max Hook Load 2557 Decal	1	-
232-137-01	Shackle Assembly	4	4
232-140-01	Forward Attach Cable Assembly	2	-
232-141-01	Aft Attach Cable Assembly	2	-
232-145-02	Swing Frame Assembly	1	-
232-165-01	Master Cylinder Assembly	1	1
270-106-02	LW Internal Harness	1	1
270-108-00	Release Internal Harness	1	1
270-125-00	Ground Strap, Fixed	1	1
290-772-00	Indicator Mount bracket	1	1
290-884-00	Connector Bracket	1	1
290-783-00	Relay Bracket	1	1
290-888-00*	Retainer	1	1
290-889-01*^	Guard	1	1
290-893-00*	Bracket	1	1
291-105-00	Friction Knob	1	1
445-005-00	Relay	1	1
500-065-00	Grommet Edging	1	1
505-014-00	Grommet	1	1
510-029-00	Nut	8	8
510-042-00	Washer	2	2
510-062-00	Washer	8	8
510-095-00	Washer	3	3
510-102-00	Nut	2	2
510-277-00	Screw	2	2
510-278-00	Washer	2	2
510-279-00	Nut	2	2
510-453-00	Bolt	2	2
510-457-00	Screw	4	4
510-475-00	Screw	3	3
510-481-00	Screw	8	8
510-486-00	Rivet	3	3
510-526-00*	Cotter Pin	2	2
512-003-00	Ty-wrap	5	5
512-005-00	Adel Clamp	4	4
610-024-00*	Seal	1	1

* Fuel drain guard component, included with kits shipped after August 2010.

^Fuel drain guard P/N 290-889-01 supersedes fuel drain guard P/N 290-889-00.

** The 210-095-04 Indicator is equipped with NVG compatible lights. It is an optional indicator that can be ordered in place of the 210-095-00 Indicator.

The following is needed for installation of the fuel drain guard and is not included with the kit and should be obtained before installation is begun.

Table 1.5 Needed Supplies

Part Number	Description	Quantity
PR1422-B	Sealant	AR

To complete the cargo hook installation the following Airbus Helicopters parts may be necessary to obtain (these parts are frequently found to be on the aircraft from the factory or are standard Airbus Helicopters parts).

These items may or may not be installed with a standard aircraft, therefore verification is recommended before purchasing them.

Table 1.6 Airbus Helicopters Part Numbers

P/N	Description	Quantity
22201BE120074L	Screw	4
23111AG120LE	Washer	8
ASN52320BH120N	Nut	4
DHS751-160.62	Grommet	1
SL211M5-1	Nut	3
A3125-2 H179	Quick Disconnect Clamp	2
350A41-1097-20	Half Clamp, Rear	2
350A-41-1099-20	Half Clamp Pad	2
350A86-0020-33	Bracket	1
ASNA0078A403	Rivet	3

The cargo hook electrical system interfaces with the aircraft's electrical panel. Earlier versions (pre-mod. #07-3274) of the AS350 utilize a fuse type switch panel. The following electrical panel components for these versions are typically found to be on the aircraft, but may be necessary to obtain.

Table 1.7 Airbus Helicopters Electrical Parts – Pre-mod. #07-3274

P/N	Description	Qty
DHS775-160-42	Indicator Light Body	1
DHS775-240-22	Indicator Light	1
EN2240-6839	Lamp	4
DI-2-5*	Fuse 2.5A	1
DA8-16A*	Fuse 16A	1

AS350B2 and B3 aircraft with modification #07-3274 incorporated utilize a circuit breaker type switch panel. The following electrical panel components for these versions are typically found to be on the aircraft, but may be necessary to obtain.

Bill of Materials continued

Table 1.8 Airbus Helicopters Electrical Parts – with Mod. #07-3274

P/N	Description	Qty
045004A127A	Cargo Hook Sling Switch	1
1180-01-2.5A (Alt. P/N: ECS0744A02A5)	Circuit Breaker 2.5A	1
1170-01-15A (Alt. P/N: ECS0744B15A0)	Circuit Breaker 15A	1

AS350B2 and B3 aircraft with modification #073475 (introduced by Airbus Helicopters Service Bulletin No. AS350-25.01.09) incorporated use a 10 amp circuit breaker installed in the 44 ALPHA panel rather than the 15 amp circuit breaker listed in Table 1.8. This 10 amp circuit breaker (P/N EN2495-10AM) may be necessary to obtain.

For AS350 B2 and B3 aircraft post-mod 07-4280 refer to the Airbus Helicopters IPC for Sling switch and circuit breaker P/Ns.

For early B2 models with a Honeywell switch panel, to connect the electrical harnesses to the switch panel it will be necessary to obtain two IDC contacts that are not supplied with the kit (Amp P/N 640632-3, Onboard Systems P/N 410-194-00).

Kit P/N 200-280-01, which uses a cargo hook with a mechanical release cable as the means for secondary release, can be converted to Kit P/N 200-280-02 using upgrade kit P/N 200-306-00 or to Kit P/N 200-280-03 using upgrade kit P/N 200-306-01. See Section 2.9 for installation instructions.

Table 1.9 Onboard Systems Bill of Materials – Upgrade Kit P/N 200-306-00

Part No.	Description	Qty
270-129-00	Electrical Release Harness	1
232-209-00	Cargo Hook/Slave Cylinder Assembly	1
290-839-02	Hook Bumper	1
232-165-00	Master Cylinder Assembly	1
290-884-00	Connector Bracket	1
505-014-00	Grommet	1
212-014-00	Bleed Kit	1
510-062-00	Washer	8
510-102-00	Nut	2
510-178-00	Cotter Pin	1
510-277-00	Screw	2
510-278-00	Washer	2
510-279-00	Nut	2
510-453-00	Bolt	2
512-005-00	Adel Clamp	4
512-003-00	Ty-wrap	5
510-486-00	Cherry Max Rivet	3
120-104-02	Owner's Manual	1
121-012-02	RFMS	1
122-015-00	Cargo Hook Service Manual	1
123-011-02	ICA	1

Bill of Materials continued

Table 1.10 Onboard Systems Bill of Materials – Upgrade Kit P/N 200-306-01

Part No.	Description	Qty
270-129-00	Electrical Release Harness	1
232-209-00	Cargo Hook/Slave Cylinder Assembly	1
290-839-02	Hook Bumper	1
232-165-01	Master Cylinder Assembly	1
290-884-00	Connector Bracket	1
291-105-00	Friction Knob	1
505-014-00	Grommet	1
212-014-00	Bleed Kit	1
510-062-00	Washer	8
510-102-00	Nut	2
510-178-00	Cotter Pin	1
510-277-00	Screw	2
510-278-00	Washer	2
510-279-00	Nut	2
510-453-00	Bolt	2
512-005-00	Adel Clamp	4
512-003-00	Ty-wrap	5
510-486-00	Cherry Max Rivet	3
120-104-02	Owner's Manual	1
121-012-02	RFMS	1
122-015-00	Cargo Hook Service Manual	1
123-011-02	ICA	1

Theory of Operation

The Cargo Hook Swing Suspension System is composed of:

- A suspended pyramid frame that supports the cargo hook.
- An electrical release system that provides means for release by pilot actuation of the push-button switch in the cockpit. When the push-button switch is pressed, it energizes the DC solenoid in the cargo hook, and the solenoid opens the latch in the internal mechanism.
- A hydraulic release system, which provides a means of releasing a cargo hook load in the event of an electrical release system failure. A lever mounted to the collective actuates it.
- Ground personnel may also release a load by the actuation of a lever located on the side of the cargo hook.
- A load weigh system, which is comprised of an indicator mounted to the RH door pillar within the cockpit connected to a load cell between the cargo hook and frame.

A load is attached to the cargo hook by passing a cargo sling ring into the throat of the load beam and pushing the ring against the upper portion of the load beam throat, which will cause the hook to close. In the closed position, a latch engages the load beam and latches it in this position.

To release the load, the latch is disengaged from the load beam. With the latch disengaged, the weight of the load causes the load beam to swing to its open position, and the cargo sling ring slides off the load beam. The load beam then remains in the open position awaiting the next load.

The Cargo Hook Swing Kit includes a fuel drain guard. The fuel drain guard protects the fuel drain valve on the helicopter from accidentally being opened or damaged by the cargo swing. The fuel drain valve is located on the bottom of the fuel tank and extends below the belly skin of the helicopter. In this position it is vulnerable to damage or uncommanded opening. The most common occurrence of the cargo hook swing suspension striking the fuel drain valve happens when the helicopter lands on snow or on uneven terrain. The swing has limited ground clearance and when the skid gear sinks into the snow, the swing suspension is pushed upward into the fuel drain valve, opening it and causing fuel to drain. The fuel drain valve can also be opened in flight by the swing suspension flying vertically due to aerodynamics when ferrying with no load or from recoil effects from releasing large cargo hook loads.

The systems feature several improvements over the original 200-280-00 system.

- The swing frame now has an additional gimbal near where the hook attaches and spherical rod ends on the sides. This allows the frame to distribute the loads between the four cables more effectively. As the swing moves from side to side under the helicopter the four cables all describe their own arcs. With a load applied to a rigid swing frame, something has to give to accommodate this geometry. This can be observed by moving the Onboard Systems gimbaled swing frame side to side under the helicopter and watching the gimbaled joint and the parallelism of the frame tubes. The frame flexes to accommodate the paths of the cable ends.

  With a rigid frame, such as supplied by Airbus Helicopters, swinging side to side with no load one or more of the cables goes slack. With a load on the hook, the cables don't go slack and the geometry is accommodated by flexure of the swing frame and the airframe. The more rigid the swing frame is, the more loads get transferred to the airframe. It is this principal that lead to the articulated swing frame of the current design. This reduces the stresses in both the airframe and the swing frame to a minimum.

- The swing frame supporting cables are now a simplified design with fewer parts and less weight.
- The attach fittings have been redesigned with larger replaceable bushings for longer life and more economical service.
- A ground strap is included to reduce the effects of static discharge.
- Improved routing of electrical cables from the hook to the belly plug.
- Hydraulic Manual Release System
  - No rigging required.
  - Improved cable routing for increased durability.
  - Inherently safer design than cable.

Section 2

Installation Instructions

These procedures are provided for the benefit of experienced aircraft maintenance facilities capable of carrying out the procedures. Those lacking the necessary expertise must not attempt them.

2.1 Electrical Wiring Installation

Install electrical harnesses (P/N 270-106-02 and P/N 270-108-00). Route them along the existing harnesses (reference Figure 2.1.1) while observing the following precautions:

- Pick up existing wire runs by opening existing cable clamps. Nylon ties alone may not be used for primary support.
- The distance between supports should not exceed 21 inches.
- Bend radius of wire or harness must not be less than 10 times the wire or harness diameter.
- Inspect and verify that the wire harness may not be manually deflected into a structure with a bend radius of less than 0.13".

Make the appropriate connections with terminal lugs. Secure the C-39 indicator harness along the canopy with clamps and connect to the C-39 indicator. Refer to Figure 2.1.6 for electrical schematic.

The P/N 270-106-02 electrical harness includes a data line for use with an Onboard Systems Data Recorder or Analog Meter. These items are not included under this STC. Attach connectors to data line per pin out in Table 2.1.1 to connect the Analog Slave Meter or Data Recorder to the electrical harness "DATA" line. If a data connector is present on the data line use harness P/N 270-059-00 to connect to Analog Slave Meter. If the accessory connector is not used, stow this line of the harness.

NOTICE

The data line may or may not be terminated with a data connector depending on manufacture date.

Table 2.1.1 Optional Equipment Connectors

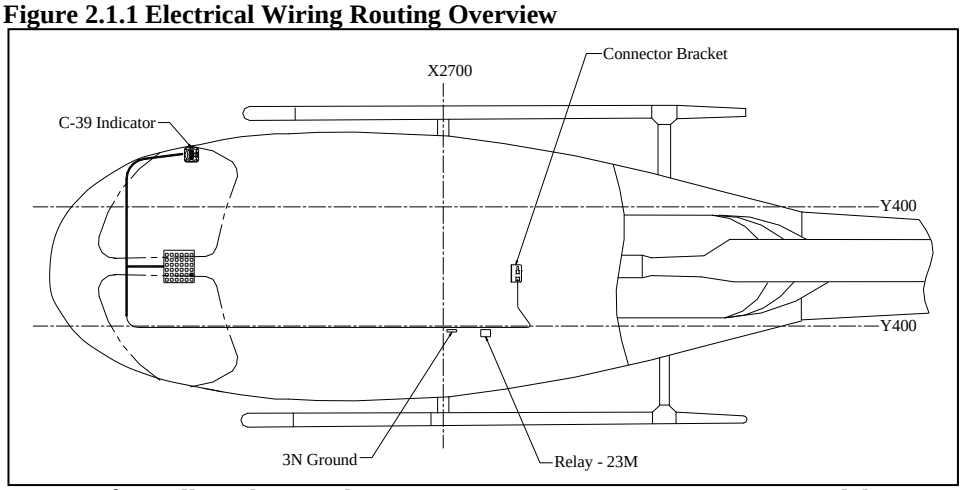
Analog Meter Connector P/N 410-130-00 Mfg P/N: MS3126F10-6P		
Pin	Color	Function
A	WH	Power
B	WH/GN	Clock
C	WH/OR	Data
D	WH/BL	Ground
E	Shield	Shield

Data Recorder Connector P/N 410-011-00, 410-057-00 & 410-020-00		
Pin	Color	Function
1	WH/BL	Ground
3	WH	Power
5	Shield	Shield
7	WH/GN	Clock Signal
9	WH/OR	Data Signal
4	Red*	Flight Switch
2	Purple*	Cap. Switch

*Optional

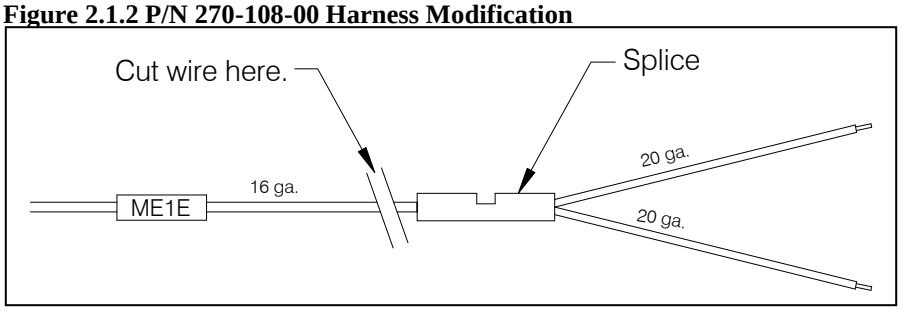
2.1 Electrical Wiring Installation continued

Figure 2.1.1 Electrical Wiring Routing Overview



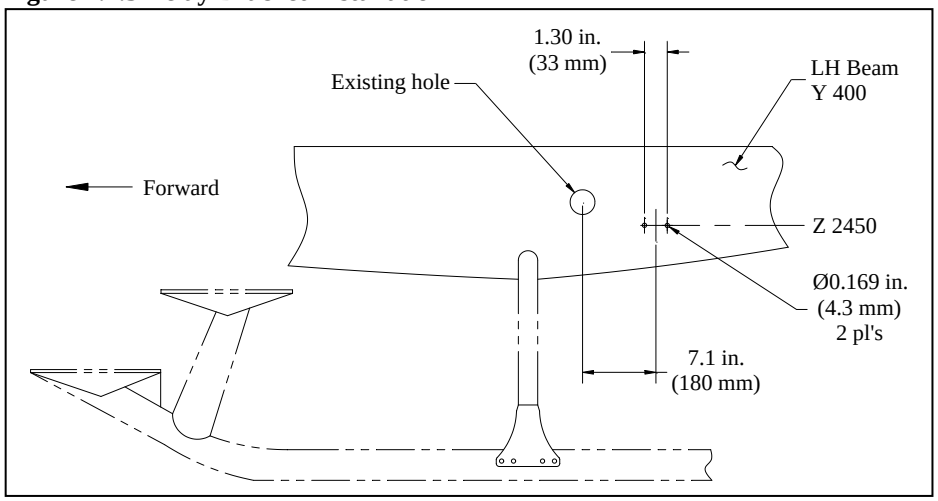
If installing the wire harnesses on a newer AS350B2 or B3 model equipped with a switch panel of circuit breaker design (Airbus Helicopters mod. #07-3274 incorporated) the electrical harness P/N 270-108-00 requires modification. Cut the ME1E wire off just prior to the butt splice and discard the splice and the 20 ga. wires.

Figure 2.1.2 P/N 270-108-00 Harness Modification



- In preparation for installing the Relay Bracket (P/N 290-783-00) create two holes in the LH beam at Y400 as illustrated in Figure 2.1.3.

Figure 2.1.3 Relay Bracket Installation

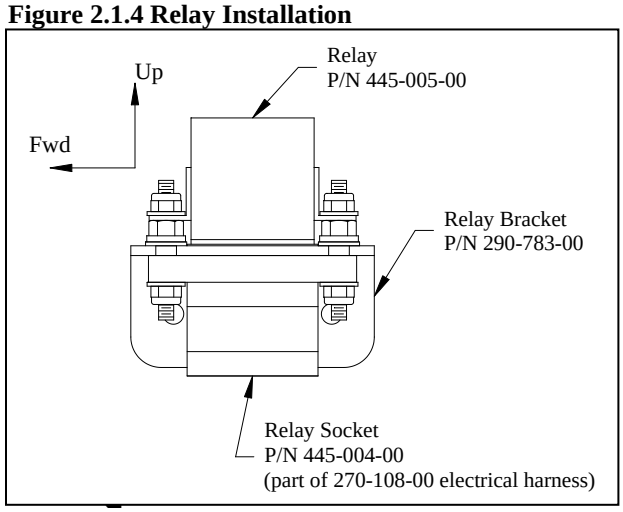


- Secure Relay Bracket with two screws (P/N 510-277-00), two washers (P/N 510-278-00), and two nuts (P/N 510-279-00).

2.1 Electrical Wiring Installation continued

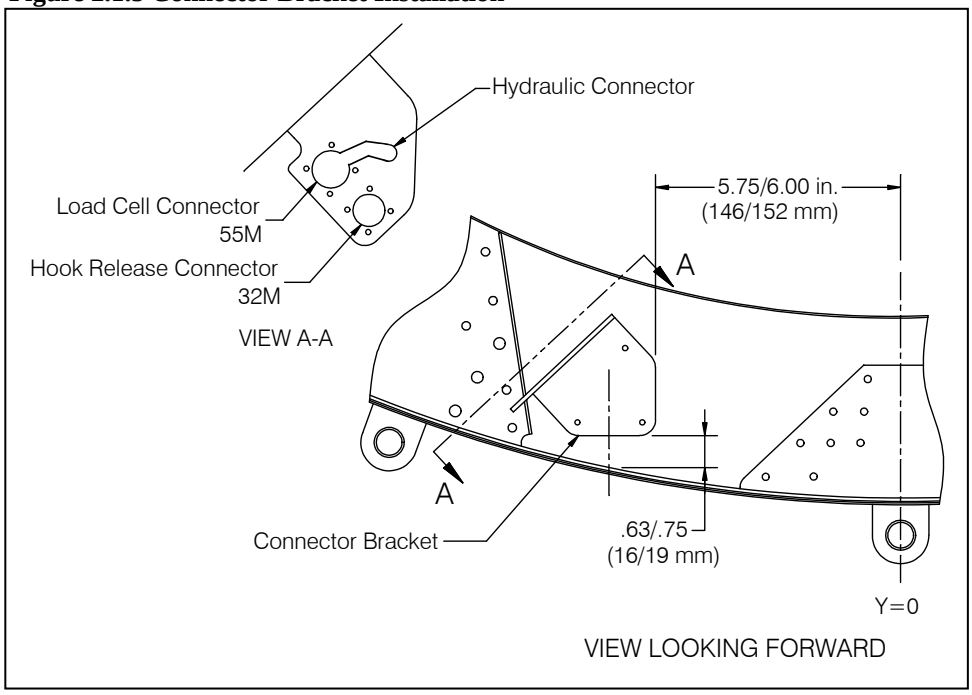
- Place relay socket (part of 270-108-00 electrical harness) into relay bracket mounting holes from below and secure to relay and relay bracket with hardware provided with relay (as illustrated below).

Figure 2.1.4 Relay Installation



- Locate Connector Bracket (P/N 290-884-00) at forward fuel tank support frame as illustrated below.
- Drill out pilot holes in bracket to 0.129/0.132" (3.2/3.4 mm) diameter and drill fuel tank support to match.
- Secure Connector Bracket to fuel tank support with three rivets (P/N 510-486-00).
- The hook release connector, 32M, and load cell connector, 55M, will be installed on the Connector Bracket later in the installation. The hydraulic connector must be installed prior to installing the connectors.

Figure 2.1.5 Connector Bracket Installation

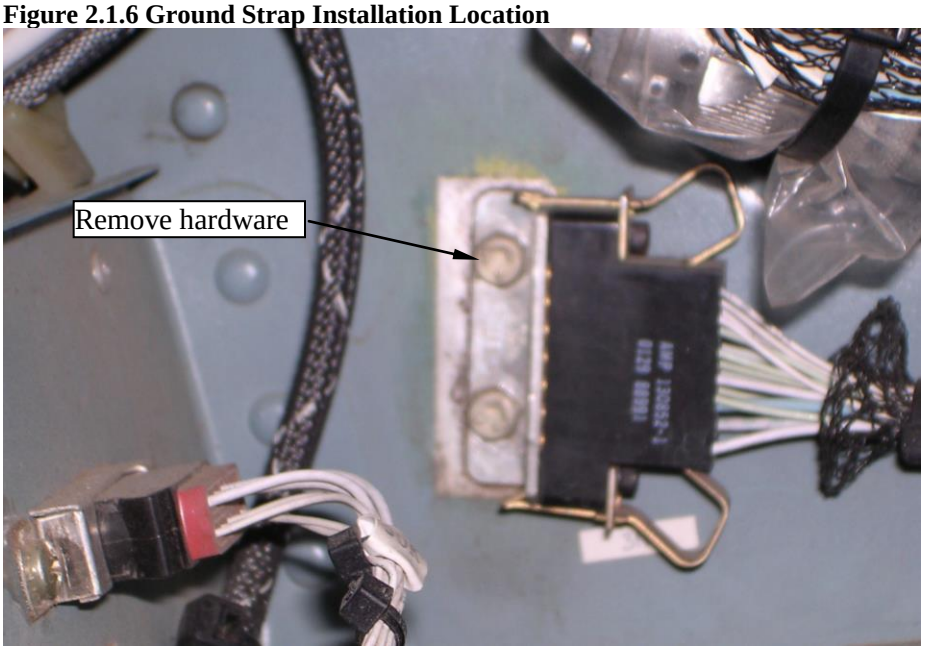


2.1 Electrical Wiring Installation continued

Install the Ground Strap (P/N 270-125-00) terminal at the 3N ground connector bracket at the LH airframe beam at Y400 per the following:

- Remove upper mounting hardware (see below) for 3N connector bracket and retain.
- Prepare the surface for electrical bonding per Airbus Helicopters electrical bonding procedure. Refer to section 20.02.07 of the Airbus Helicopters Standard Practices Manual.
- Install ground strap terminal, re-using hardware removed per above step.

Figure 2.1.6 Ground Strap Installation Location



- Route the Ground Strap to the load weigh and electrical release harnesses installed previously and route with these harnesses, while securing with ty-wraps, to their termination points at the Connector Bracket.
- Route the Ground Strap around the Connector Bracket and position it such that the disconnect fitting at the end can be routed through the connector access hole and extended below the lower fairing (when installed).
- Re-install lower fairings if both the electric and hydraulic systems have been installed.

2.1 Electrical Wiring Installation continued

The electrical schematic for the electrical release system and the load weigh system is shown below along with the aircraft's interface points. Airbus Helicopters modification #'s 07-4280 and 07-3450 are reflected below. Earlier Airbus Helicopters configurations which affected how and where wire numbers ME1E, ME2E and ME10E of the electrical release harness and load weigh harness interface with the helicopter are shown on the following page. Refer to the applicable Airbus Helicopters Wiring Diagrams Manual for additional information and for other cargo hook aircraft side wiring configurations that may not be shown.

For the C-39 Indicator backlighting, install wire 2LK71E to an available pin in the instrument panel or console lighting circuit (31L for pre-mod 07-4280), at 28 volts the indicator's internal bulb draws 25 mA.

If existing Airbus Helicopters cargo hook or load weigh wiring is installed and terminated at the locations below, remove the wires completely or remove from connectors and cap and stow them.

Figure 2.1.7 Electrical Schematic

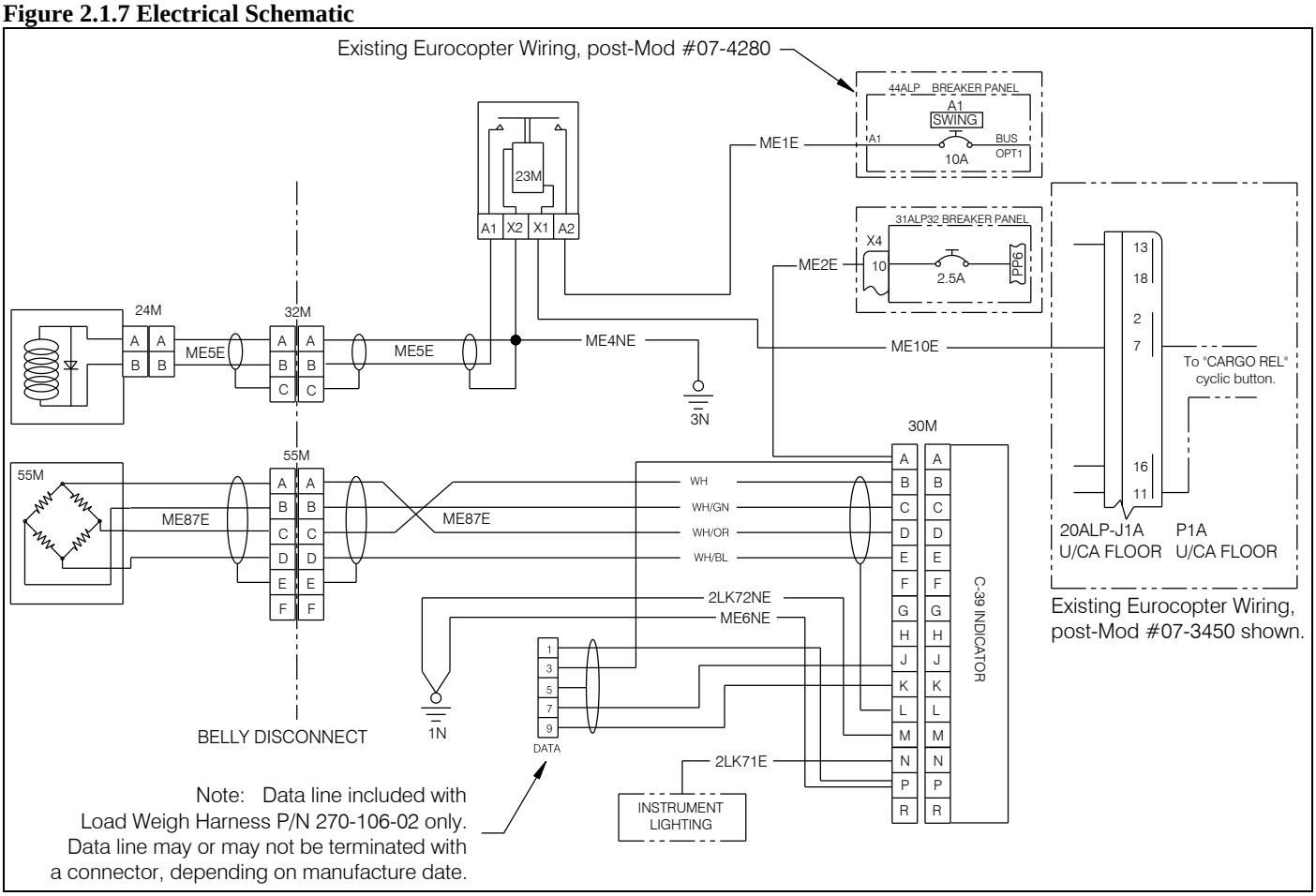
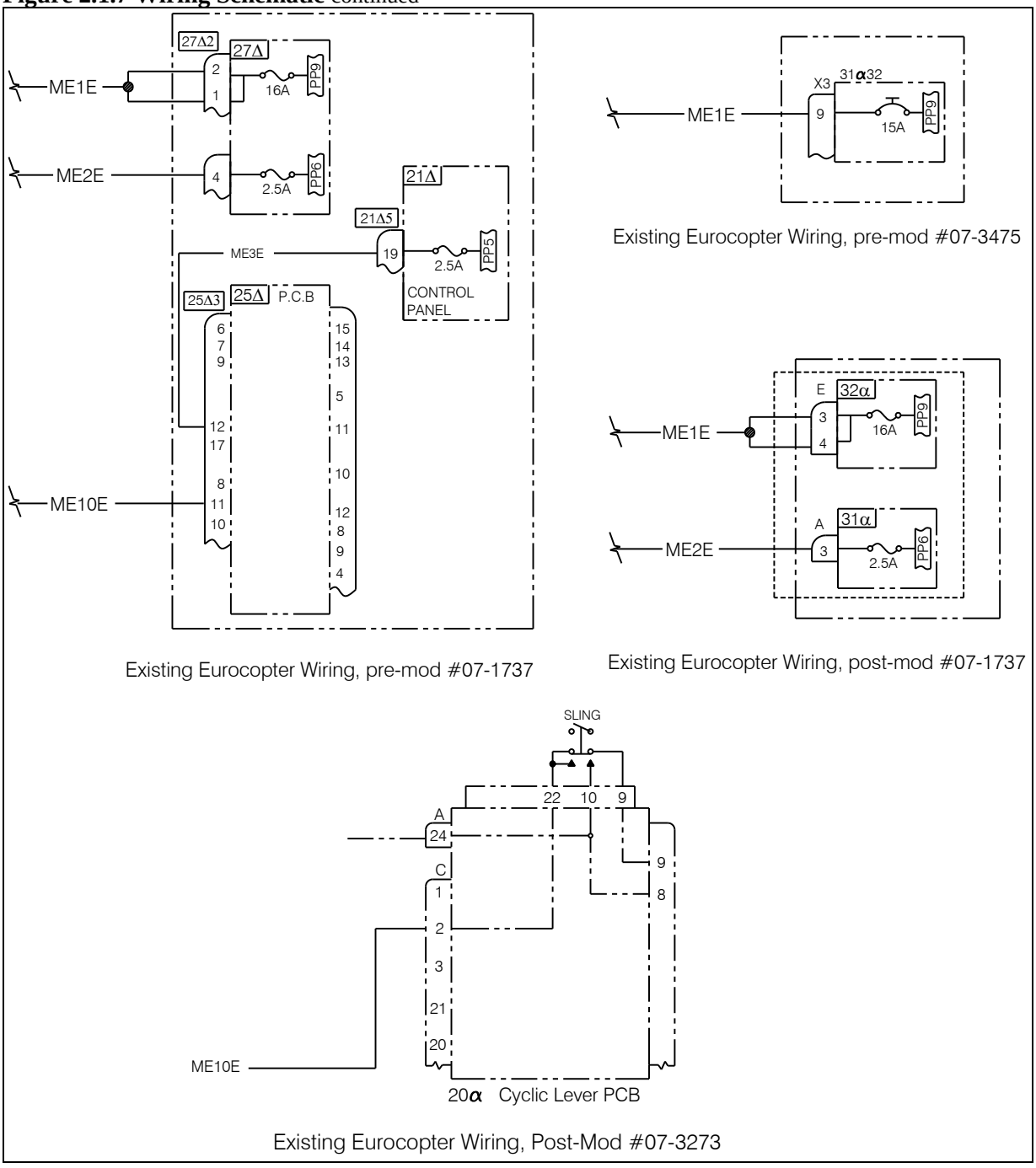


Figure 2.1.7 Wiring Schematic continued

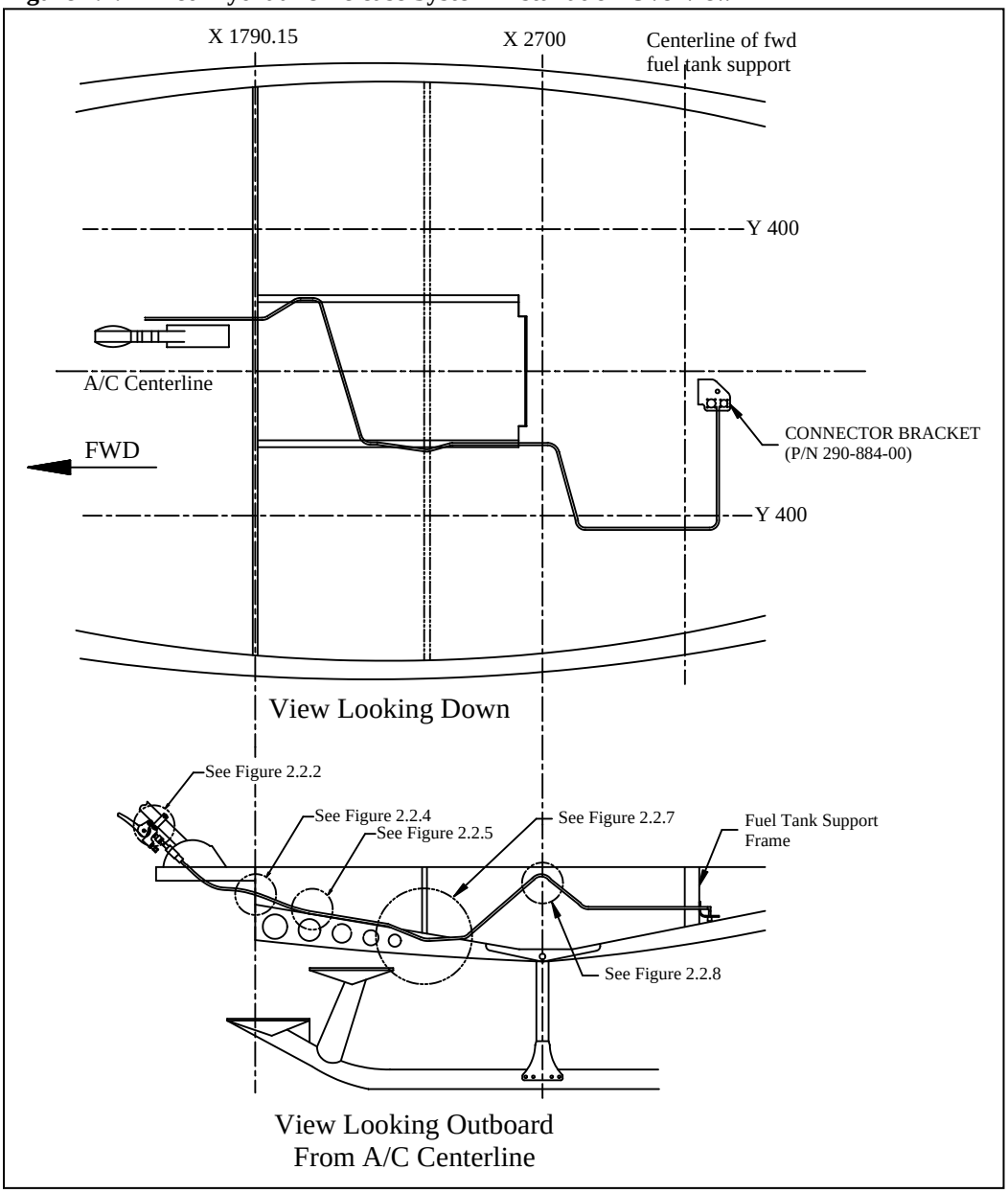


2.2 Fixed Hydraulic Release System Installation

Remove the lower fairings on the helicopter in order to obtain access to hydraulic hose routing areas.

The hydraulic release system installation consists of a fixed section and a removable section. The fixed section is routed from the release lever at the collective, underneath the cabin floor, and aft to meet up with the electrical release harness (as shown in Figure 2.2.1). Figure 2.2.1 is an overview of the hose routing and the figures following detail the support installations at various points.

Figure 2.2.1 Fixed Hydraulic Release System Installation Overview



2.2 Fixed Hydraulic Release System Installation continued

The hydraulic release system is supplied dry. It is recommended that the system be filled and bled on the bench before installing on the helicopter. Refer to section 2.6 for filling and bleeding instructions.

Mount the Release Lever Assembly to the collective stick with the Clamp Half (P/N 290-753-00) and two screws (P/N 510-390-00) provided pre-assembled on the assembly, as illustrated below.

NOTICE

An optional clamp half (P/N 291-523-00) which can be installed in place of clamp half P/N 290-753-00 and includes a 10 mm (.394 in.) mounting hole (e.g. – for an accessory) is available from the factory.

NOTICE

If installing kit P/N 200-280-03 (for the AS350B3 model) a shorter friction adjustment knob for the collective is required because of interference with the master cylinder assembly reservoir. This item is supplied as P/N 291-105-00. See below for installation instructions.

Figure 2.2.2 Release Lever Installation

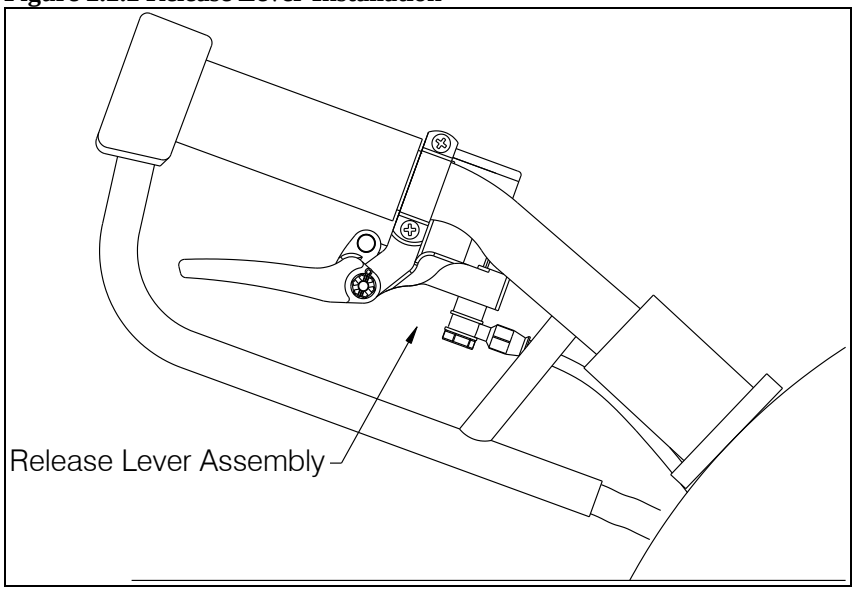
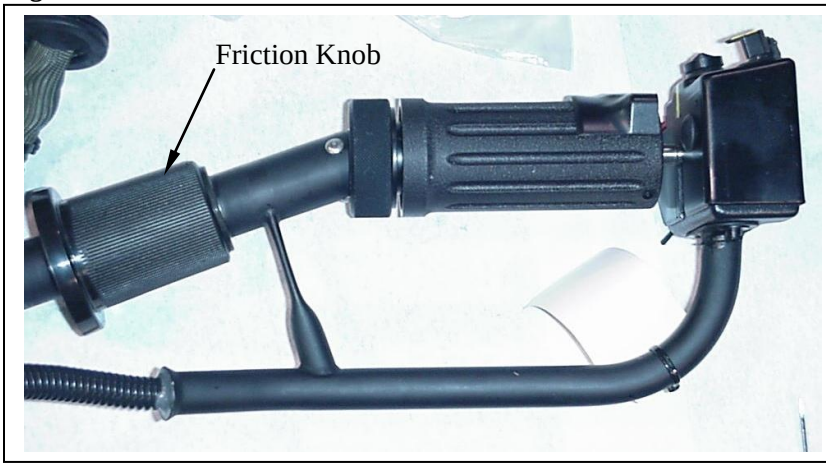


Figure 2.2.3 B3 Collective



- Remove the collective to the extent necessary to remove the friction knob. Refer to Airbus Helicopters maintenance manual for removal and re-installation instructions.
- Replace the OEM friction knob with the friction knob P/N 291-105-00 provided.
- Re-install collective per Airbus Helicopters maintenance manual instructions.
- Mount the Release Lever Assembly (P/N 232-165-01) to the collective stick with the Clamp Half (P/N 290-753-00) and two screws (P/N 510-390-00) provided pre-assembled on the assembly, as illustrated below.

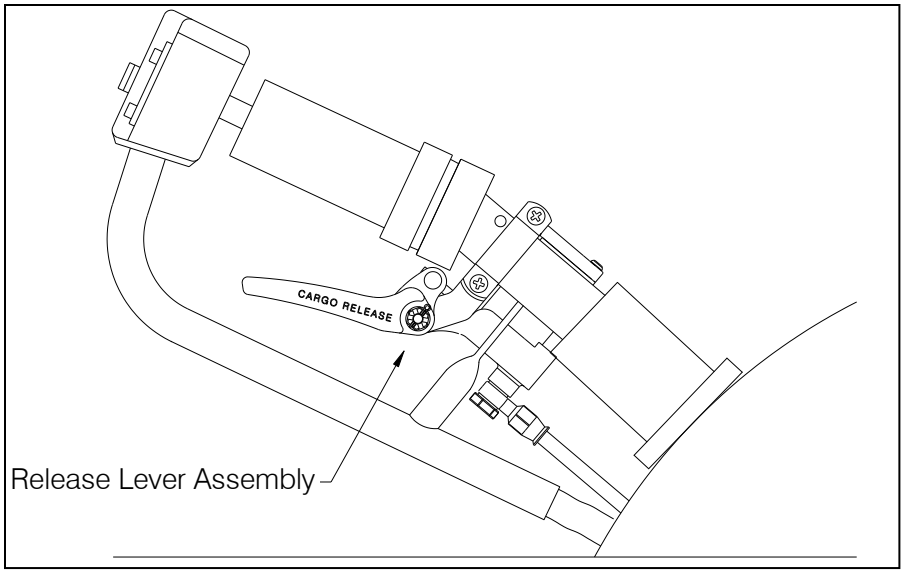
NOTICE

Locate the Release Lever Assembly on the collective shaft as close as possible to the throttle twist grip (as shown in Figure 2.2.4).

- Adjust the Friction Knob to its outermost position and verify that there is clearance with the reservoir on the release lever assembly.

2.2 Fixed Hydraulic Release System Installation continued

Figure 2.2.4 AS350B3 Release Lever Installation



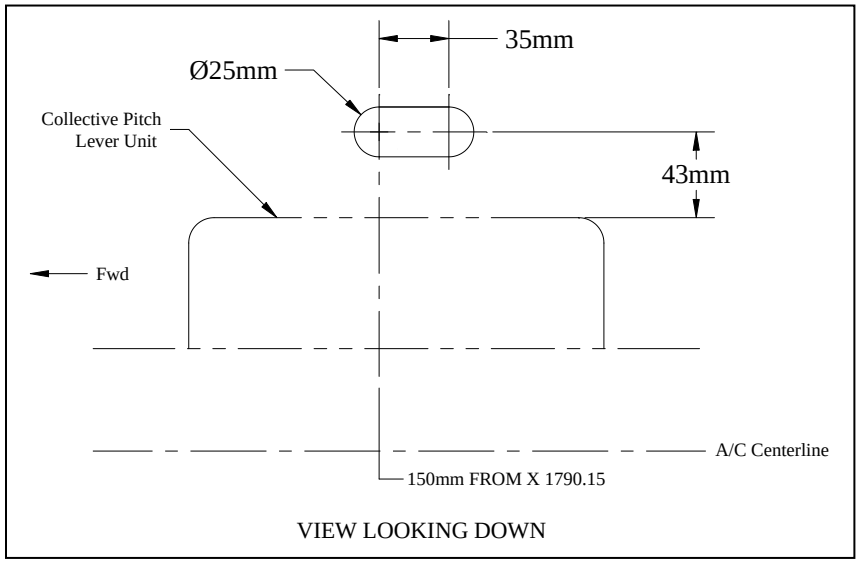
NOTICE

The lever position is to be adjusted, if necessary, at installation check out, after the system is filled and bled.

- ❑ Route the hose to underneath the cabin floor through the existing slot.

  If the slot in floor does not exist, create one with dimensions as shown below in the cabin floor 43 mm from the collective pitch lever unit and 150 mm forward of X1790.15 (see below) and install the grommet (Airbus Helicopters P/N DG-38).

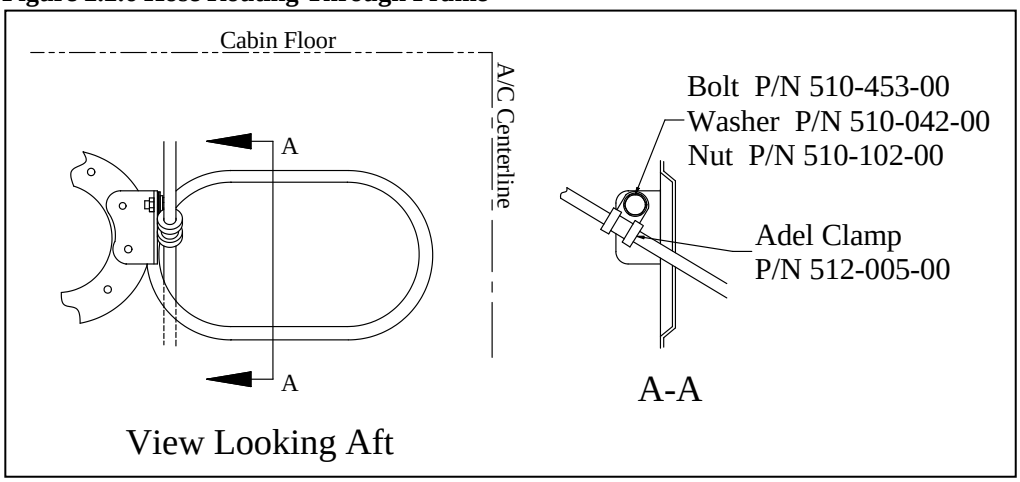
Figure 2.2.5 Cabin Floor Hole Detail



2.2 Fixed Hydraulic Release System Installation continued

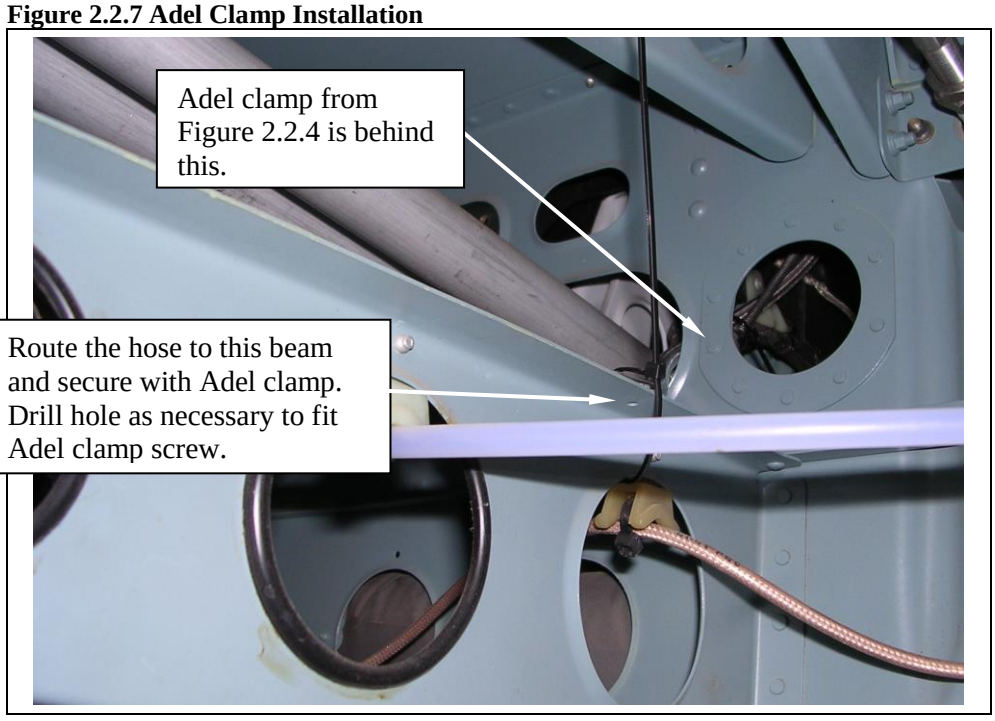
- Underneath the floor, route the hydraulic hose through an existing hole in the frame immediately aft of the collective. Secure the hose at this point with an adel clamp (P/N 512-005-00). Fasten the adel clamp to the existing bracket (Airbus Helicopters P/N 350A86-0020-33) with hardware as illustrated below.

Figure 2.2.6 Hose Routing Through Frame



- Aft of the frame, route the hose along the top of the structural member (shown below) and secure with adel clamp (P/N 512-005-00) at location shown.

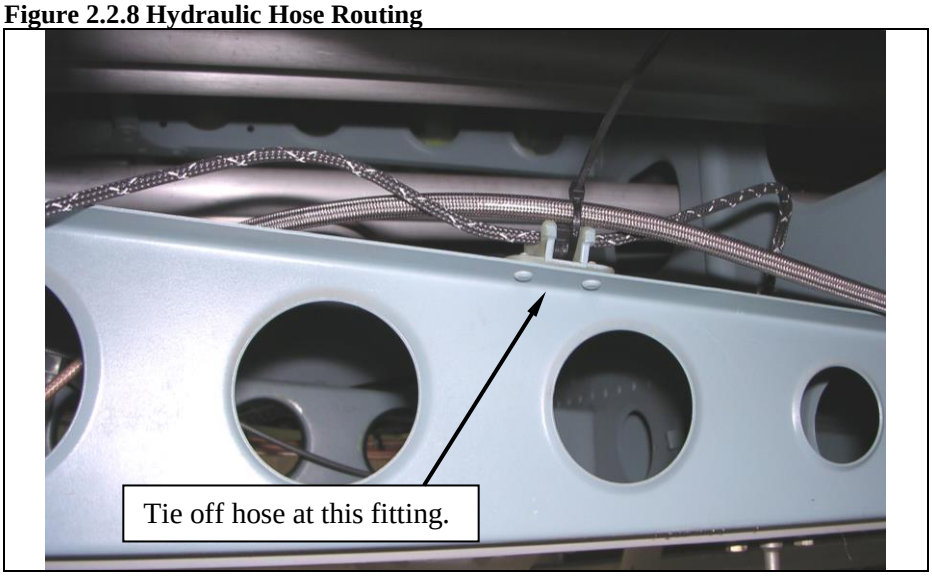
Figure 2.2.7 Adel Clamp Installation


Adel clamp from Figure 2.2.4 is behind this.
Route the hose to this beam and secure with Adel clamp. Drill hole as necessary to fit Adel clamp screw.

2.2 Fixed Hydraulic Release System Installation continued

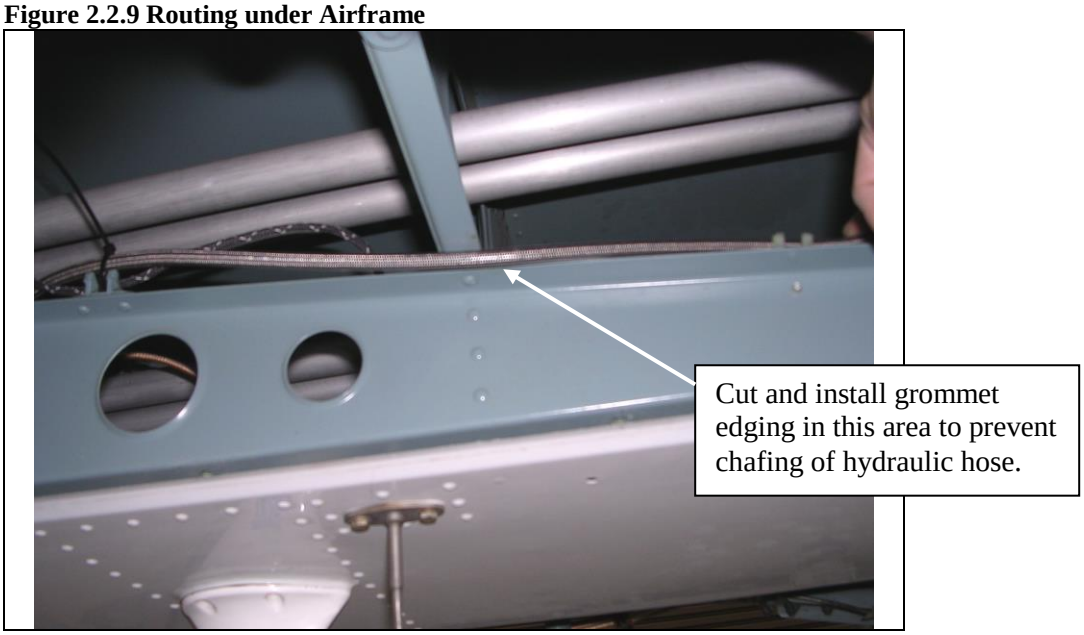
- Aft of the clamp installed in Figure 2.2.3, route the hose inboard and aft across the airframe centerline to the identical structural member on the left side of the airframe. Secure hose to fitting on top of structural member with ty-wrap as shown below. Ensure the hose is secured so that it does not interfere with the control rods.

Figure 2.2.8 Hydraulic Hose Routing



- Route the hose under the airframe support (as shown below) and secure the hose to the fitting on top of the structural member aft of the airframe support. Install grommet edging (P/N 500-065-00) as necessary to protect hydraulic hose from chafing.

Figure 2.2.9 Routing under Airframe



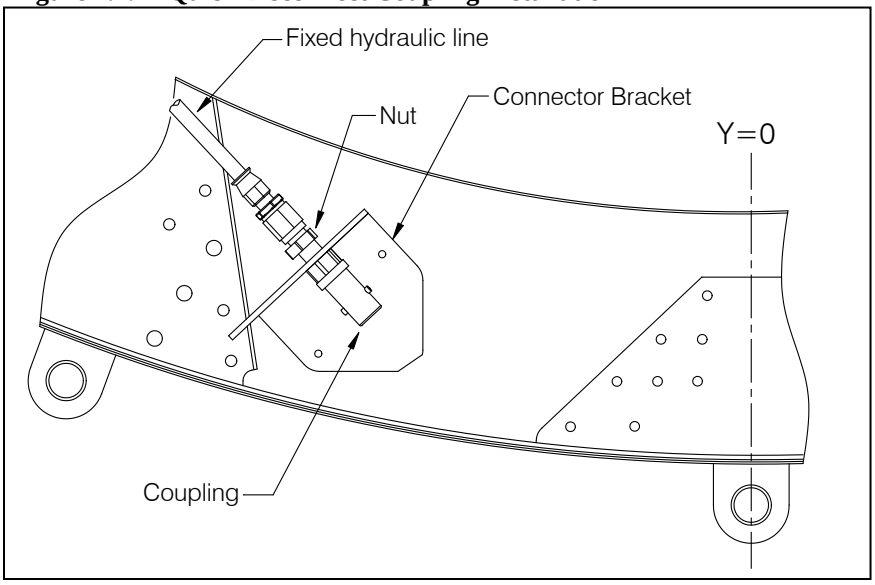
2.2 Fixed Hydraulic Release System Installation continued

- Route the hose up through the rear cabin bulkhead as shown in Figure 2.2.10. Split and install grommet (P/N 505-014-00) in hole after hose is routed through.

Figure 2.2.10 Hose Routing through Rear Cabin Bulkhead

- Aft of the rear cabin bulkhead pick up existing electrical harness runs and secure hydraulic hose using ty-wraps. The hose will route outboard of Y400 and follow the electrical release harness (installed previously) to the connector bracket.
- Pass the hydraulic quick disconnect coupling through the load cell electrical connector hole. Slide the fitting to the end of the slot and tighten the jamb nut securely against the Connector Bracket.

Figure 2.2.11 Quick Disconnect Coupling Installation



2.2 Fixed Hydraulic Release System Installation continued

- Install the hook release connector 32M and load cell connector 55M with screws (P/N 510-481-00), washers (P/N 510-062-00), and nuts (P/N 510-029-00).

Install screws with their heads on the bottom side of bracket flange (if nuts are installed on bottom side they will interfere with mating connector). See figure 2.4.1 for location of each connector.

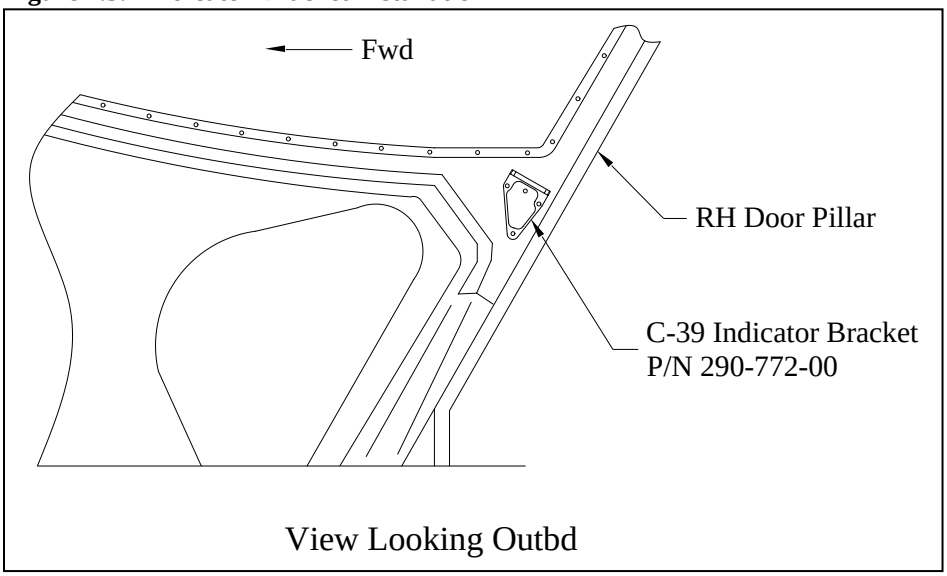
- Install electrical markers (P/N 215-165-00) on the Connector Bracket adjacent to the corresponding connectors.

2.3 Cockpit Indicator Installation

The Indicator is mounted on the RH door pillar. If nut clips are not pre-installed in the door pillar, install them per the following.

- Hold the Indicator Bracket (P/N 290-772-00) at a location as shown below and transfer its hole pattern to the door pillar.

Figure 2.3.1 Indicator Bracket Installation

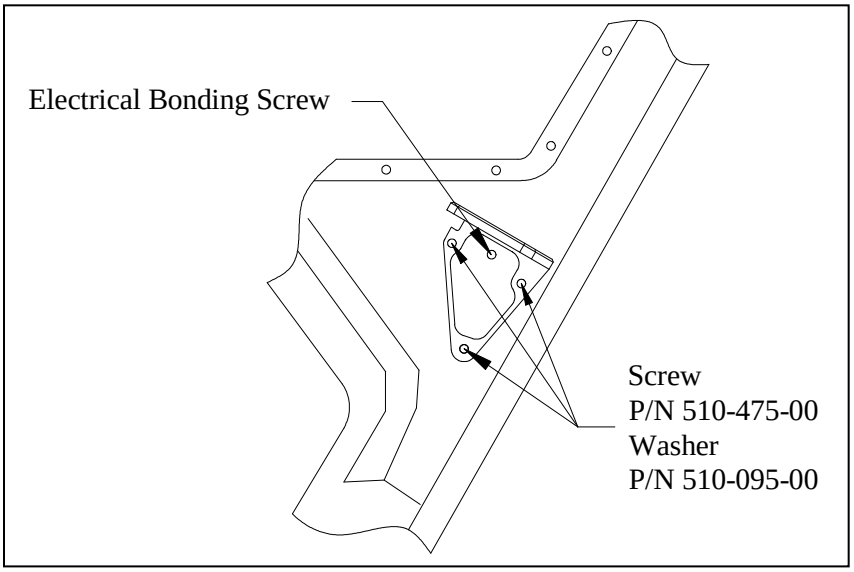


2.3 Cockpit Indicator Installation continued

- Drill three mounting holes in the RH door pillar to install the nut clips. Reuse the electrical bonding screw at the fourth location (see below).

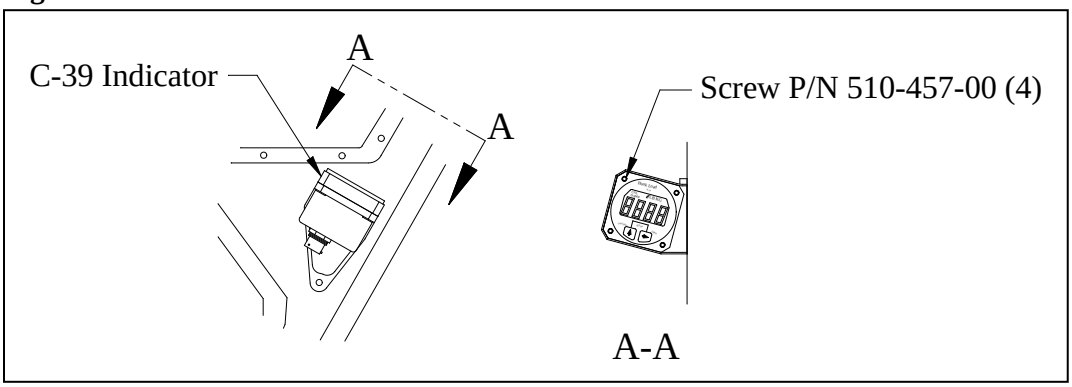
- After completing electrical bonding, install the three nut clips (Airbus Helicopters P/N SL211M5-1) and fasten Indicator Bracket with three screws (P/N 510-475-00) and three washers (P/N 510-095-00).

Figure 2.3.2 Indicator Bracket Hardware



- Install C-39 Indicator (P/N 210-095-00 or P/N 210-095-04) onto bracket with hardware as illustrated below.

Figure 2.3.3 C-39 Indicator Installation



2.4 Fuel Drain Guard Installation

NOTICE

If installing the Swing Kit on a helicopter that has NOT been retrofitted with the B-2 style dual fuel pump type tank (Airbus Helicopters P/N 350A55-1015-0251), the fuel drain guard installation is omitted. Skip to section 2.5.

In preparation for installing the fuel drain guard:

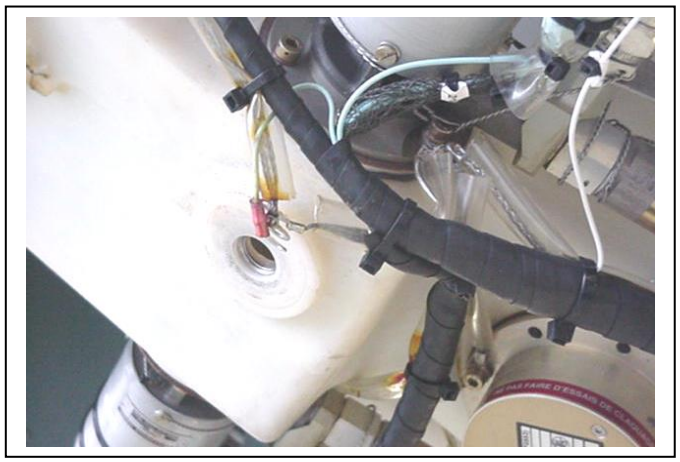
- Obtain ProSeal sealant (P/N PR1422-B).
- Verify that the fuel drain lever has been modified per AD 2005-03-08 requirements.
- Completely drain the fuel from the fuel tank.
- Begin disassembly of the fuel drain assembly by disconnecting the fuel drain control cable from the Lever (Airbus Helicopters P/N 350A55-1043-20). To free the control cable, remove the Sleeve (Airbus Helicopters P/N N1-5ALU) and Cable Grip (Airbus Helicopters P/N 58-2-009). See Figure 2.4.1.

Figure 2.4.1 Removing Control Cable

2.4 Fuel Drain Guard Installation continued

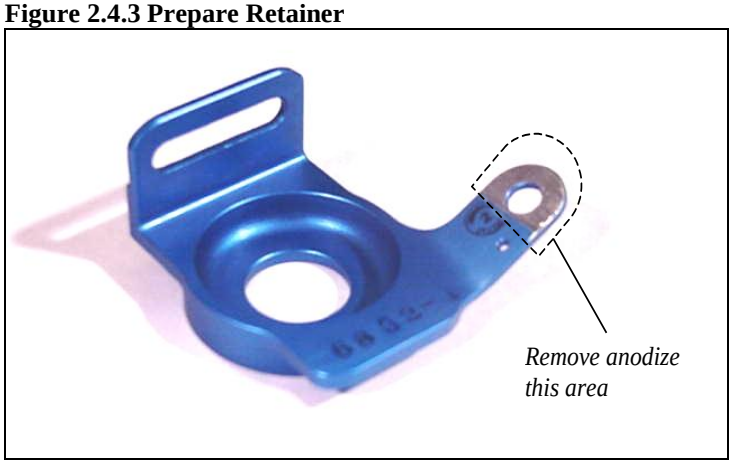
- Remove and retain spring (Airbus Helicopters P/N 350A55-1044-21) and Lever.
- Remove the cable support bracket. Retain the two attachment screws.
- Remove the connections from the common ground point on the lever retainer.
- Remove the safety wire securing the Fuel Drain Valve (Airbus Helicopters P/N A1282Z3). Remove the Fuel Drain Valve and the Retainer from the tank. Discard used Fuel Valve Seal (Airbus Helicopters P/N SD16X-21P).
- Remove the residual sealant from the tank, taking care to not mar the sealing surface. Prepare the area for sealing per Airbus Helicopters Standard Practices Manual.

Figure 2.4.2 Fuel Drain Disassembly Complete



- Prepare the Retainer (P/N 290-888-00) for electrical bonding by removing anodize from the area shown in Figure 2.4.3.

Figure 2.4.3 Prepare Retainer



2.4 Fuel Drain Guard Installation continued

- Prepare PR1422-B or equivalent fuel tank sealant per Airbus Helicopters Standard Practices Manual. Apply sealant to Retainer as shown in Figure 2.4.4. Retain unused sealant to ensure proper cure.

Figure 2.4.4 Apply Sealant

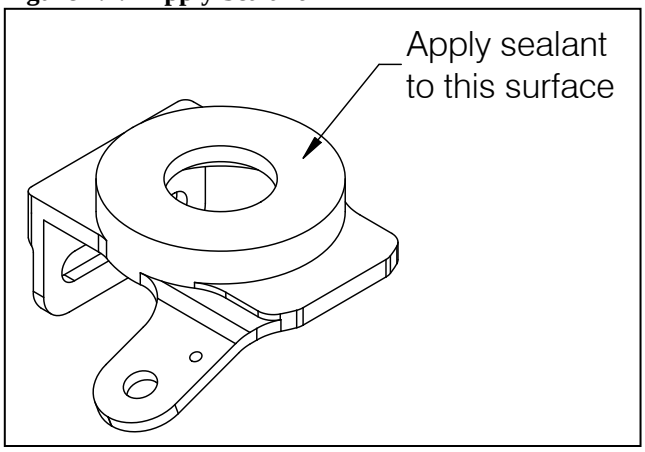
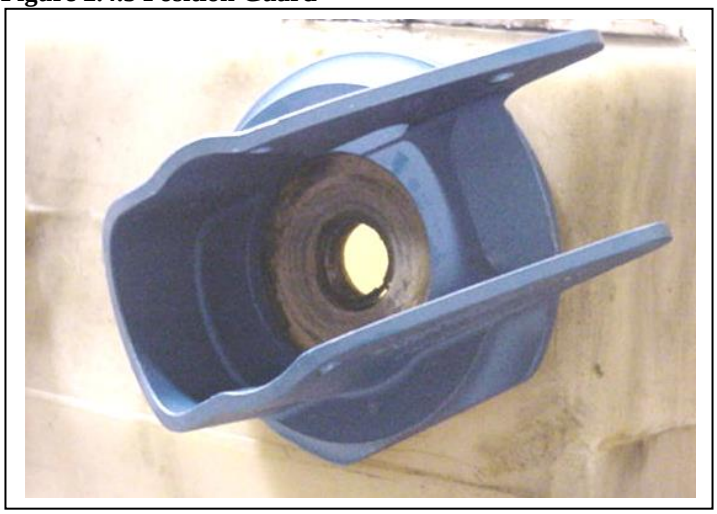


Figure 2.4.5 Position Guard



NOTICE

On some aircraft, the guard may have to be modified to fit the key on the fuel tank. In these cases, the sides of the clearance slot on the guard should be widened the minimum amount necessary in order to fit over the key. See figures 2.4.6 and 2.4.7.

2.4 Fuel Drain Guard Installation continued

Figure 2.4.6 Fuel Tank Key

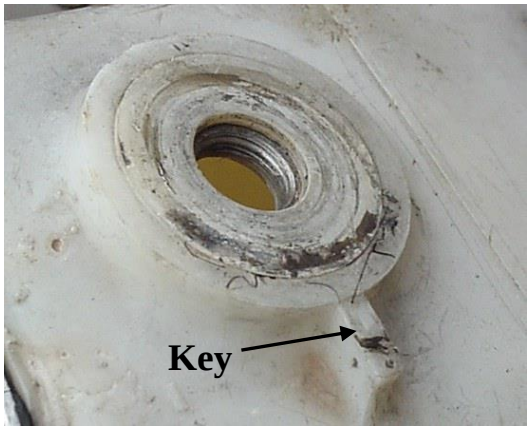
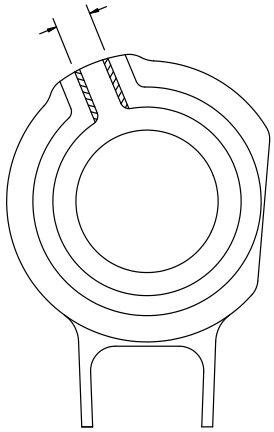


Figure 2.4.7 Modification of Guard Slot



- Carefully place the Retainer inside the Guard by inserting the tab through the slot in the Guard. Press the retainer to the tank firmly and center it about the drain hole. See Figure 2.4.8.

Figure 2.4.8 Retainer/Guard Assembly

- Secure the Guard and Retainer by re-installing the Fuel Drain Valve with Fuel Valve Seal, P/N 610-024-00 (Airbus Helicopters P/N SD16X-21P). Use a flat-blade screwdriver to prevent the Retainer from twisting when tightening the Fuel Drain Valve. Torque per Airbus Helicopters specifications.

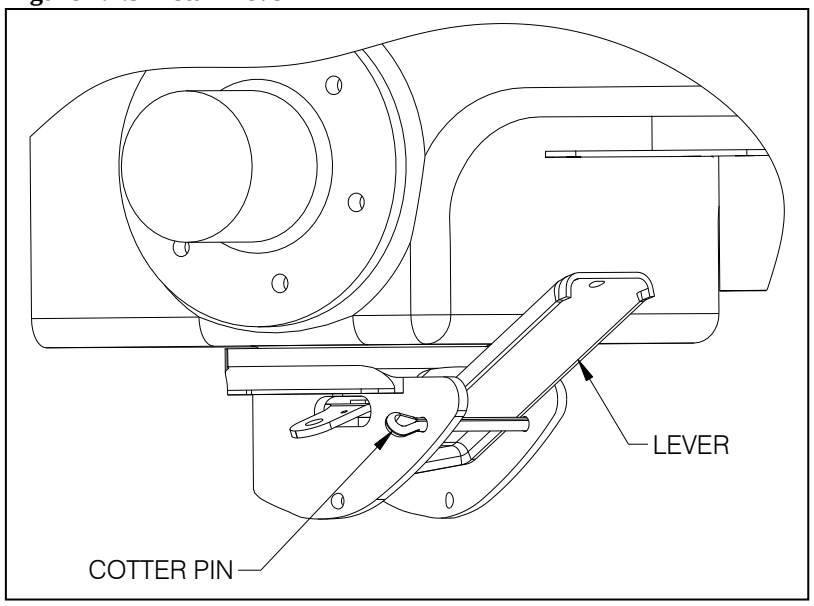
NOTICE

The guard is not intended to fit tightly with the fuel tank. When properly installed, the guard should have freedom to move slightly.

2.4 Fuel Drain Guard Installation continued

- Secure the Fuel Drain Valve with safety wire using the small hole in the retainer tab.
- Re-install the electrical connections to the new Retainer tab per Airbus Helicopters Electrical Bonding Procedure. Refer to Airbus Helicopters Standard Practices Manual, 20.02.07.
- Install the Lever by placing it in Retainer slot and rotating upwards. Secure with cotter pin (P/N 510-526-00). See Figure 2.4.9.

Figure 2.4.9 Install Lever



- Install a second cotter pin through the other holes in the Guard (this cotter pin is for valve protection only and is not used for rigging purposes).
- Prepare to install Bracket (P/N 290-893-00) by threading the control cable through the Bracket hole. Install Bracket using the two screws removed previously.
- Pass the cable through the spring and then the Lever. Install the Sleeve and Cable Grip.

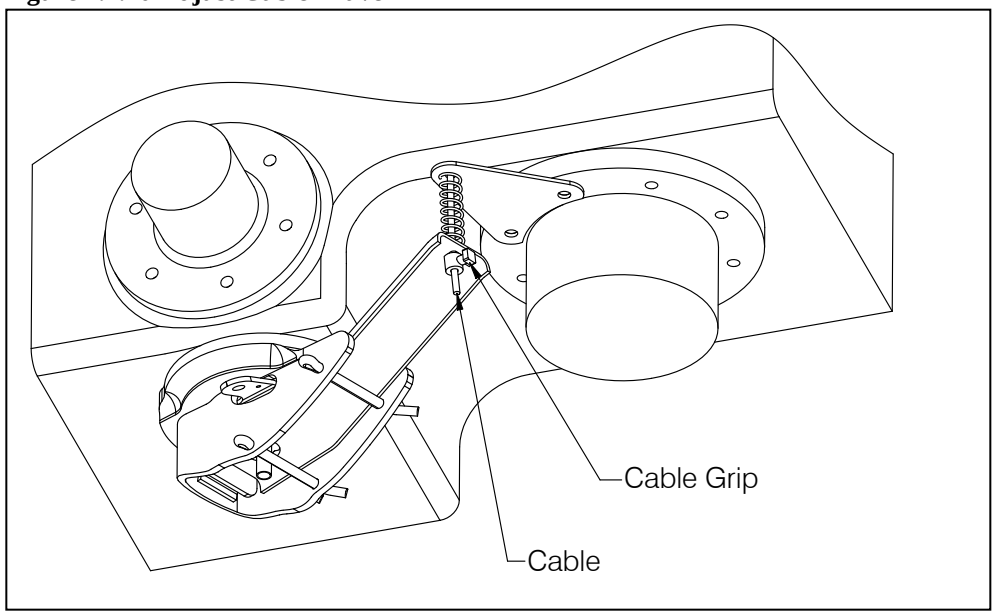
⚠ CAUTION

To avoid inadvertent fuel loss, Airbus Helicopters P/N 58-2-009 Cable Grip must be used with this installation.

- Adjust the cable travel by doing the following: allow the lever to rest against the cotter pin stop. Slide the Cable Grip up to the bottom of the lever and secure. See Figure 2.4.10.

2.4 Fuel Drain Guard Installation continued

Figure 2.4.10 Adjust Cable Travel



- Check the cable adjustment with the release handle on the side of the aircraft. There should be a minimum of .25 inch (6mm) cable travel before valve opens. Adjust the Cable Grip as required.
- Trim excess cable to within .25" of cable grip.
- Allow the sealant to cure per Airbus Helicopters Standard Practices Manual before adding fuel. Verify proper cure of unused sealant.
- Add fuel to the tank and check for leakage.
- Re-install aft lower cowling. Check for clearance between Guard and cowling. If required trim cowling cutout to provide a min of .125 inch (3.5mm) clearance between the cowling and guard. See Figure 2.4.11 for completed installation.

Figure 2.4.11 Installation Complete

2.5 Shackle Assembly Installation

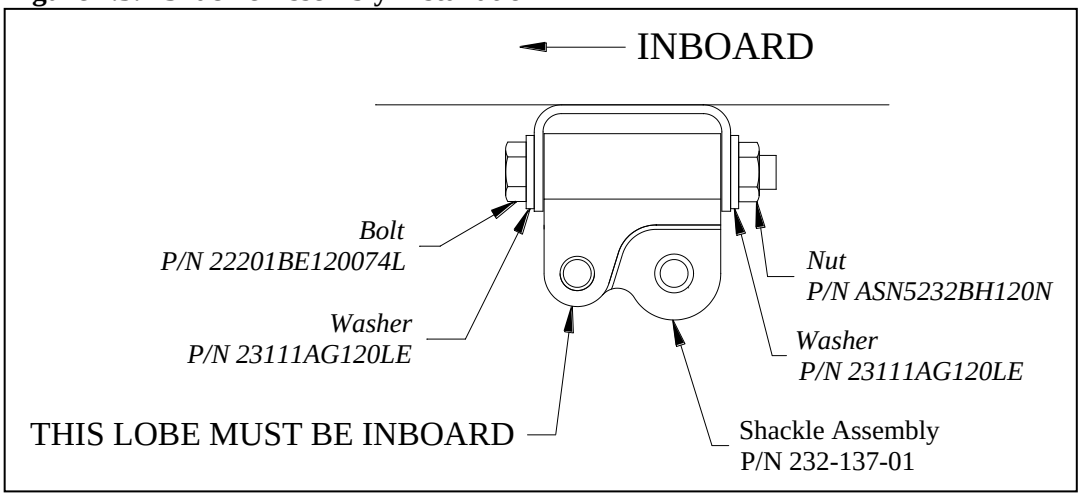
Airbus Helicopters landing gear fittings (P/N 350A41-1097-20) are standard on B3 and optional on earlier models. If not present, install per Airbus Helicopters modification 07-2772.

Attach the four Shackle Assemblies (P/N 232-137-01) to the helicopter landing gear fittings with hardware as illustrated in Figure 2.5.1 (Airbus Helicopters part numbers are shown in italics).

NOTICE

The smaller lobe of the shackle must be inboard as shown in the Figure 2.5.1 below.

Figure 2.5.1 Shackle Assembly Installation



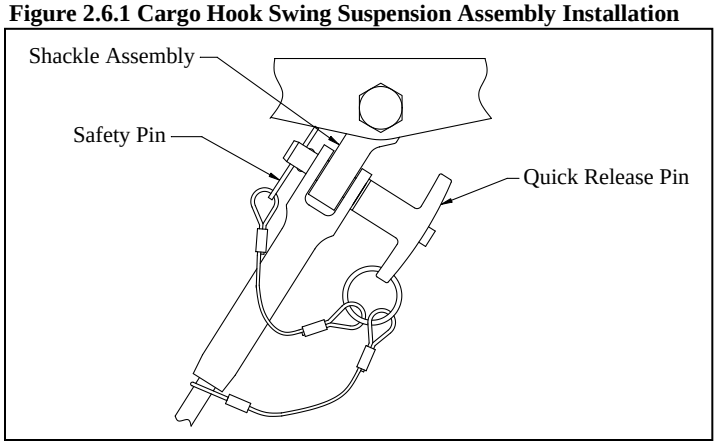
- Install the cable assemblies onto the swing suspension frame with hardware provided pre-assembled onto the cable clevis end. Disassemble hardware from cable clevis end and apply grease to pivot bolts, recommended lubricants are AeroShell 17, MIL-G-21164 or Mobilgrease 28, MIL-G-81322. Fasten the two shorter cable assemblies (P/N 232-140-01) onto the forward pivot points of the suspension frame and the two longer cable assemblies (P/N 232-141-01) to the aft pivot points. The forward end of the suspension is determined by the orientation of the cargo hook. **When the suspension is installed, the cargo hook load beam must point to the left side of the helicopter.**

2.6 Swing Suspension Installation

- Install the Swing Suspension onto the aircraft by attaching the suspension cable clevises to the **INBOARD** holes of the Shackle Assemblies with the Quick Release Pins* (P/N 290-851-00) as shown in Figure 2.6.1. Install the attached safety pins at each Quick Release Pin.

  * Apply grease to quick release pins before installing, recommended lubricants are AeroShell 17, MIL-G-21164 or Mobilgrease 28, MIL-G-81322.

Figure 2.6.1 Cargo Hook Swing Suspension Assembly Installation



- Connect the end of the cargo hook electrical release cable to the fixed electrical release connector (32M) installed per Section 2.1. See Table 2.2 for pin out information.

Table 2.2 Cargo Hook Connector

Pin	Function
A	Ground
B	Power

CAUTION

The cargo hook is equipped with a suppression diode that will be damaged if the cargo hook electrical connection is reversed.

- Connect the Ground Strap from the hook to the fixed ground strap installed per Section 2.1.
- Connect the end of the load cell cable to the fixed load weigh harness connector (55M) installed per Section 2.1.
- Connect the hydraulic hose from the hook to the fitting installed at the belly of the helicopter.

2.7 Filling Hydraulic Release System

Each hydraulic system is typically shipped dry. A label affixed to the Master Cylinder and Slave Cylinder assemblies will state if each hydraulic assembly has been filled and bled. Proper bleeding is critical to the operation of the hydraulic release system. An improperly bled system will not release the cargo hook mechanism.

A reservoir seal is installed beneath the reservoir lid. This seal serves to prevent hydraulic fluid left over from the testing process from leaking during shipping.

⚠ CAUTION

The reservoir seal is for shipping purposes only and must be removed and discarded before bleeding or installation of the hydraulic release system.

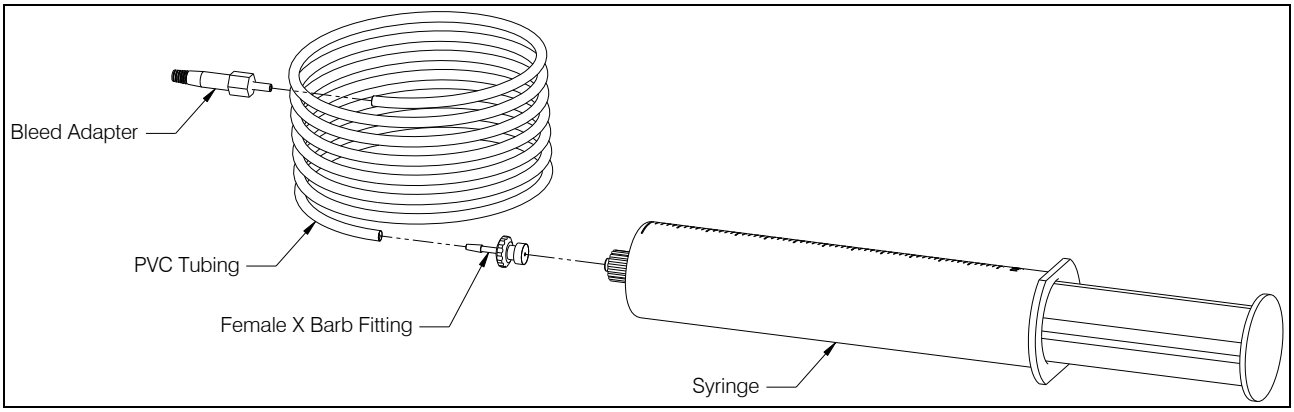
If there is a need to fill and/or bleed the system, follow the procedures listed below. If you need to remove and repair any items in the hydraulic system, refer to 123-011-02, Instruction for Continued Airworthiness.

Filling and bleeding the hydraulic release system is most easily accomplished on the bench, prior to installation on the aircraft. This process may also be accomplished after the system is installed. Filling and bleeding requires two persons, one to inject hydraulic fluid through the system and the other to observe the reservoir.

Bleeding procedure:

1. Obtain the hydraulic hook bleed kit, 212-014-01. This kit consists of 2 ounces of MIL-PRF-5606 fluid, a syringe, a female barb fitting, a length of PVC tubing, and a bleed adapter fitting. The bleed kit is included in new Hydraulic Hook kits. Assemble the bleed kit by press fitting each component together as shown.

Figure 2.7.1 Hydraulic Hook Bleed Kit



2. If the system is already installed on the aircraft, place an absorbent towel under the master cylinder. If the master cylinder is not installed on the aircraft, lightly clamp the master cylinder in a vise to hold it in a vertical position and position the slave cylinder so that its level is below the level of the master cylinder.

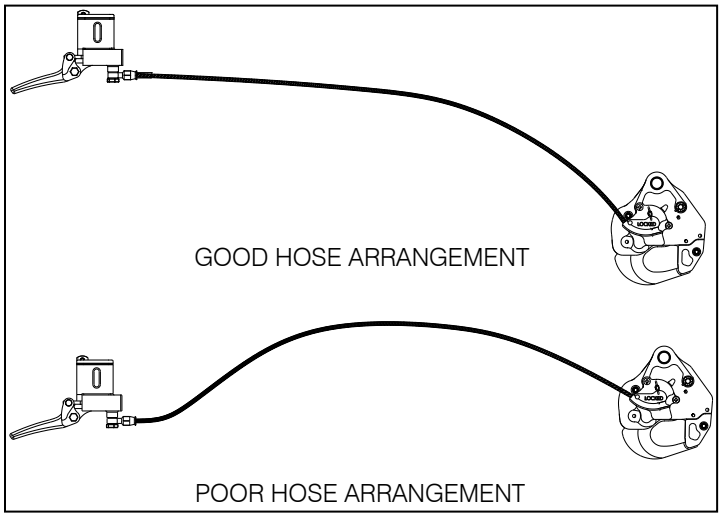
2.7 Filling Hydraulic Release System continued

NOTICE

Use best shop practices to keep foreign material out of the hydraulic system. FOD will plug orifices, damage seals and/or scratch sealing surfaces necessitating system rebuild. Use only clean hydraulic fluid from sealed containers.

3. Connect the master cylinder assembly to the slave cylinder assembly if not already done. If filling or bleeding on the bench, as much as possible, arrange the hoses uncoiled, straight and running uphill as much as possible. See figure 2.7.2.

Figure 2.7.2 Hose Arrangements



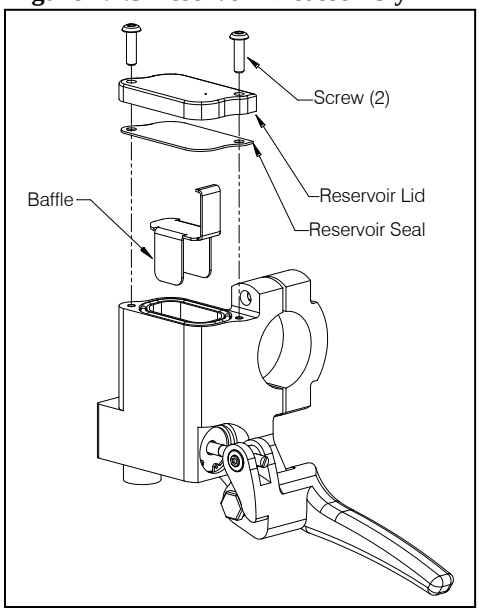
4. Remove screws, reservoir lid, reservoir seal, and baffle from the master cylinder reservoir as shown in Figure 2.7.3 (the reservoir seal is for shipping purposes only, after removal discard or retain for future shipping or storage).

NOTICE

The reservoir seal is for shipping purposes only and must be removed and discarded before bleeding or installation of the hydraulic release system.

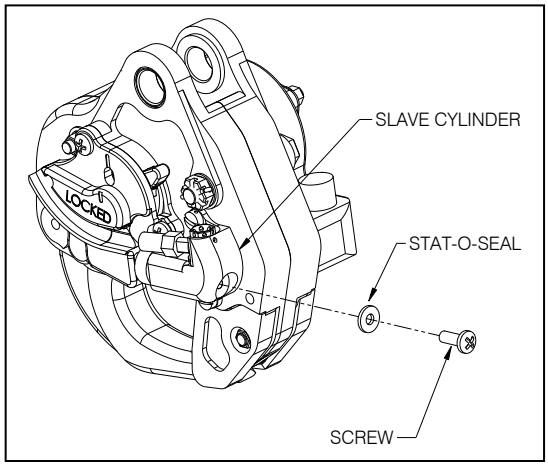
2.7 Filling Hydraulic Release System continued

Figure 2.7.3 Reservoir Disassembly



5. Remove the screw and stat-o-seal on the slave cylinder, see Figure 2.7.4.

Figure 2.7.4 Screw and Stat-o-seal Removal



6. Fill the syringe with approximately 35 cc of MIL-H-5606 fluid and purge any remaining air in the syringe and tubing. Screw the end of the bleed adapter into the screw hole on the slave cylinder to create a tight seal. See Figure 2.7.5.

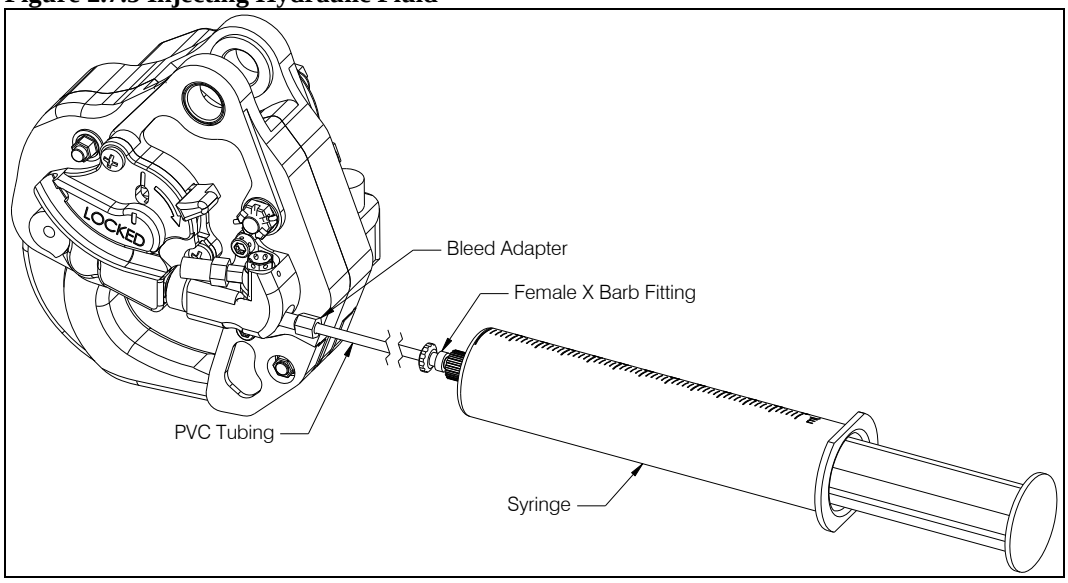
7. While observing the reservoir, **slowly** push on the syringe plunger to force fluid through the slave cylinder, hydraulic hose, and up to the master cylinder reservoir. There will be some resistance during filling—this is normal.

2.7 Filling Hydraulic Release System continued

CAUTION

Injecting the fluid into the system too rapidly may cause the fluid to spray up and out of the master cylinder reservoir. Wear safety glasses when observing fluid reservoir while filling.

Figure 2.7.5 Injecting Hydraulic Fluid



8. Continue to force fluid into the master cylinder reservoir until the reservoir is approximately half full.

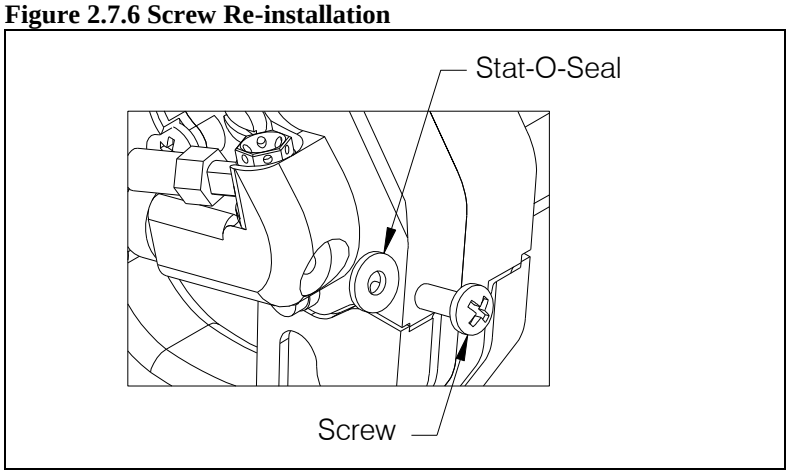
NOTICE

If bleeding an already filled system, you may need to draw fluid from the master cylinder reservoir during this step to prevent overflow.

9. Remove the bleed adapter from the screw hole. Re-install the Stat-O-Seal (P/N 510-496-00) and screw (P/N 510-493-00), see Figure 2.7.6.

2.7 Filling Hydraulic Release System continued

Figure 2.7.6 Screw Re-installation



10. Allow the system to rest for several minutes. This will allow any air to rise through the system.

11. Very **slowly** pull the release lever on the master cylinder and watch for bubbles. If bubbles are observed rising within the reservoir, continue to slowly cycle the lever until there are no more. Actuating the lever releases air trapped within the master cylinder.

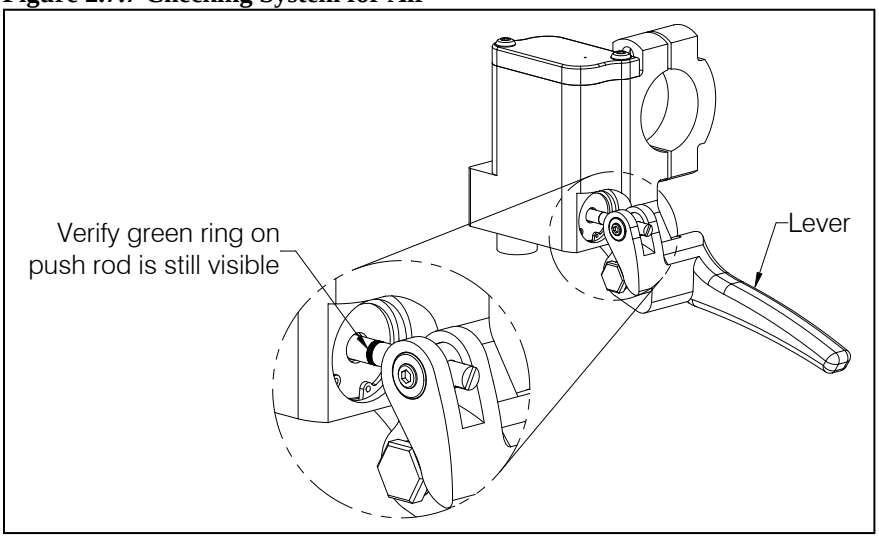
CAUTION

Pull the lever very slowly! When the reservoir is not baffled and capped, a hard pull will cause fluid to erupt over the edge of the reservoir.

2.7 Filling Hydraulic Release System continued

12. Check the system for air by actuating the lever firmly until it bottoms out. Check the push rod position (see Figure 2.7.7). If the green area on the push rod is visible, proceed to step 13. If the green on the push rod is not visible with the lever completely pulled, the system has too much air in it and needs further bleeding. To do this, repeat steps 5 – 11.

Figure 2.7.7 Checking System for Air



13. After the system is properly bled, verify that the reservoir is approximately half full of hydraulic fluid. Fluid should be visible above the baffle.
14. Re-install the baffle, and the reservoir lid.
15. Check the system for proper operation. Fully actuate the release lever. The hook must open and the lever must have a firm feel.
16. Disassemble and thoroughly clean the hydraulic hook bleed kit with isopropyl alcohol. Allow it to dry. Not cleaning the kit will render it unusable. Reassemble and store for next use.

2.8 Placard Installation

Install appropriate load limitation placard, P/N 215-166-00 (3086 lb. max. hook load) or P/N 215-168-00 (2557 lb. max. hook load), dependent on the model of AS350 on which the system is being installed. Consult the Airbus Helicopters Flight Manual Supplement applicable to your particular helicopter for the external load limitation. Locate the placard on the belly of the helicopter, visible to the ground operator and near the hook.

The provided load limitation placards P/N 215-166-00 and P/N 215-168-00 feature black letters on a transparent background. If being installed on a dark-colored helicopter the lettering may not be readily visible. In this instance new placards of the same size and with the same text may be fabricated with a different background to provide visibility. The lettering must contrast with the background.

If installation is being done under a Brazilian CHST, bilingual placards are required. The required placards are illustrated in Section 7, fabricate as shown or contact Onboard Systems for price and availability. Install appropriate load limitation placard, P/N 215-200-00 (1400 kg max. hook load) or P/N 215-201-00 (1160 kg max. hook load), dependent on the model of AS350 on which the system is being installed. Install "SOLTA-CARGA" placard (P/N 215-202-00) on the manual release lever in the cockpit over, or next to, the "CARGO RELEASE" engraving.

2.9 Installation Check-Out

After installation of the Cargo Hook Swing Suspension System, perform the following functional checks.

- Swing the installed cargo hook and suspension to their full extremes to ensure that the hydraulic hose and the electrical harnesses have enough slack to allow full swing without straining or damaging the cables. The hose or harnesses must not be the stops that prevent the cargo hook and suspension from swinging freely in all directions.
- With no load on the cargo hook load beam, pull the release lever on the collective to operate the cargo hook hydraulic release, the Cargo Hook should release. Reset the cargo hook load beam.
- Actuate the release lever on the collective and observe its range of motion. Ensure that it is accessible and that there is no interference throughout its range of motion (refer to Figure 2.9.1). If necessary adjust the lever position by loosening the set screw and turning the lever adjustment screw (ref. Figure 2.9.2) in the required direction. Re-tighten set screw.

Figure 2.9.1 Lever Position

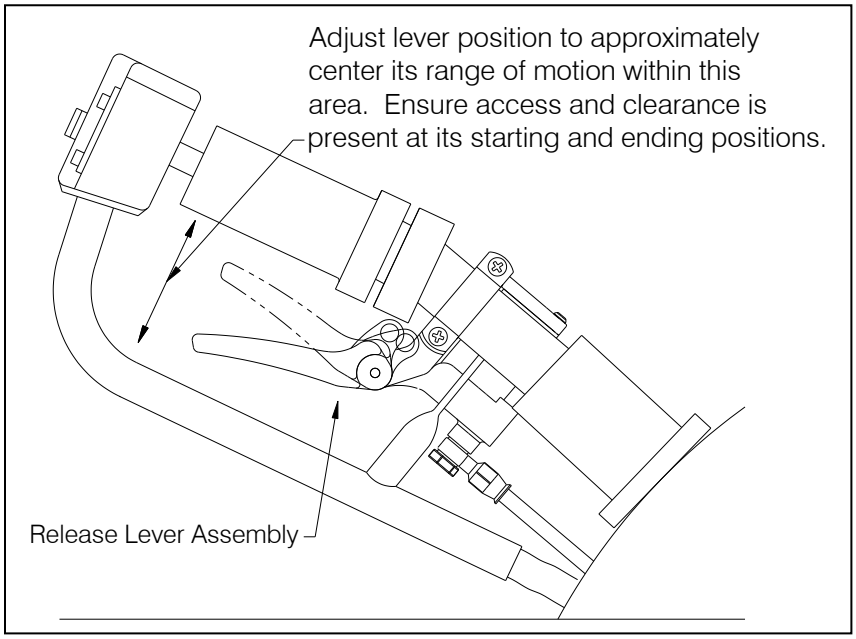
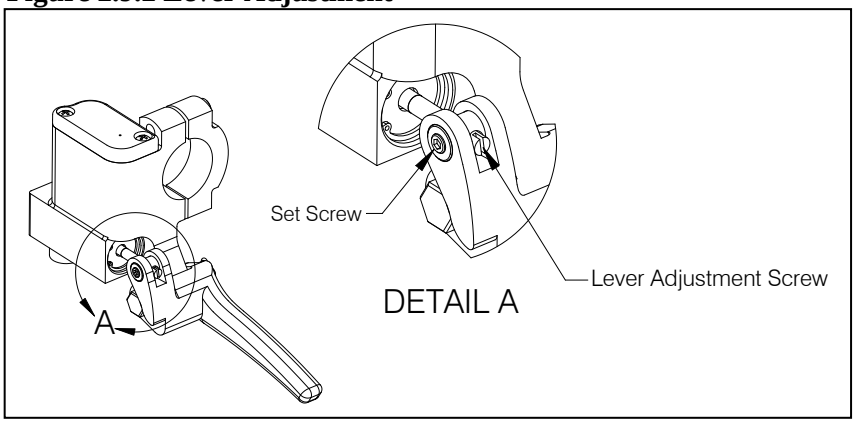

Adjust lever position to approximately center its range of motion within this area. Ensure access and clearance is present at its starting and ending positions.
Release Lever Assembly

Figure 2.9.2 Lever Adjustment



2.9 Installation Check-Out continued

- With no load on the cargo hook load beam, depress the cargo hook electrical release button, the Cargo Hook should release. Reset the cargo hook load beam.
- Perform an EMI ground test per AC 43.13-1b section 11-107. For equipment that can only be checked in flight an EMI flight test may be required.

NOTICE

The cargo hook is of a class of equipment not known to have a high potential for interference. This class of equipment does not require special EMI installation testing (i.e. FADEC) as required in paragraphs 7 and 8 of FAA policy memorandum ASW-2001-02.

- Power on the Indicator and allow it to warm up for 5 minutes (with no load on the hook). Press both Indicator buttons at the same time to go to the Setup Mode. Scroll through the menu until the symbol "0 in" is displayed, then press the right button. Remove any weight that is not to be zeroed out and press either button to complete the procedure.
- If fuel drain guard was installed, pull the handle on the side of the helicopter and verify that fuel is dispensed from the valve.

Component Weights

The weights and cgs of the Cargo Hook Swing Suspension System components are listed below.

Table 2.3 Component Weights

Item	Weight	Station
Removable Provisions*	30.0 lbs (13.6 kg)	133 in (3375 mm)
Fixed Provisions**	5.9 lbs (2.7 kg)	110 in (2794 mm)
Fuel Drain Guard***	0.4 lbs (.18 kg)	135 in. (3430 mm)
Total	36.3 lbs (16.5 kg)	129.4 in (3288 mm)

***** The removable provisions include the swing suspension w/ hook, external manual release cable, and external electrical release cable. These items are easily removed if they are not needed on the helicopter's mission.

****** The fixed provisions are those items of the kit that remain on the aircraft. These include the fixed manual release cable, internal electrical wire harnesses, the load weigh indicator, and the miscellaneous brackets that support these items.

*** Included with newer kits (kits shipped after August 2010).

Paper Work

In the US, fill in FAA form 337 for the initial installation. This procedure may vary in different countries. Make the appropriate aircraft log book entry. Place the Rotorcraft Flight Manual Supplement P/N 121-012-02 in the aircraft Flight Manual.

2.10 Upgrade Kit Installation

These instructions are provided for installing upgrade kit P/N 200-306-00 or P/N 200-306-01 which are used to convert the P/N 200-280-01 to the P/N 200-280-02 or P/N 200-280-03 configuration respectively.

Convert the fixed provisions per the following.

- Remove the internal manual release cable and all supporting hardware.
- Remove existing connector bracket at aft fuel tank support frame.
- Install Connector Bracket (P/N 290-884-00) per Section 2.1 of this manual.
- If installing kit P/N 200-306-01 (for the B3 model) install friction knob (P/N 291-105-00) per Section 2.2 of this manual.
- Install the hydraulic release lever assembly and plumbing (P/N 232-165-00 or P/N 232-165-01) per Section 2.2 of this manual.

At the swing frame assembly perform the following.

- Remove the external manual release cable and all supporting hardware.
- Cut ty-wraps securing electrical harnesses and ground strap at cargo hook bumper and remove the existing cargo hook (P/N 528-023-01) and bumper. Retain the attach bolt (P/N 290-775-00) and hardware.
- Remove the ground strap (P/N 270-126-00) from the existing cargo hook and retain the screw (P/N 510-391-00).
- Prepare cargo hook surface for installation of ground strap by lightly removing the anodize layer around the tapped hole on the new cargo hook (the ground strap is installed in the same location as on P/N 528-023-01). Apply a suitable conductive chemical surface treatment, such as Alodine 1201, to the bare metal.
- Attach the ground strap using the screw removed from previous cargo hook.
- Attach the new electrical release harness (P/N 270-129-00) onto the cargo hook connector.
- Fit the new Bumper (P/N 290-839-02) over the cargo hook.
- Align the Bumper and Cargo Hook with the load cell and assemble attach bolt and hardware removed previously and cotter pin P/N 510-178-00 (refer to ICA 123-011-02 section 25.17 for orientation and hardware).
- Route the load cell harness, slave cylinder plumbing, ground strap, and electrical release harness through the bumper grooves (refer to ICA 123-011-02 section 25.17 for routing).
- Connect the electrical connectors, ground strap connector, and slave cylinder plumbing connector to the mating fittings at the Connector Bracket.
- Bleed the hydraulic system per Section 2.7.

Perform installation check-out per Section 2.9.

Section 3

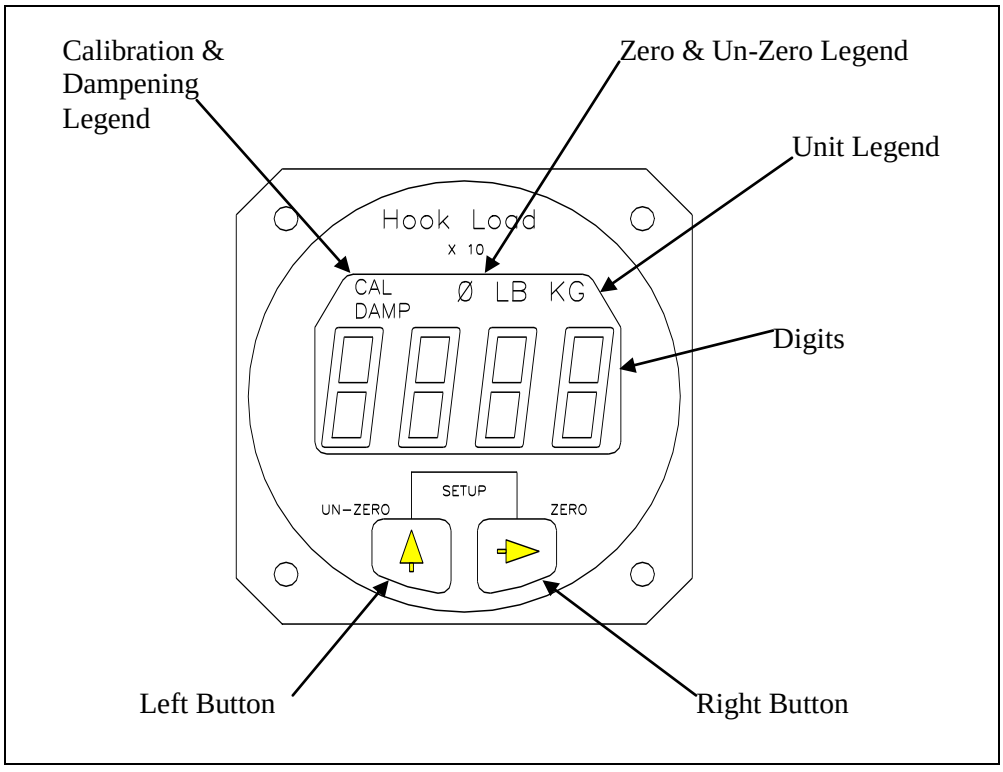
Load Weigh System Operation Instructions

Indicator Front Panel

The C-39 Indicator front panel includes the following features.

- The four 7 segment LCD digits show the weight on the Cargo Hook and display various setup information.
- The Legends clarify the digital display, i.e. - when the LB Legend is turned on, the display will be pounds, etc.
- The right button is used to Zero the display in the Run Mode and select the digit to be changed in the Setup Mode.
- The left button is used to Un-Zero the display in the Run Mode and scroll the selected digit in the Setup Mode.

Figure 3.1 Front Panel

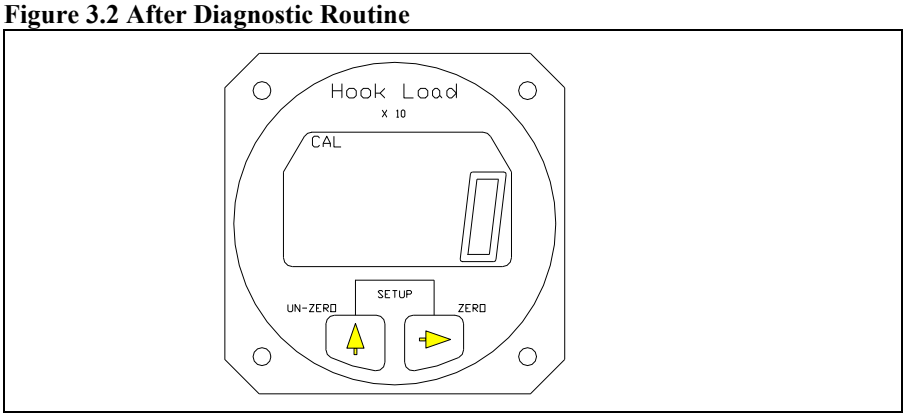


The Run Mode

The C-39 Indicator has two operating modes, Run and Setup. The Run Mode is used to display the cargo hook weight and the Setup Mode is used to setup or configure the Indicator to the helicopter and to the Load Cell. When powered up, the Indicator always comes on in the Run Mode.

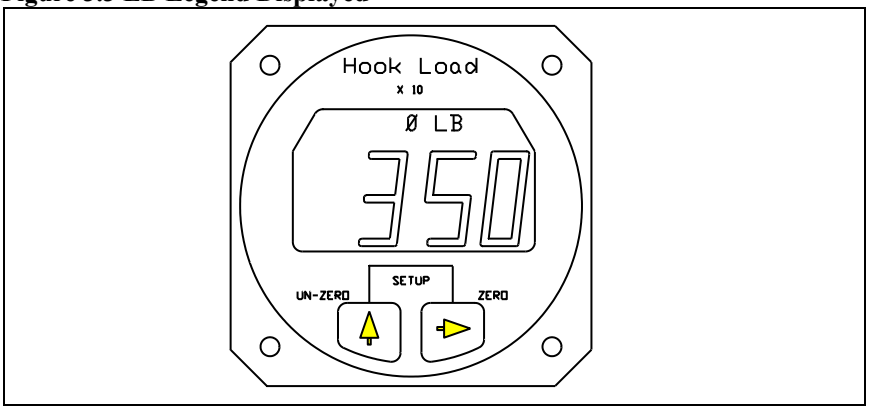
After the Indicator has been correctly installed, power it up by activating the aircraft electrical system. The Indicator will go through a self-diagnostic routine. During this routine the display will display all of the digits and legends. If a problem is found during the routine an Error Code will be displayed. For an explanation of Error Codes see the section *Error Codes*. After the diagnostic routine the display should look like this:

Figure 3.2 After Diagnostic Routine



The illustration is of the Indicator in the Run Mode with no load on the hook. Note the LB legend displayed.

Figure 3.3 LB Legend Displayed

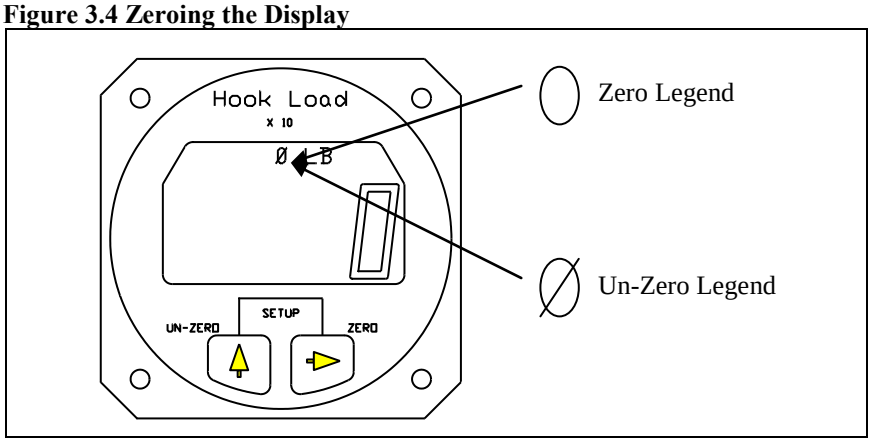


The illustration is a typical hook load reading. The display is 3,500 pounds, note the last digit is not displayed.

To Zero or Tare the Display

The zero feature is used to zero or tare the weight on the Cargo Hook that is not wanted, such as the weight of a cargo net or long line. The Right button is used to zero the Indicator reading. When the Right button is pressed the display is zeroed. The zero legend is turned on and the zeroed number is stored in memory. If the Right button is pressed again, before the Un-zero button is pressed, the display blinks in response to the button closure. Zero is only available in the Run Mode.

Figure 3.4 Zeroing the Display



To Un-Zero the Display

The Left button is used to add the zeroed value back into the current Indicator reading or Un-zero the display. When the Left button is pressed, the number previously zeroed is added to the current display and the Un-zero legend is turned on. If the Left button is again pressed before the zero button is pressed, the display blinks in response to the button closure. Un-Zero is only available in the Run Mode.

Error Codes

Error Codes are the result of difficulties discovered during the Indicator diagnostic tests. Diagnostic tests occur at power up and during the execution of certain routines. Listed below is a matrix of the Error Code displays, their meaning and possible corrective action. Pressing either button will usually bypass the error code, however, the displayed information may be suspect.

Table 3.1 Indicator Error Codes

DISPLAY	CAUSE	POSSIBLE CORRECTIVE ACTION
Err 1	A/D or D/A circuit failure	Potential short in the optional analog meter cable. Clear short and power cycle the Indicator by turning the power to the Indicator off for a few moments. If Error Code continues, return the Indicator to the factory.
Err 2	NV Ram failure	Power cycle the Indicator; if Error Code continues, return the Indicator to the factory.
Err 3	NV Ram write failure	Re-enter data, if Error Code continues, return the Indicator to the factory.
Err 4	NV Ram busy failure	Power cycle the Indicator, if Error Code continues return the Indicator to the factory.

The Setup Mode

The C-39 Indicator can be used with a wide range of helicopters and load cells. The Setup Mode on the Indicator matches the Indicator to the Load Cell and to the helicopter. This is done by entering data into the Indicator. Entered data includes the load cell Calibration Code, the units that the Indicator should read-out (pounds or kilograms), and several other items.

The Indicator has a group of Setup routines, arranged in menu form, that are used to configure the Indicator. Shown on the next page is a matrix of the Setup routines and a brief discussion of their function and how they are programmed. A complete discussion of each setup item is presented later in this section.

To enter the Setup Mode press both the Right and Left buttons at the same time while the Indicator is powered up and in the Run Mode. To exit the Setup Mode and return to the Run Mode, press both the buttons at the same time. If you are in a Setup routine and have started to change an entry, but you change your mind before completing the procedure, power cycle the Indicator to exit the Setup Mode and then go to the Run Mode without changing the item. The Indicator is power cycled by turning the Indicator power off for a few moments.

The Setup Mode, continued

Table 3.2 Indicator Setup Routines

MENU	FUNCTION	DISPLAY
Press the Left button to scroll through the menu	Press the Right button to view or change the menu item.	To return to the Run Mode press both the Right and Left buttons at the same time.
DAMP	Dampening Level, sets the pilots preference for display dampening.	Blinking display is previously entered Dampening Level. Select the desired dampening level by pressing the Left button.
CODE	Calibration Code, matches the Indicator to the Load Cell.	Display is previously entered CAL Code. The Code is changed by selecting the digit to be changed with the Right button. The selected digit will blink. Change the blinking digit by pressing the Left button.
0 in	Installation ZERO, matches the Indicator to the installed Load Cell and to the helicopter. After this procedure the display will be zero when no load is on the Cargo Hook.	Display is a combination of load on the Load Cell, and normal load cell zero offset. Remove all weight from the installed Load Cell except the Cargo Hook, and press any button to complete the procedure and return to the Run Mode.
LOAD	Load, is used to calibrate the system by lifting a known load.	No previous display is shown. Enter the known load using the Right button to select the digit to be changed and Left button to enter the number. Known load is entered "X 10" i.e.; 5000 kilograms is entered as 500. After the known load is entered, press both buttons at the same time and lift the known load. When the load is stabilized press either button. A new Calibration Code will be calculated and the known load will be displayed. This completes the procedure.
Scale	Scale, matches the analog output of the Indicator to an optional remote analog meter.	Display is previously entered number. To change the number use the Right button to select a digit, use the Left button to scroll the digit to the desired number. Entry is times 10.
LB KG	Units, selects the Indicator units (pounds or kilograms).	Display is previously selected unit. To change the unit, use the Left button.
XX - V	Version, is the revision level of the Indicator hardware and software.	Version is for information only, it cannot be changed.

The Setup Mode, continued

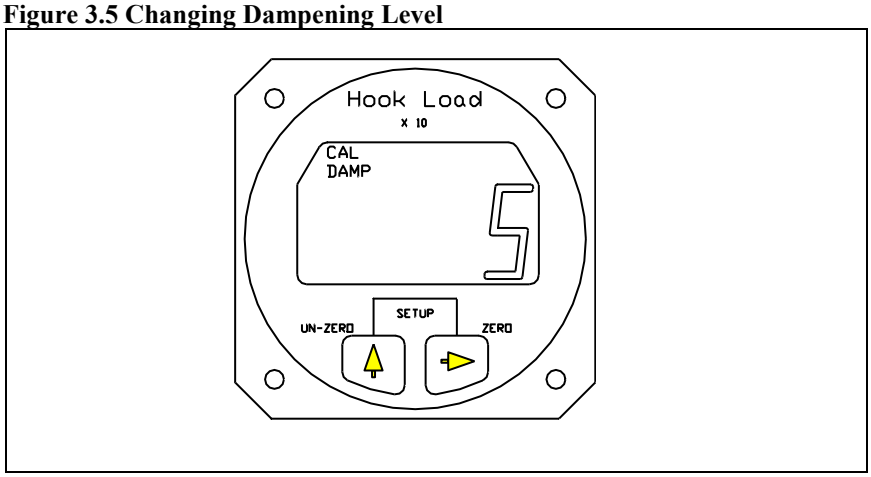
Indicator Dampening

The Damp or dampening routine allows the pilot to adjust the Indicator dampening level to his preference. The dampening routine is a program that stabilizes the Indicator reading. It offers a trade-off between Indicator responsiveness and stability. Ten dampening levels are available, from 0 through 9. At level 0 the display responds to the slightest change in weight. However, if the load bounced even slightly, the display digits would respond instantly, making the display look unstable. With a dampening level of 9, the display would be stable under the most turbulent conditions, however, it would take several seconds for the display to respond to a change in weight. The ideal dampening level will depend on the flying conditions. A mid range setting of 5 or 6 is usually adequate.

To Look at or Change the Dampening Level

With the Indicator powered up and in the Run Mode, press both buttons at the same time to go to Setup. Scroll through the menu, using the Left button, until the word DAMP is displayed. To look at or change the Dampening Level press the Right button. The display should look like this:

Figure 3.5 Changing Dampening Level



The CAL and the DAMP legend is turned on and the previously set dampening level is displayed. To return to Run without changing the current dampening level press both the Right and Left buttons at the same time. To change the dampening number, use the Left button to scroll the blinking digit to the desired number. After the selection has been made press both the Right and Left buttons at the same time to return to Run.

Indicator Calibration

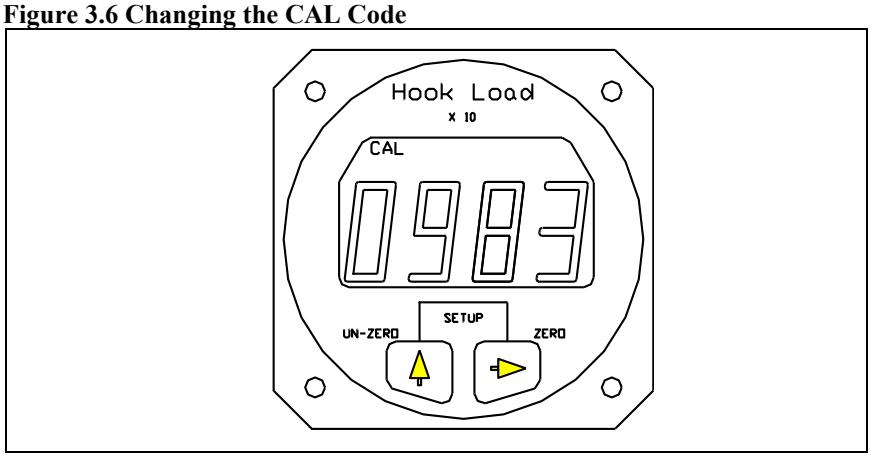
The Calibration Code, or CAL code, is a mandatory input. The Indicator will not accurately display the load without the correct Calibration Code. The Calibration Code scales the signal from the Load Cell.

If the C-39 Indicator was supplied as part of a Load Weigh System, the Calibration Code will have been entered into the Indicator by the factory, however, it should be confirmed. If the Indicator is to be mated to a different Load Cell, it must be calibrated before use. Calibration can be done by entering a known Calibration Code or by lifting a known load and having the Indicator calibrate itself. Both options are discussed below.

To Look at or Change the Calibration Code

With the Indicator powered up and in the Run Mode, press both buttons at the same time to go to Setup. Scroll through the menu until the word CODE is displayed, then press the Right button. The display should look like this:

Figure 3.6 Changing the CAL Code



The CAL legend is turned on and the previously entered or computed Calibration Code is displayed. To return to Run without changing the CAL Code, press both the Right and Left buttons at the same time. To change the Calibration Code, use the Right button to select the digit to be changed, then use the Left button to scroll the blinking digit to the desired number. When the Calibration Code has been entered, press both the Right and Left button at the same time to return to Run.

NOTICE

Depending on the type of Load Cell, the Calibration code could be a 3 or 4 digit number. If the Calibration Code is a 3 digit number a leading zero (0) must be used. For example if a Load Cell had a CAL Code of 395 it would be entered as 0395.

If the load cell Calibration Code is not known or as a cross check, the Indicator can generate the Calibration Code. This is done by entering the weight of a known load into the Indicator LOAD routine and then lifting the load. See the section *Calibration by Lifting a Known Load*.

Installation Zero

Installation zero is a routine that matches the Indicator to the ***INSTALLED*** Load Cell. It adjusts the Indicator reading to compensate for the weight of the Cargo Hook on the Load Cell and whatever zero offset is built into the Load Cell. The Installation Zero procedure is not mandatory. If done the Indicator will read zero when the Un-Zero button is pressed and there is no weight on the Cargo Hook. If the Installation Zero is not done, the Indicator will show the weight of the Cargo Hook plus the value of the Load Cell zero offset.

To Run the Installation Zero Routine

With the Indicator powered up and in the Run Mode, press both buttons at the same time to go to Setup. Scroll through the menu until the symbol "0 in" is displayed, then press the Right button. The CAL legend will be turned on and the current weight on the Cargo Hook will be displayed and blinking. Remove any weight that is not to be zeroed out and press either button to complete the procedure and return to the Run Mode.

Calibration by Lifting a Known Weight

Calibration by lifting a known weight is a Setup routine that calculates the Calibration Code for the Load Cell attached to the Indicator. It is useful if the load cell Calibration Code is not known or as a cross check to the accuracy of a known Calibration Code. The procedure is done by entering the known weight into the Indicator and then lifting the weight. This procedure can be done in the shop or on the helicopter. The accuracy of the procedure is directly related to the weight of the known load. If for example the procedure was done with a 1,000 pound load that was assumed to weigh only 900 pounds, all subsequent lifts would be displayed 10% light.

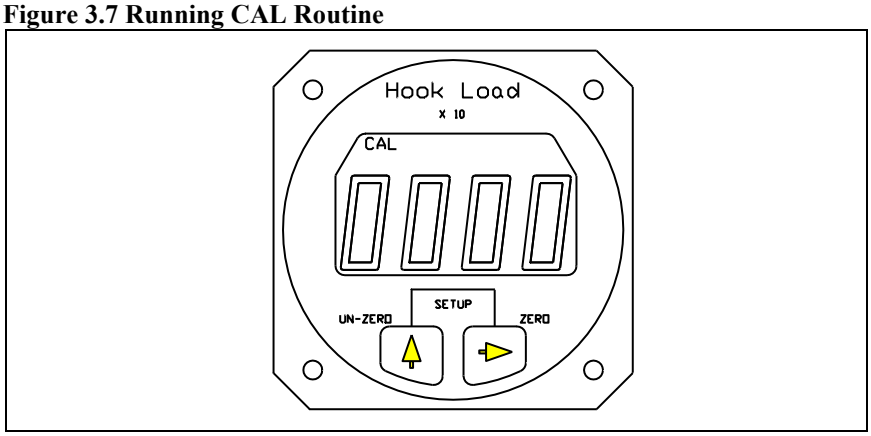
Be sure to include the weight of everything between the Cargo Hook and the load, i.e. the cable, net, dirt, etc.

The closer the known load approaches the lifting capacity of the helicopter, the more accurate the calculated Calibration Code will be.

To Run the Calibration by Lifting a Known Weight Routine

With the Indicator powered up and in the Run Mode, press both buttons at the same time to go to Setup. Scroll through the menu until the word LOAD is displayed, then press the Right button. The display should look like this:

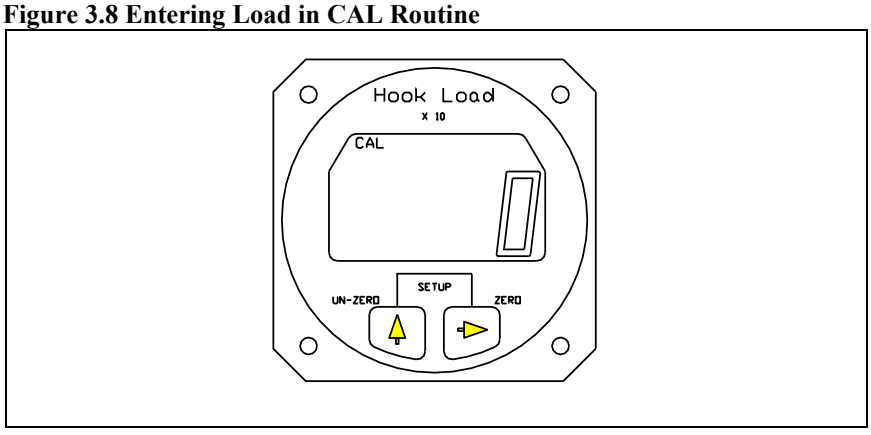
Figure 3.7 Running CAL Routine



The CAL legend is turned on and the first digit is blinking. The previous load is not displayed. At this point if you wish to return to the Run Mode without changing the Calibration Code, power cycle the Indicator. At this point it is not possible to return to the Run Mode without changing the Calibration Code by using the buttons on the Indicator front panel.

To proceed with the procedure, use the Right button to select the digit to be changed, then use the Left button to scroll the blinking digit to the desired number. Note that the known weight is entered "X 10"; a 1000 pound load is entered as 100. When the known load has been entered, press both the Right and Left button at the same time. The display will look like this:

Figure 3.8 Entering Load in CAL Routine



Calibration by Lifting a Known Weight, continued

The CAL legend and the digits will be blinking. Again, at this point if you wish to return to the Run Mode without changing the Calibration Code, power cycle the Indicator. It is not possible to return to the Run Mode by using the buttons on the Indicator front panel without changing the Calibration Code. If you wish to proceed, lift the known load and when it is stabilized, press either button to complete the procedure. The Indicator will display the load. This ends the procedure. The Indicator is now calibrated to the Load Cell. It is a good practice to go to the Code routine and record the new Calibration code for later reference.

Setting the Scale for a remote analog meter

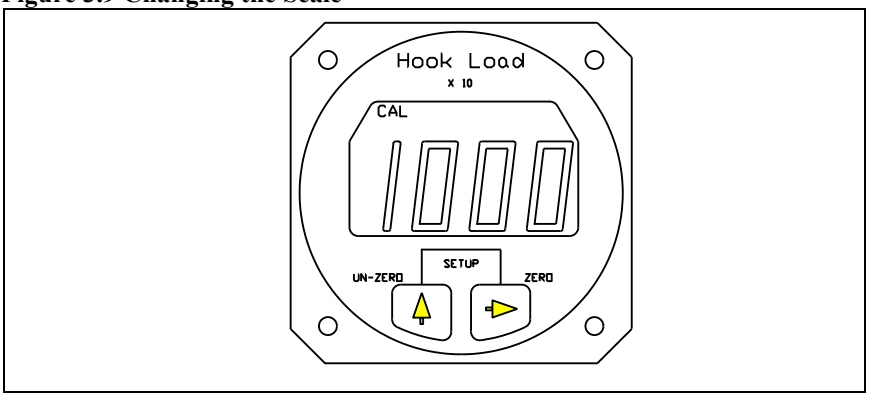
The Scale routine is used when a user supplied analog meter is connected to the Indicator. It is used to match or calibrate the analog meter to the Indicator. The Indicator outputs a 0 to 5 VDC analog signal, which is proportional to the Load Cell load. The Scale number tells the Indicator at what point in pounds or kilograms it should reach the 5 VDC output. If for example a 5 volt analog meter is used and its full scale reading is 10,000 pounds, the number entered into the Indicator Scale routine would be 1000 (the number is entered X 10). This number tells the Indicator that it should output the proportional 0 to 5 VDC signal between zero pounds and 10,000 pounds.

The Scale number does not affect Onboard Slave Meters, P/N 210-106-00 or 210-180-00. This number only affects user supplied instruments connected to the analog out signal.

To Look at or Change the Scale

With the Indicator powered up and in the Run Mode, press both buttons at the same time to go to Setup. Scroll through the menu until the word SCALE is displayed, then press the Right button. The display should look like this:

Figure 3.9 Changing the Scale



To Look at or Change the Scale, continued

The CAL legend is turned on and the previously set Scale number is displayed. To return to Run without changing the Scale, press both the Right and Left button at the same time. To change the Scale number, use the Right button to select a digit to be changed, then use the Left button to scroll the blinking digit to the desired number. When the complete Scale number has been entered, press both the Right and Left button at the same time to return to Run.

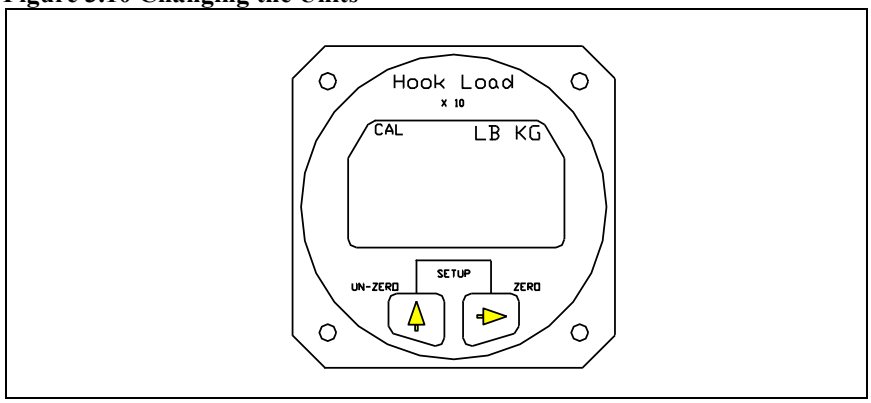
Select KG or LB Units

The units routine sets the display to read in pounds (LB) or kilograms (KG).

To look at or change the Units

With the Indicator powered up and in the Run Mode, press both buttons at the same time to go to Setup. Scroll through the menu until the word LB or KG is displayed, then press the Right button. The display should look like this:

Figure 3.10 Changing the Units



The CAL legend is turned on and the previously set unit is displayed. To return to Run without changing the units, press both the Right and Left button at the same time. To change the units press the Left button. When the selection has been made, press both the Right and Left button at the same time to return to Run.

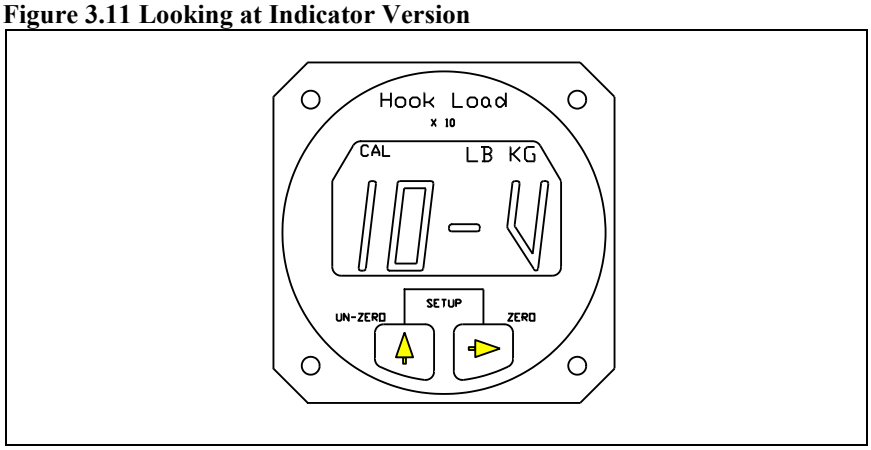
NOTICE

The selected units are displayed when in the Run Mode.

Indicator Version

The Version routine displays the Indicator's hardware and software revision levels. Version is set at the factory and cannot be changed.

Figure 3.11 Looking at Indicator Version



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Section 4

Operation Instructions

Operating Procedures

Prior to a flight involving external load operations perform the following:

1. Ensure that the Cargo Hook Kit has been properly installed and that the hydraulic hose, electrical harnesses and ground strap do not limit the movement of the hook.
2. Be completely familiar with this Owner's Manual, Cargo Hook Service Manual 122-015-00 and the ICA Manual 123-011-02.
3. Activate the electrical system and press the Cargo Hook release button to ensure the cargo hook electrical release is operating correctly. The Cargo Hook must release. Reset the hook by hand after the release. If the hook does not release or re-latch, do not use the unit until the difficulty is resolved.

The release solenoid is intended to be energized only intermittently. Depressing the electrical release button continuously in excess of 20 seconds will cause the release solenoid to overheat, possibly causing permanent damage.

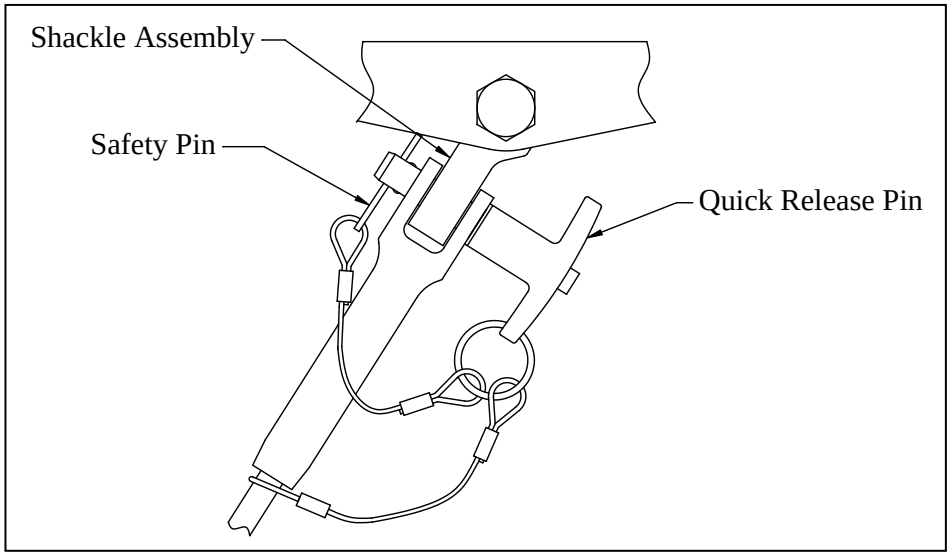
4. Activate the manual release lever on the collective to test the cargo hook manual release mechanism. The Cargo Hook must release. Reset the hook by hand after release. If the hook does not release or re-latch, do not use the unit until the difficulty is resolved.
5. Swing the installed Cargo Hook and the suspension to ensure that the hydraulic hose, electrical harnesses and ground strap have enough slack to allow full swing of each component without straining or damaging the harnesses and hose. The hose and/or harnesses must not be the stops that prevent the Cargo Hook or the suspension from swinging freely in all directions.
6. Visually check for presence and security of fasteners, and condition of cables. Swing the Cargo Hook and the suspension in fore and aft and side to side directions to check for freedom of rotation at all joints.

Disconnecting Removable Provisions

For helicopter missions in which the cargo hook swing suspension system is not needed, its removable provisions may be removed per the following instructions.

1. Disconnect the electrical cables, the ground strap, and the hydraulic release hose at the belly of the helicopter.
2. Remove the Swing Suspension by removing the safety pins and then the quick release pins that secure the cables to each of Shackle Assemblies at the aircraft hard points.

Figure 4.1 Suspension Removal



Re-installing Removable Provisions

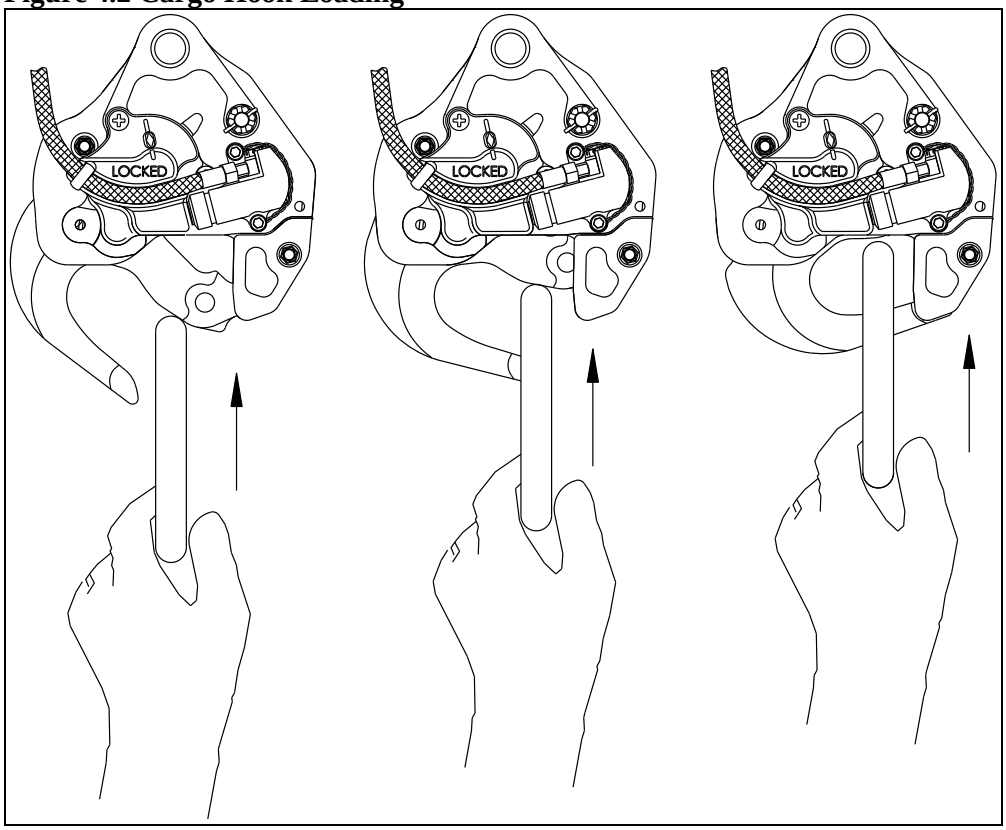
Re-install the removable provisions per the following instructions:

1. Wipe off electrical connectors and the hydraulic quick disconnect fitting to remove any debris.
2. Connect the electrical connector and the hydraulic quick disconnect fitting to the connector bracket.

Cargo Hook Loading

The cargo hook can easily be loaded with one hand. A load is attached to the hook by pushing the ring upward against the upper portion of the load beam throat, as illustrated in Figure 4.2, until an internal latch engages the load beam and latches it in the closed position.

Figure 4.2 Cargo Hook Loading



Cargo Hook Rigging

Extreme care must be exercised when rigging a load to the Cargo Hook. Steel load rings are recommended to provide consistent release performance and resistance to fouling. The following illustration shows the recommended rigging and rigging to avoid, but is not intended to represent all rigging possibilities.

It is the responsibility of the operator to assure the cargo hook will function properly with each rigging.

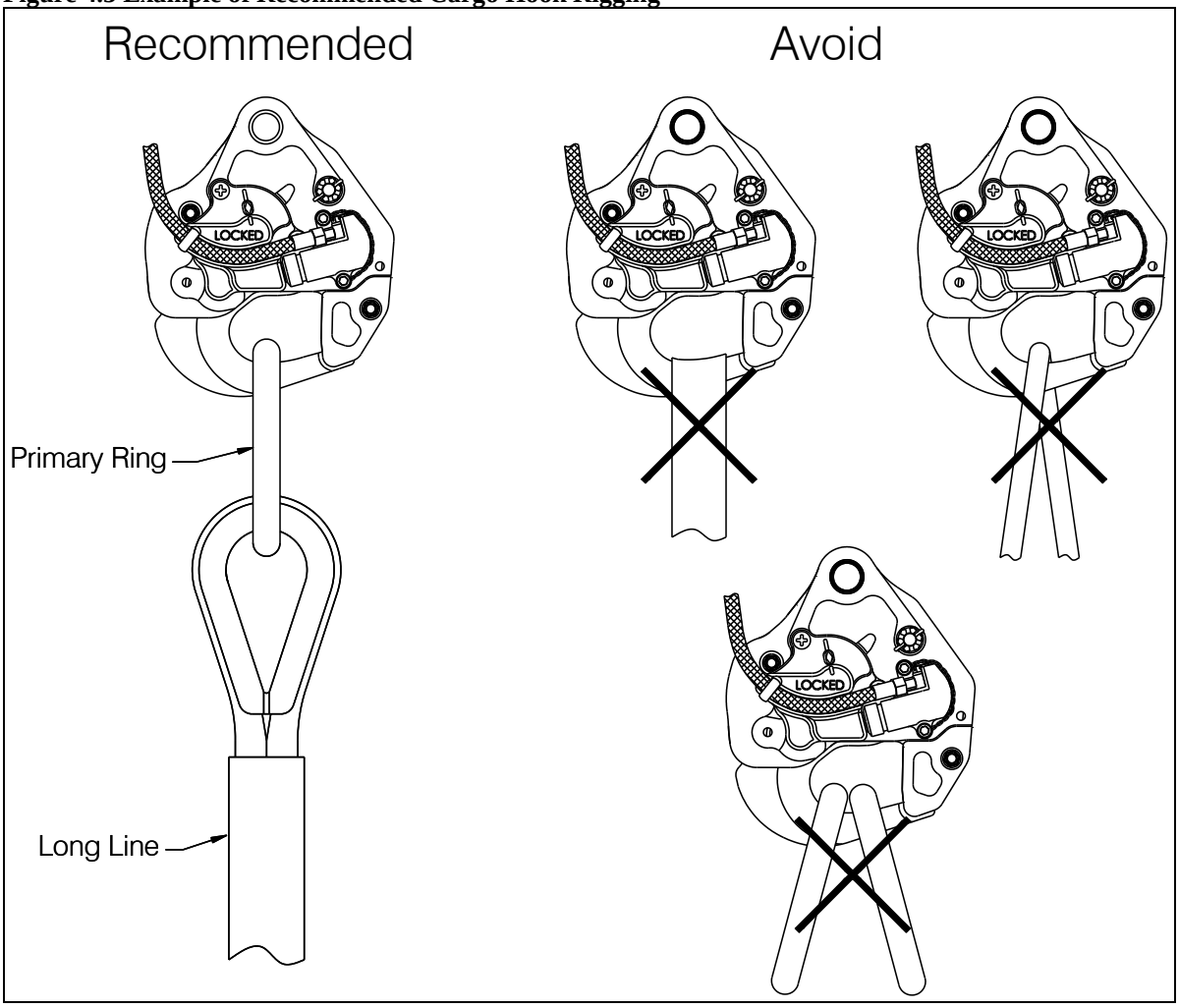
Cargo Hook Rigging, continued

Nylon Type Straps and Rope

WARNING

Nylon type straps (or similar material) or rope must not be used directly on the cargo hook load beam. If nylon straps or rope must be used they should be first attached to a steel primary ring. Verify that the ring will freely slide off the load beam when it is opened. Only the primary ring should be in contact with the cargo hook load beam.

Figure 4.3 Example of Recommended Cargo Hook Rigging



Section 5
Maintenance

Refer to the Instructions for Continued Airworthiness (ICA) manual 123-011-02 for maintenance of the cargo hook suspension system. For maintenance of the cargo hook refer to Cargo Hook Service Manual 122-015-00.

Instructions for Returning Equipment to the Factory

If an Onboard Systems product must be returned to the factory for any reason (including returns, service, repairs, overhaul, etc) obtain an RMA number before shipping your return.

NOTICE

An RMA number is required for all equipment returns.

- To obtain an RMA, please use one of the listed methods.
    - Contact Technical Support by phone or e-mail (Techhelp@OnboardSystems.com).
    - Generate an RMA number at our website: http://www.onboardsystems.com/rma.php
- After you have obtained the RMA number, please be sure to:
    - Package the component carefully to ensure safe transit.
    - Write the RMA number on the outside of the box or on the mailing label.
    - Include the RMA number and reason for the return on your purchase or work order.
    - Include your name, address, phone and fax number and email (as applicable).
    - Return the components freight, cartage, insurance and customs prepaid to:

      Onboard Systems
      13915 NW 3rd Court
      Vancouver, Washington 98685
      USA
      Phone: 360-546-3072

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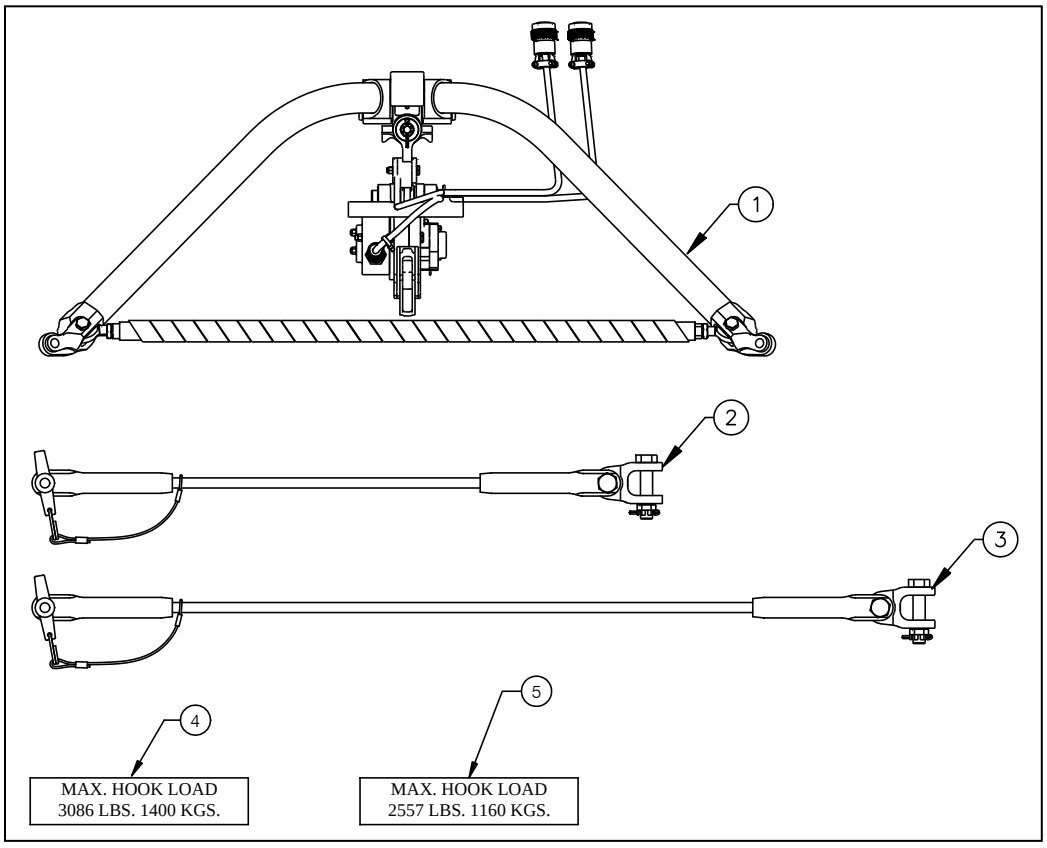
Section 6
System Part Numbers

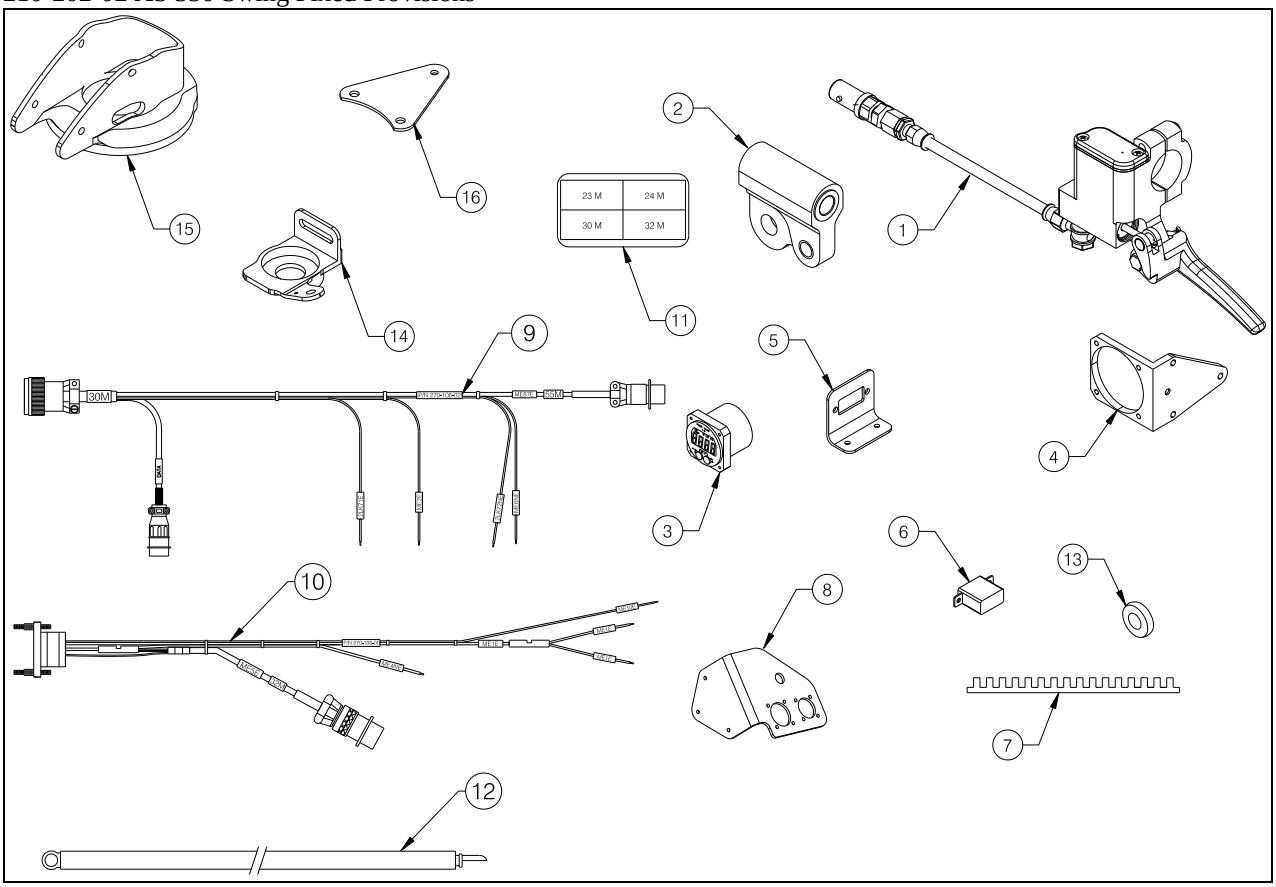
210-201-02 AS 350 Swing Removable Provisions



Item	Part Number	Description	Qty
1	232-145-02	Swing Hook Frame Assembly	1
2	232-140-01	Forward Attach Cable Assembly	2
3	232-141-01	Aft Attach Cable Assembly	2
4	215-166-00	Max Hook Load 3086 Decal	1
5	215-168-00	Max Hook Load 2557 Decal	1

System Part Numbers **continued**

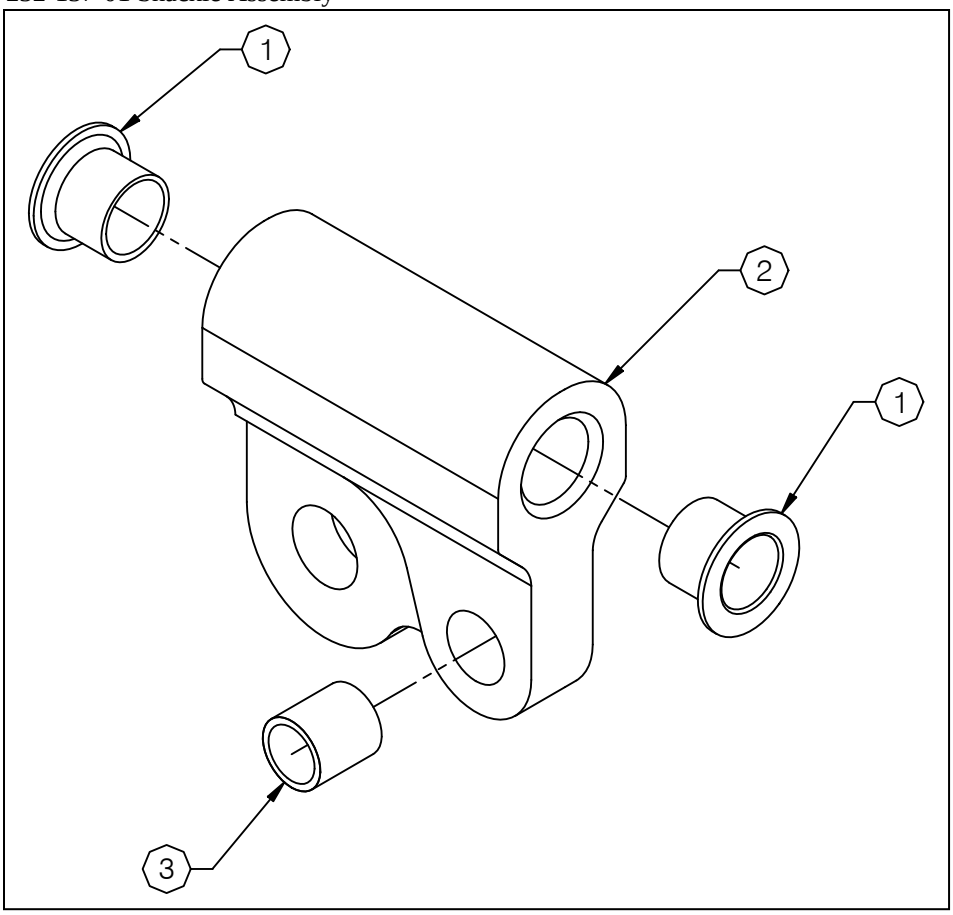
210-202-02 AS 350 Swing Fixed Provisions



ITEM	P/N	DESCRIPTION	QTY.
1	232-165-00	Master Cylinder Assembly	1
2	232-137-01	Shackle Assembly	4
3	210-095-00	C-39 Indicator Assembly	1
4	290-772-00	Indicator Mount Bracket	1
5	290-783-00	Relay Bracket	1
6	445-005-00	Relay	1
7	500-065-00	Grommet Edging	1
8	290-884-00	Connector Bracket	1
9	270-106-02	Load Weigh Internal Harness	1
10	270-108-00	Electrical Release Internal Harness	1
11	215-165-00	Multiple Sticker Sheet	1
12	270-125-00	Ground Strap, Fixed	1
13	505-014-00	Grommet	1
14	290-888-00	Retainer	1
15	290-889-01	Guard	1
16	290-893-00	Bracket	1

System Part Numbers continued

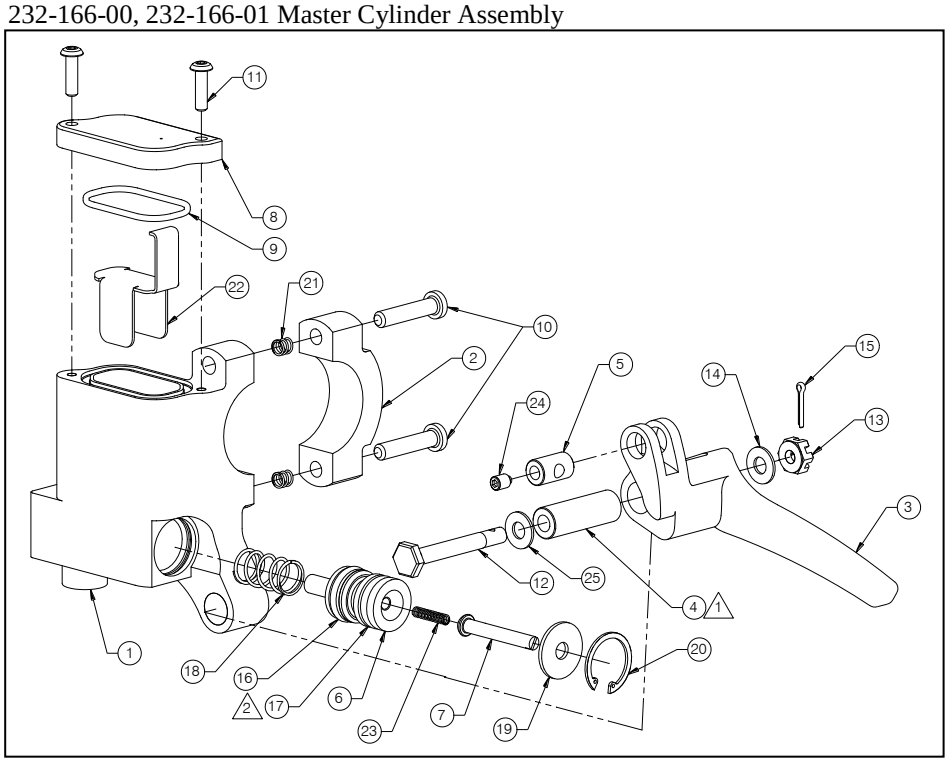
232-137-01 Shackle Assembly



Item	Part Number	Description	Qty
1	517-047-00	Bushing	2
2	290-850-00	Airframe Attach Fitting	1
3	517-016-00	Bushing	1

System Part Numbers continued

232-166-00, 232-166-01 Master Cylinder Assembly

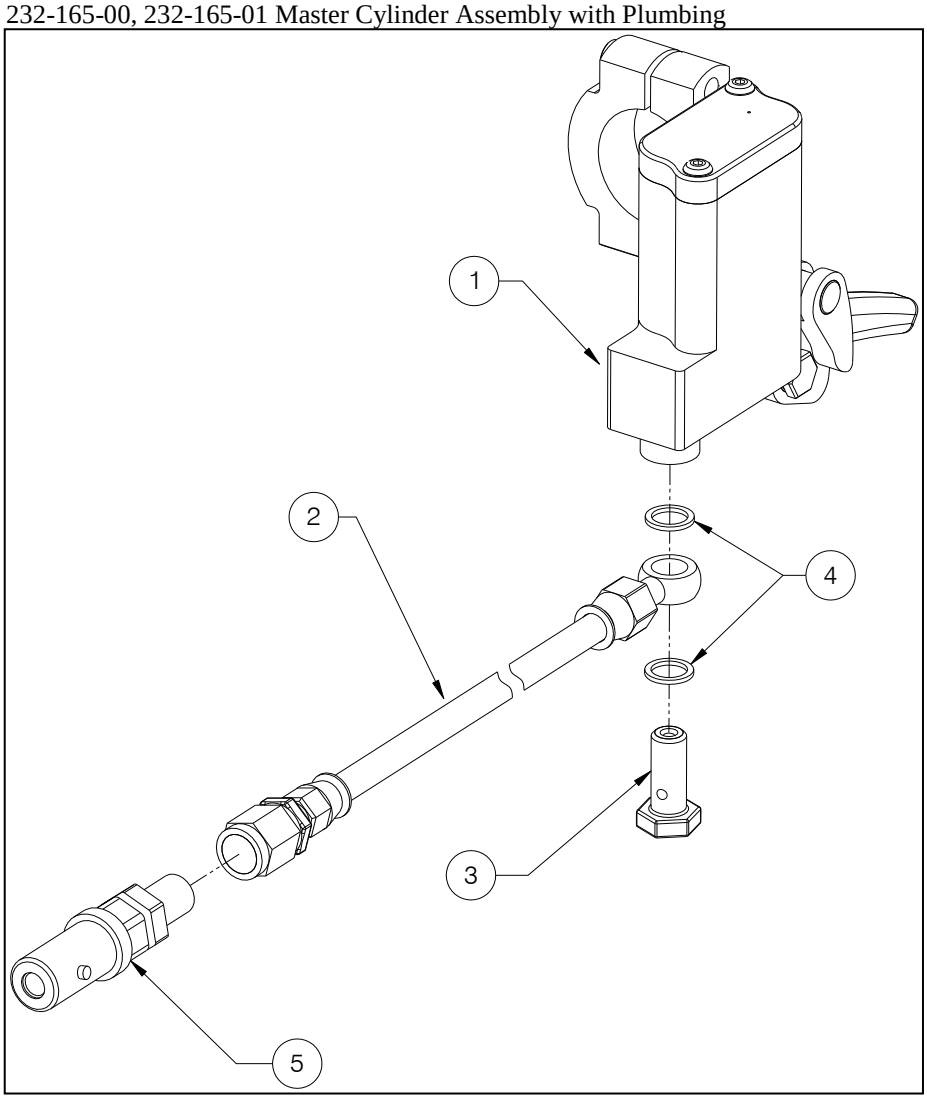


ITEM	P/N	DESCRIPTION	QTY
1	290-810-01	Master Cylinder	1
2	290-753-00	Clamp Half	1
3	290-811-00**	Lever	1
4	290-816-00	Shaft	1
5	290-812-00	Barrel Nut	1
6	290-814-01	Piston	1
7	290-813-00	Push Rod	1
8	290-921-00	Reservoir Lid	1
9	556-044-00	O-Ring	1
10	510-390-00	Screw	2
11	510-157-00	#6-32 x ½" Button Head Cap Screw	2
12	510-487-00	Bolt	1
13	510-082-00	Nut	1
14	510-095-00	Washer	1
15	510-125-00	Cotter Pin	1
16	556-048-00	Cup Seal	1
17	556-047-00	O-Ring	1
18	514-055-00	Compression Spring	1
19	510-532-00	Washer – Piston Stop	1
20	515-008-00	Snap Ring	1
21	510-248-00	Helicoil	2
22	235-118-00	Master Cylinder Baffle	1
23	514-060-00	Compression Spring	1
24	510-530-00	#8-32 x 3/16" Nylon Tip Set Screw	1
25	510-042-00	Washer	1

**** 232-166-01 (for the B3 collective) uses lever P/N 290-811-01.**

System Part Numbers **continued**

232-165-00, 232-165-01 Master Cylinder Assembly with Plumbing

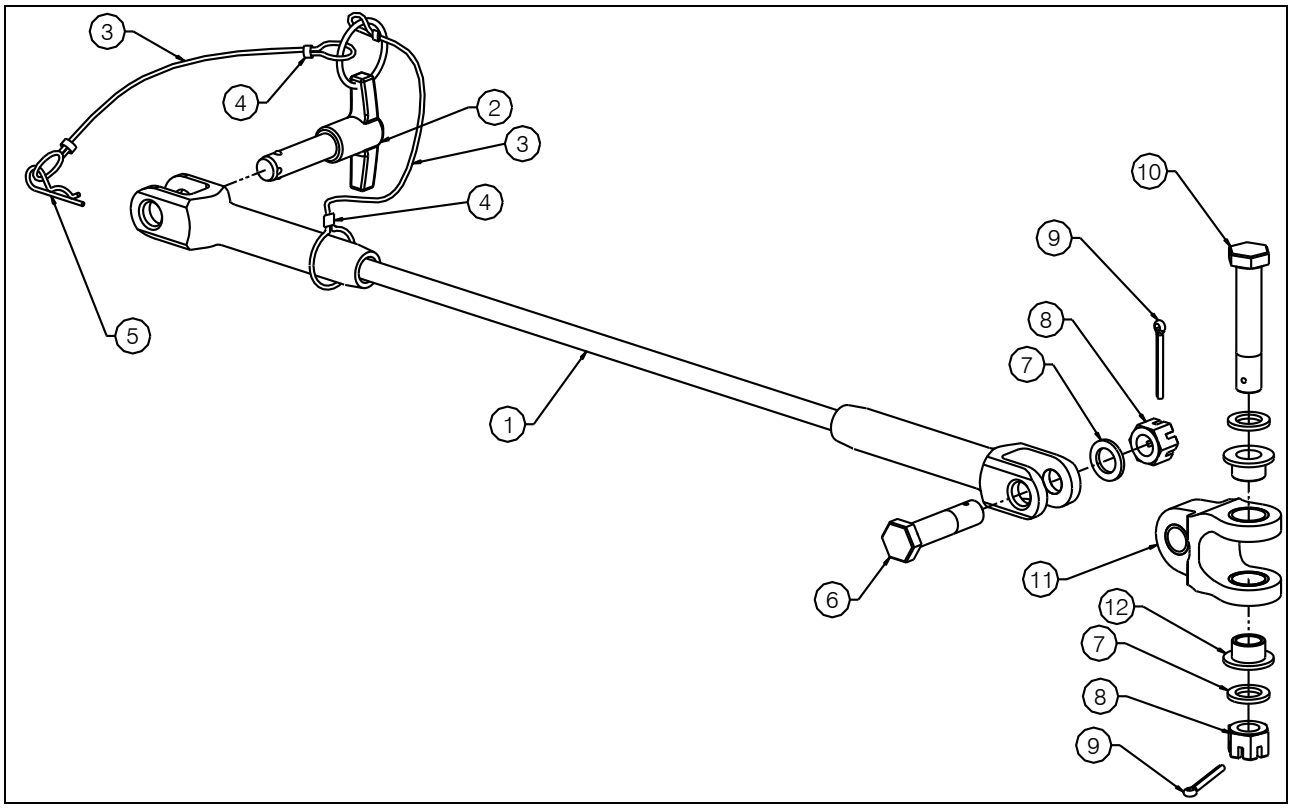


ITEM	P/N	DESCRIPTION	QTY
1**	232-166-00	Master Cylinder Assembly	1
2	232-165-01	Master Cyl Plumbing Assembly	1
3	558-021-00	Banjo Bolt	1
4	556-040-00	Crush Washer	2
5	560-005-00	Quick Disconnect	1

** 232-165-01 (for the B3 collective) uses Master Cylinder Assembly P/N 232-166-01.

System Part Numbers continued

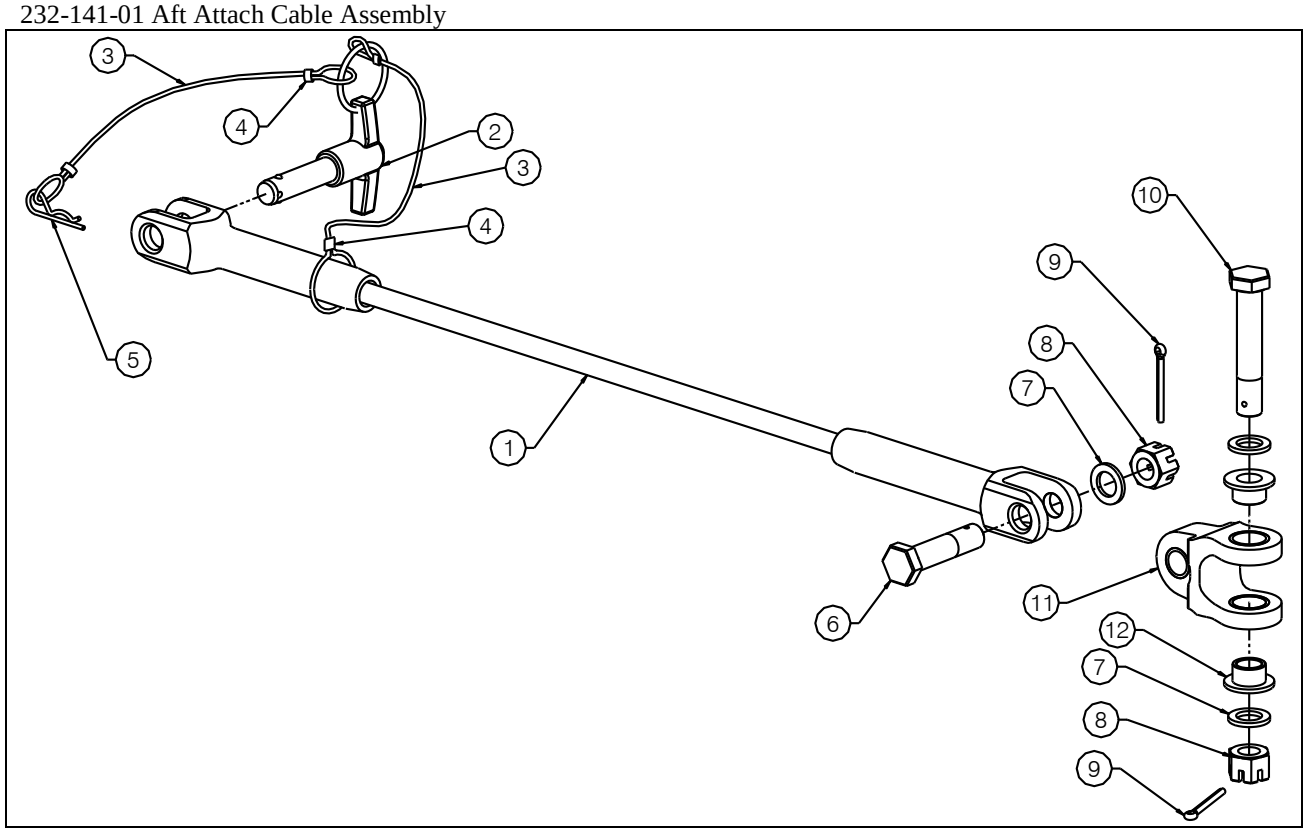
232-140-01 Forward Attach Cable



Item	Part Number	Description	Qty
1	232-177-00	Forward Attach Cable	1
2	290-851-00	Quick Release Pin	1
3	531-015-00	Lanyard Cable	2
4	531-016-00	Crimp Sleeve	2
5	510-464-00	Hitch Pin	1
6	510-438-00	Bolt	1
7	510-221-00	Washer	3
8	510-440-00	3/8" Castellated Nut	2
9	510-178-00	Cotter Pin	2
10	510-439-00	Bolt	1
11	232-142-00	Lower Attach Gimbal Assembly	1
12	290-749-00	Standoff Bushing	2

System Part Numbers continued

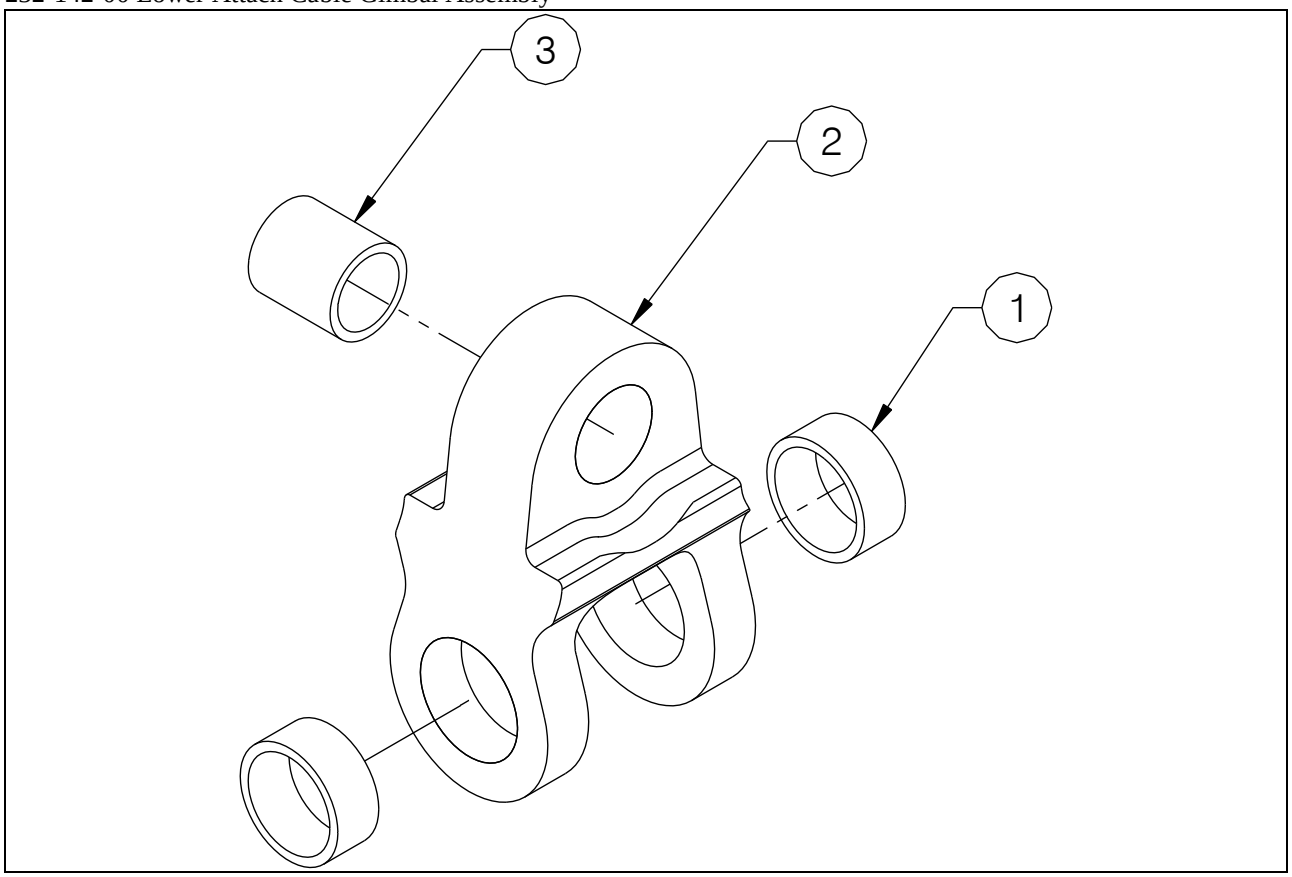
232-141-01 Aft Attach Cable Assembly



Item	Part Number	Description	Qty
1	232-178-00	Aft Attach Cable	1
2	290-851-00	Quick Release Pin	1
3	531-015-00	Lanyard Cable	2
4	531-016-00	Crimp Sleeve	2
5	510-464-00	Hitch Pin	1
6	510-438-00	Bolt	1
7	510-221-00	Washer	3
8	510-440-00	3/8" Castellated Nut	2
9	510-178-00	Cotter Pin	2
10	510-439-00	Bolt	1
11	232-142-00	Lower Attach Gimbal Assembly	1
12	290-749-00	Standoff Bushing	2

System Part Numbers continued

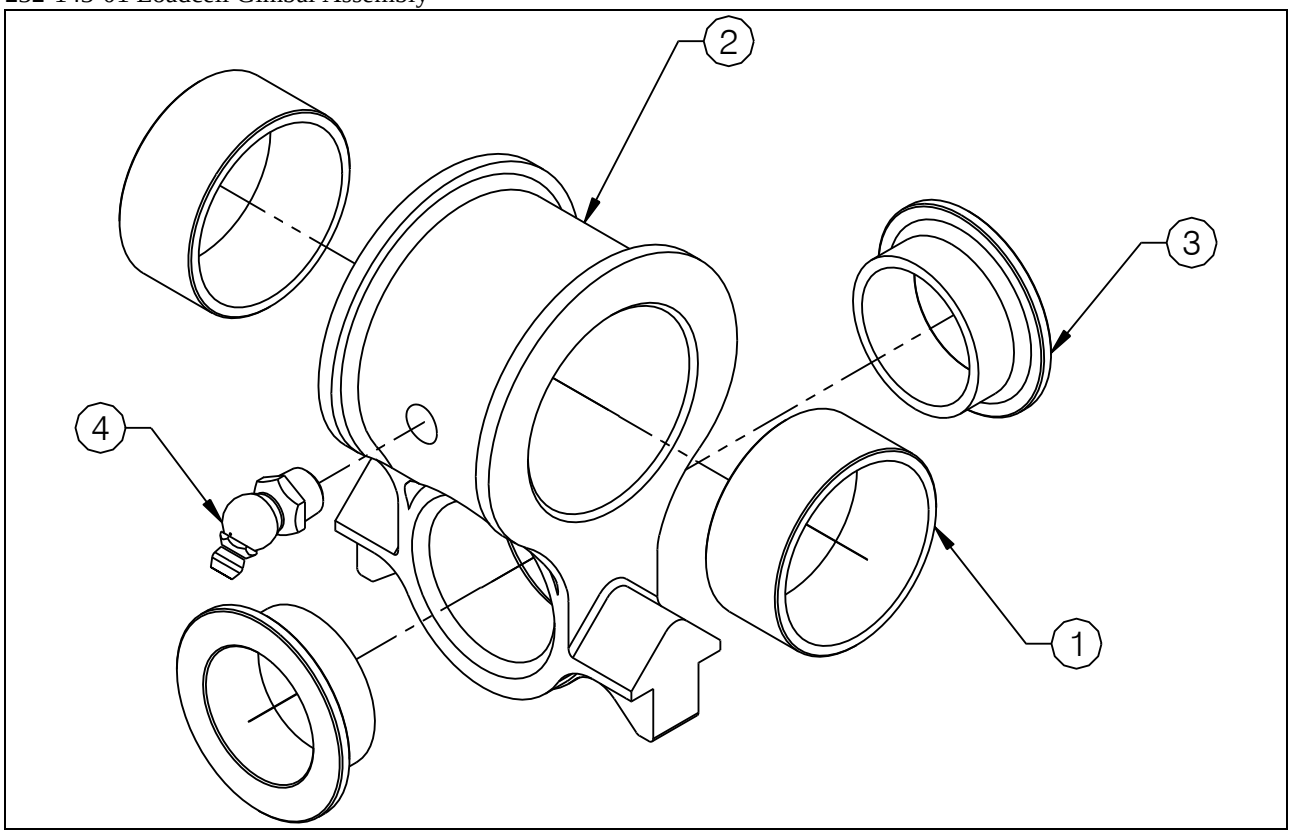
232-142-00 Lower Attach Cable Gimbal Assembly



Item	Part Number	Description	Qty
1	517-048-00	Bushing	2
2	290-746-00	Lower Cable Gimbal	1
3	517-016-00	Bushing	1

System Part Numbers continued

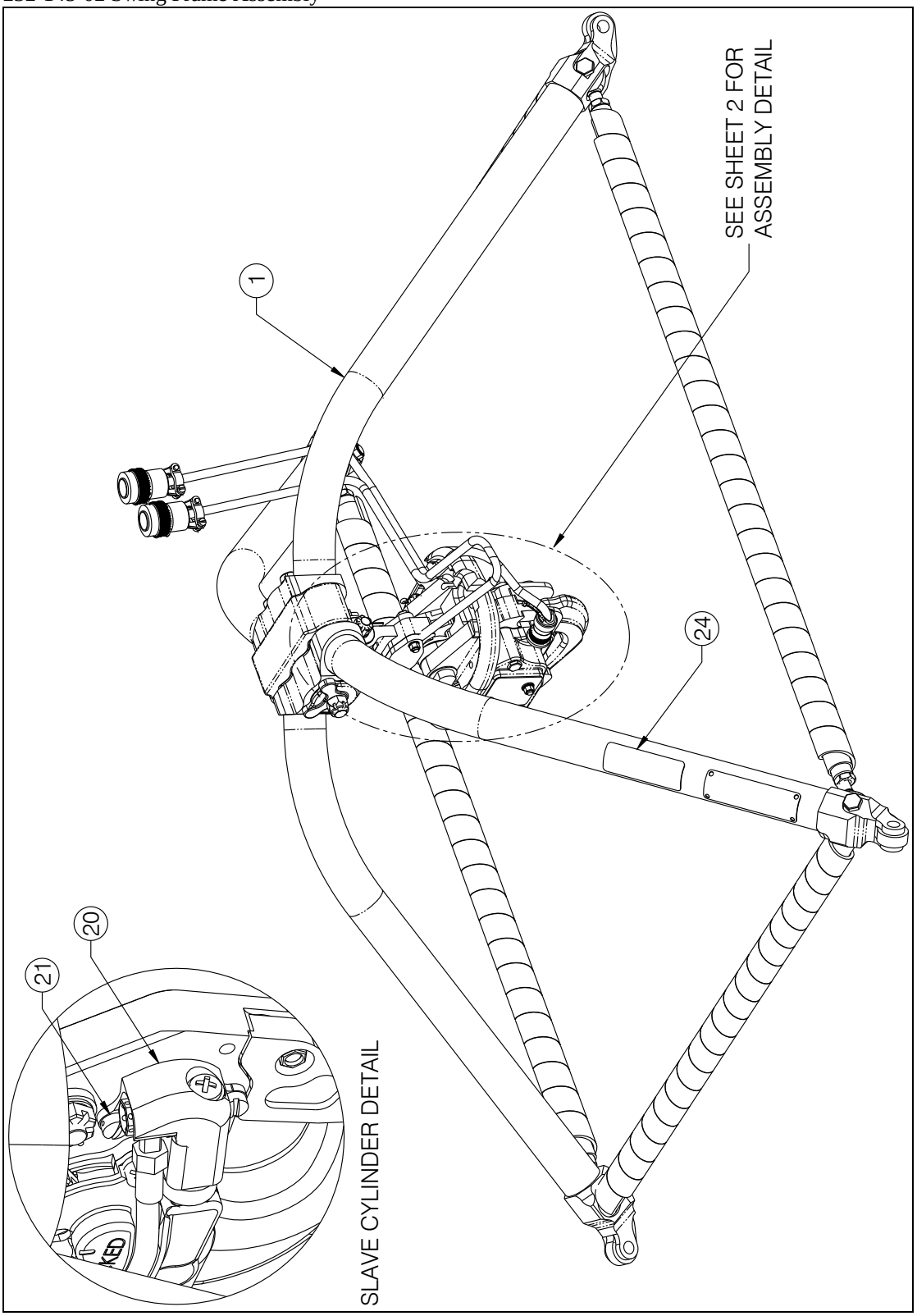
232-143-01 Loadcell Gimbal Assembly



Item	Part Number	Description	Qty
1	517-056-00	Bushing Upper Hook Gimbal	2
2	290-841-00	Gimbal Link	1
3	517-046-00	Bushing Lower Hook Gimbal	2
4	518-003-00	Grease Fitting	1

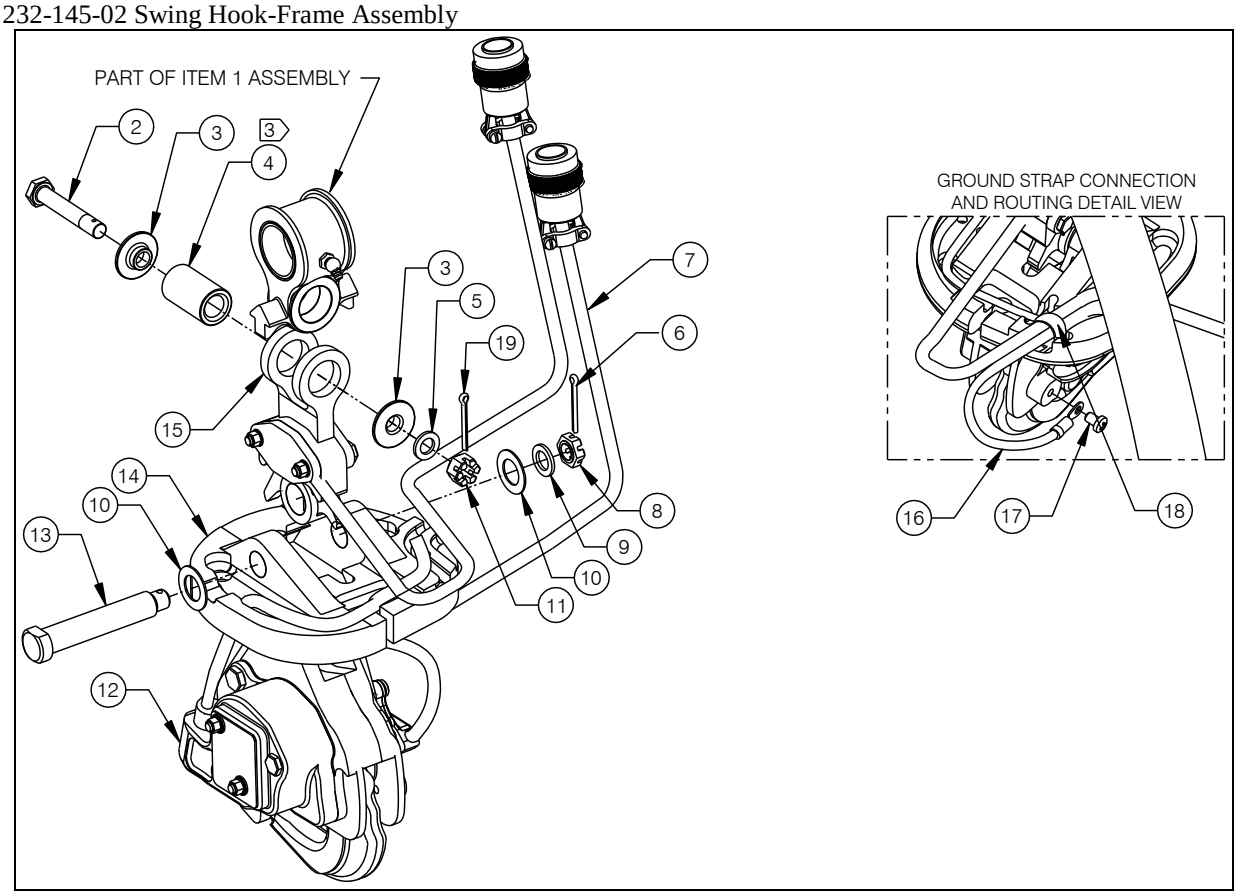
System Part Numbers **continued**

232-145-02 Swing Frame Assembly



System Part Numbers continued

232-145-02 Swing Hook-Frame Assembly



Item	Part Number	Description	Qty
1	232-145-25	Swing Frame Assembly	1
2	510-443-00	Bolt	1
3	290-740-00	Shaft Retaining Bushing	2
4	290-739-00	Shaft – Gimbal	1
5	510-220-00	Washer	1
6	510-178-00	Cotter Pin	1
7	270-129-00	Lower Electrical Release Cable	1
8	510-170-00	Nut	1
9	510-174-00	Washer	1
10	510-183-00	Washer	2
11	510-320-00	Nut, Castellated	1
12	528-028-00	Talon LC Hydraulic Cargo Hook	1
13	290-775-00	Long Hook Attach Bolt	1
14	290-839-02**	Hook Bumper	1
15	210-249-03*	Swing Load Cell Assembly	1
16	270-126-00	Ground Strap	1

* Supersedes P/N 210-243-00 & P/N 210-199-01 and P/N 210-199-00. These assemblies are interchangeable.

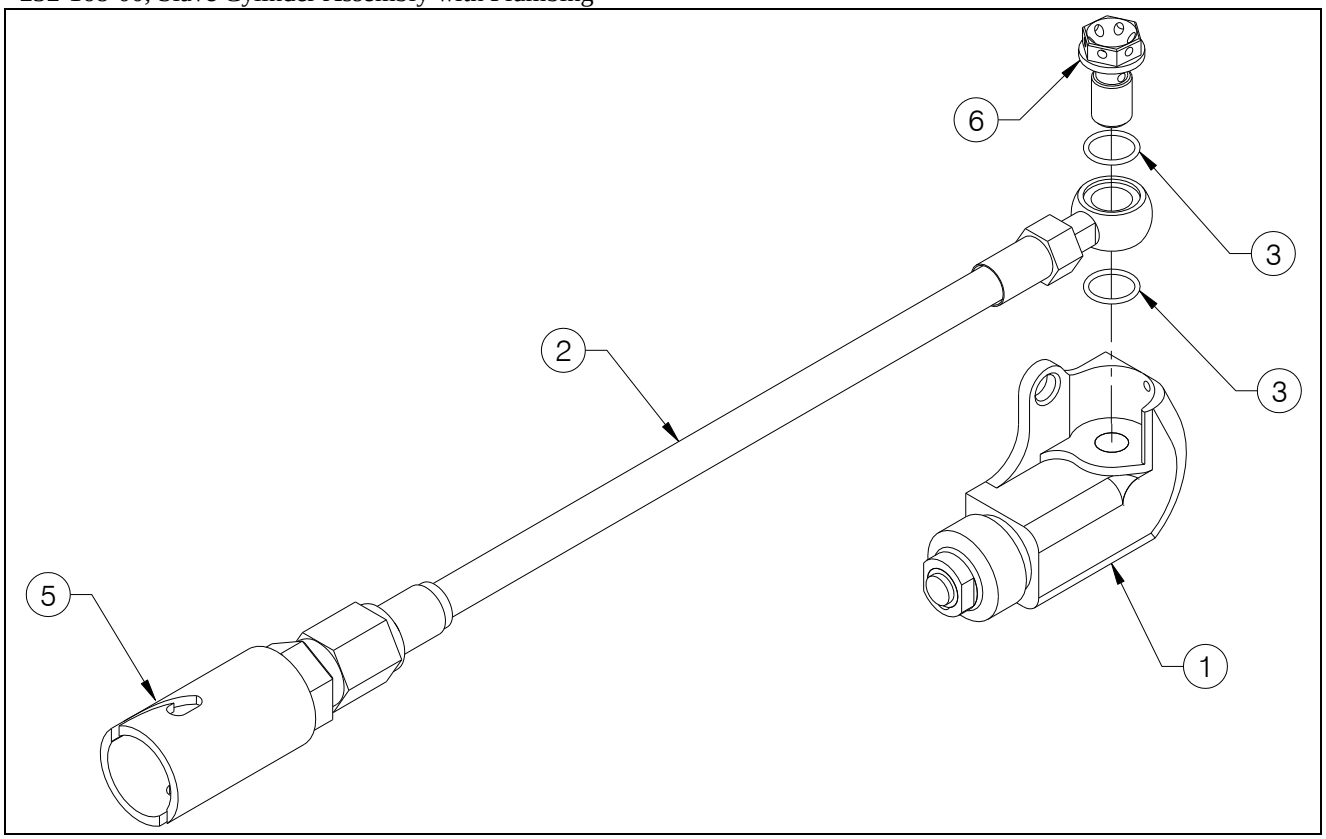
** Supersedes 290-839-01 and 290-839-00. These parts are interchangeable.

232-145-02 Swing Hook-Frame Assembly, continued

Item	Part Number	Description	Qty
17	510-391-00	Screw	1
18	512-011-00	Ty-Wrap	4
19	510-115-00	Cotter Pin	1
20	232-168-00	Slave Cylinder Assembly	1
21	510-531-00	Screw	2
22	420-033-00	Safety Wire	AR
23	512-003-00	Ty-Wrap	1
24	215-271-00	Fuel Drain Warning Placard	1

System Part Numbers continued

232-168-00, Slave Cylinder Assembly with Plumbing

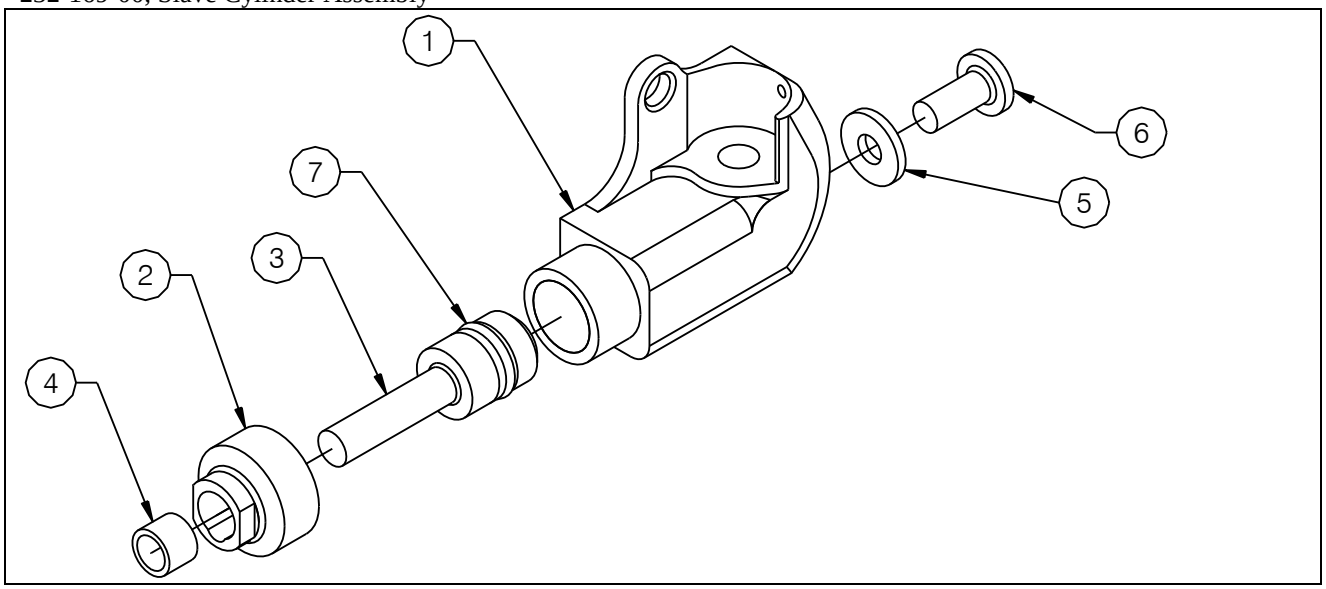


Item	Part Number	Description	Qty
1	232-169-00	Slave Cylinder Assembly	1
2	232-170-02*	Slave Cylinder Plumbing Assy	1
3	556-041-00	O-Ring	2
4	558-031-00	Banjo Bolt	1
5	560-006-00	Quick Disconnect	1

*This item supersedes P/N 232-170-00 & 232-170-01. These parts are fully interchangeable.

System Part Numbers continued

232-169-00, Slave Cylinder Assembly



Item	Part Number	Description	Qty
1	290-803-00	Slave Cylinder	1
2	290-802-00	Cylinder Cap	1
3	290-805-00	Piston	1
4	517-040-00	Bushing	1
5	510-496-00	Stat-O-Seal	1
6	510-493-00	Screw	1
7	556-097-00	Quad Ring	1

Section 7
Certification

FAA STC

United States of America
Department of Transportation Federal Aviation Administration

Supplemental Type Certificate

Number **SR01164SE**

This certificate, issued to: **Onboard Systems International**
13915 NW 3rd Court
Vancouver, WA 98685

certifies that the change in the type design for the following product with the limitations and conditions therefore as specified hereon meets the airworthiness requirements of Part **27** *of the* **Federal Aviation Regulations.**

Original Product—Type Certificate Number: H9EU
Make: Eurocopter France
Model: AS350B, AS350B1, AS350B2, AS350B3, AS350BA, and AS350D

Description of the Type Design Change: Fabrication of Onboard Systems Model 200-280-01, 200-280-02, 200-280-03, or 200-280-04, Cargo Hook Swing Suspension System in accordance with FAA-approved Onboard Systems Master Drawing List No. 155-086-00, Revision 27, dated October 30, 2009, or later FAA-approved revision.

Installation of the systems in accordance with FAA-approved Onboard Systems Owner's Manual No. listed in the table below, or later FAA-approved revision. This modification must be inspected and maintained in accordance with Section 5 of the FAA-approved Onboard Systems Instructions for Continued Airworthiness (ICA) and Onboard Systems Cargo Hook Service Manual listed in the table below, or later FAA-approved revision. (See Continuation Sheet on Page 3)

Limitations and Conditions: Approval of this change in type design applies to the Eurocopter AS350 model rotorcraft listed above. This approval should not be extended to other rotorcraft of these models on which other previously approved modifications are incorporated unless it is determined by the installer that the relationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that rotorcraft. Rotorcraft modified in accordance with this STC must be operated in accordance with a copy of a FAA-approved Onboard Systems Rotorcraft Flight Manual Supplement (RFMS) listed in the table below, or later FAA-approved revision. (See Continuation Sheet on Page 3)

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: March 29, 2002 *Date reissued:*
Date of issuance: January 22, 2003 *Date amended:* October 26, 2004; February 04, 2005; October 30, 2007; April 13, 2010

By direction of the Administrator

(Signature)
Acting Manager, Seattle Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding $1,000, or imprisonment not exceeding 3 years, or both.
This certificate may be transferred in accordance with FAR 21.47.

FAA Form 8110-2 (10-68) PAGE 1 OF 3 PAGES

FAA STC continued

United States of America
Department of Transportation—Federal Aviation Administration

Supplemental Type Certificate

(Continuation Sheet)

Number **SR01164SE**

Onboard Systems International

Issued: January 22, 2003

Amended: October 26, 2004; February 04, 2005; October 30, 2007; April 13, 2010

Description of the Type Design Change continued:

SYSTEM PART NUMBER (P/N)	OWNER'S MANUAL NO.	INSTRUCTIONS FOR CONTINUED AIRWORTHINESS AND CARGO HOOK SERVICE MANUAL:
200-280-01	120-104-01 Revision 1, dated September 29, 2004	123-011-01 Revision 0, dated August 11, 2004 122-005-00 Revision 15, dated April 23, 2009
200-280-02	120-104-02 Revision 10, dated May 23, 2007	123-011-02 Revision 5, dated January 27, 2009 122-015-00 Revision 7, dated January 30, 2009
200-280-03	120-104-02 Revision 10, dated May 23, 2007	123-011-02 Revision 5, dated January 27, 2009 122-015-00 Revision 7, dated January 30, 2009
200-280-04	120-104-03 Revision 0, dated October 12, 2009	123-011-03 Revision 0, dated January 27, 2010 122-017-00 Revision 5, dated July 27, 2009

Limitations and Conditions continued:

SYSTEM PART NUMBER	ROTORCRAFT FLIGHT MANUAL SUPPLEMENT
200-280-01	121-012-01, Revision 2, dated June 12, 2007
200-280-02	121-012-02, Revision 3, dated October 25, 2007
200-280-03	121-012-02, Revision 3, dated October 25, 2007
200-280-04	121-012-03, Revision 0, dated March 4, 2010

A copy of this certificate, FAA-approved RFMS, ICA, and Service Manual must be maintained as part of the permanent records of the modified rotorcraft.

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

- END -

Any alteration of this certificate is punishable by a fine of not exceeding $1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

FAA FORM 8110-2-1 (10-69) PAGE 3 OF 3 PAGES

Canadian Approval

Transport Canada
Aviation

Transports Canada
Aviation

Suite 620
800 Burrard Street
Vancouver, B.C.
V6Z 2J8

Your file *Votre référence*
102S-GA-05-98
Our file *Notre référence*
P-05-0225

May 26, 2005

Mr. Mark Hanson
Onboard Systems
13915 NW 3rd Court
Vancouver, WA
98685 USA

Subject: Acceptance of revised FAA STC SR01164SE

Dear Mr. Hanson,

This is in response to the FAA Seattle ACO letter dated April 29, 2005, requesting Transport Canada approval of the subject STC.

In accordance with our current policy associated with the review of foreign STCs, some STCs applicable to certain categories of aircraft may be accepted solely on the basis of their foreign certification, and do not require the issue of a corresponding certificate by Transport Canada. The subject STC falls within these criteria.

This STC will be entered in the national index of STCs that have been reviewed and accepted by Transport Canada for installation on Canadian registered aeronautical products.

This letter confirms formal acceptance of the referenced STC by Transport Canada.

Yours truly,

Henry Wong
Aircraft Certification Engineer

c.c. Mr. Jeffrey E. Duven, Manager Seattle Aircraft Certification Office

Canada

1/1

Brazilian CHST

ANAC

AGÊNCIA NACIONAL DE AVIAÇÃO CIVIL - BRASIL

CERTIFICADO DE HOMOLOGAÇÃO SUPLEMENTAR DE TIPO

(Supplemental Type Certificate)

NÚMERO (Number) **2007S04-12**

Este certificado, emitido com base na Lei nº 7565 "Código Brasileiro de Aeronáutica", de 19 de dezembro de 1986,
(This certificate, issued in the basis of the Law No. 7565 "Código Brasileiro de Aeronáutica", dated 19 December 1986,

é conferido ao (à): (is granted to:) Onboard Systems
13915 NW 3rd Court
Vancouver, WA 98685
USA

por ter a modificação ao projeto de tipo do produto abaixo citado, observadas as limitações e condições
(for having the change to the type design of the product mentioned below, with the limitations and conditions therefor as)

especificadas, satisfeito aos requisitos de aeronavegabilidade aplicáveis.
(specified hereon, met the applicable airworthiness requirements.)

Produto Original - Número do Certificado de Tipo: (Original Product – Type Certificate No:) 8812 (ANAC).

Fabricante: (Manufacturer:) Eurocopter France.

Modelo(s): (Model(s):) AS 350 BA, AS 350 B1, AS 350 B2 and AS 350 B3.

DESCRIÇÃO DA MODIFICAÇÃO AO PROJETO DE TIPO:
(Description of Type Design Change:)

Installation of Onboard Systems Model 200-280-01 or 200-280-02 Cargo Hook Swing Suspension System in accordance with Onboard Systems Owner's Manual No. 120-104-01, Rev. 5, dated 3 Apr. 2007 or Owner's Manual No. 120-104-02, Rev. 9, dated 3 Apr. 2007, or later approved revisions.
This CHST validates in Brazil the STC # SR01164SE, issued by FAA (USA).

LIMITAÇÕES E CONDIÇÕES:
(Limitations and Conditions:)

See continuation sheet for applicable data.

DATAS:
(Dates of:)

Do Requerimento: (Application:) 21 Dec. 2006 **Da emissão**: (Issue:) 13 Apr. 2007 **Da reemissão**: (Reissue:)

CLÁUDIO PASSOS SIMÃO
Gerente Geral, Certificação de Produtos Aeronáuticos
(Manager, Aeronautical Products Certification)

MILTON ZUANAZZI
Diretor-Presidente
(Director President)

F-400-01C (05.06) Fl. 1 de 2 (Sheet) (of) H.02-2656-0

ANAC

AGÊNCIA NACIONAL DE AVIAÇÃO CIVIL

Folha de Continuação ao
(Continuation Sheet to)

CERTIFICADO DE HOMOLOGAÇÃO SUPLEMENTAR DE TIPO
(Supplemental Type Certificate)

NÚMERO (Number) **2007S04-12**

LIMITAÇÕES E CONDIÇÕES:
(Limitations and Conditions:)

I. The approval of this type design change should not be extended to other rotorcraft of these models on which other previously approved modifications are incorporated unless it is determined by the installer that the relationship between this change and any of those other previously approved modifications, including changes in Type Design, will introduce no adverse effect upon the airworthiness of that rotorcraft.

II. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

III. Operation must be performed in accordance with FAA approved Rotorcraft Flight Manual Supplement (RFMS), document No. 121-012-01, Rev. 0, dated 19 Oct. 2004, or Supplement No. 121-012-02, Rev. 0, dated 2 Feb. 2005, or later approved revisions.

IV. This modification must be inspected and maintained in accordance with section ATA 5 of the Onboard Systems Instructions for Continued Airworthiness, document No. 123-011-01, Rev. 0, dated 11 Aug. 2004, or document No. 123-011-02, Rev. 0, dated 13 Dec. 2004, and Service Manual No. 122-005-00, Rev. 9, dated 3 Jan. 2006, or Service Manual No. 122-015-00, Rev. 4, dated 6 Jul. 2006, or later approved revisions.

V. A copy of this Certificate, Supplement and Maintenance Manual referred on items III and IV above, respectively, shall be maintained as part of the permanent records of the modified rotorcraft.

---------- END ----------

F-400-01C (05.06) Fl. 2 de 2 (Sheet) (of) H.02-2656-0

Brazilian CHST continued

ANAC

AGÊNCIA NACIONAL DE AVIAÇÃO CIVIL - BRASIL

CERTIFICADO DE HOMOLOGAÇÃO SUPLEMENTAR DE TIPO

(Supplemental Type Certificate)

NÚMERO (Number) **2007S04-11**

Este certificado, emitido com base na Lei nº 7565 "Código Brasileiro de Aeronáutica", de 19 de dezembro de 1986,
(This certificate, issued in the basis of the Law No. 7565 "Código Brasileiro de Aeronáutica", dated 19 December 1986,

é conferido ao (à): (is granted to:) Onboard Systems
13915 NW 3rd Court
Vancouver, WA 98685
USA

por ter a modificação ao projeto de tipo do produto abaixo citado, observadas as limitações e condições
(for having the change to the type design of the product mentioned below, with the limitations and conditions therefor as)

especificadas, satisfeito aos requisitos de aeronavegabilidade aplicáveis.
(specified hereon, met the applicable airworthiness requirements.)

Produto Original - Número do Certificado de Tipo: (Original Product – Type Certificate No:) 84 (FN 157) (DGAC France)

Fabricante: (Manufacturer:) Eurocopter France.

Modelo(s): (Model(s):) AS 350 B.

DESCRIÇÃO DA MODIFICAÇÃO AO PROJETO DE TIPO:
(Description of Type Design Change:)

Installation of Onboard Systems Model 200-280-01 or 200-280-02 Cargo Hook Swing Suspension System in accordance with Onboard Systems Owner's Manual No. 120-104-01, Rev. 5, dated 3 Apr. 2007 or Owner's Manual No. 120-104-02, Rev. 9, dated 3 Apr. 2007, or later approved revisions.
This CHST validates in Brazil the STC # SR01164SE, issued by FAA (USA).

LIMITAÇÕES E CONDIÇÕES:
(Limitations and Conditions:)

See continuation sheet for applicable data.

DATAS:
(Dates of:)

Do Requerimento: (Application:) 21 Dec. 2006 **Da emissão**: (Issue:) 13 Apr. 2007 **Da reemissão**: (Reissue:)

CLÁUDIO PASSOS SIMÃO
Gerente Geral, Certificação de Produtos Aeronáuticos
(Manager, Aeronautical Products Certification)

MILTON ZUANAZZI
Diretor-Presidente
(Director President)

F-400-01C (05.06)

Fl. 1 de 2 (Sheet) (of)

H.02-2655-0

Brazilian CHST continued

ANAC

AGÊNCIA NACIONAL DE AVIAÇÃO CIVIL

Folha de Continuação ao
(Continuation Sheet to)

CERTIFICADO DE HOMOLOGAÇÃO SUPLEMENTAR DE TIPO
(Supplemental Type Certificate)

NÚMERO (Number) **2007S04-11**

LIMITAÇÕES E CONDIÇÕES:
(Limitations and Conditions:)

I. The approval of this type design change should not be extended to other rotorcraft of these models on which other previously approved modifications are incorporated unless it is determined by the installer that the relationship between this change and any of those other previously approved modifications, including changes in Type Design, will introduce no adverse effect upon the airworthiness of that rotorcraft.

II. If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

III. Operation must be performed in accordance with FAA approved Rotorcraft Flight Manual Supplement (RFMS), document No. 121-012-01, Rev. 0, dated 19 Oct. 2004 or Supplement No. 121-012-02, Rev. 0, dated 2 Feb. 2005, or later approved revisions.

IV. This modification must be inspected and maintained in accordance with section ATA 5 of the Onboard Systems Instructions for Continued Airworthiness, document No. 123-011-01, Rev. 0, dated 11 Aug. 2004 or document No. 123-011-02, Rev. 0, dated 13 Dec. 2004, and Service Manual No. 122-005-00, Rev. 9, dated 3 Jan. 2006 or Service Manual No. 122-015-00, Rev. 4, dated 6 Jul. 2006, or later approved revisions.

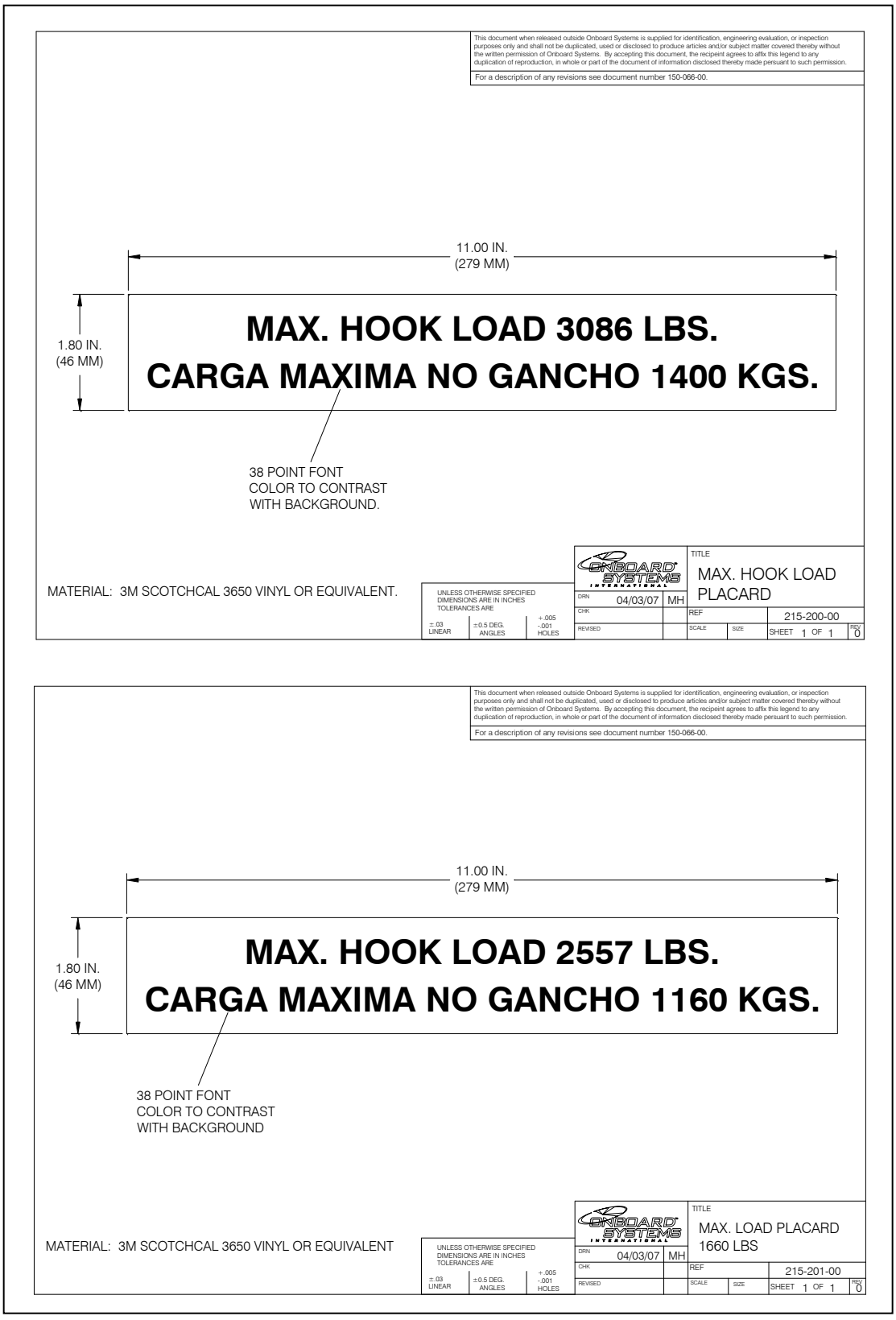
V. A copy of this Certificate, Supplement and Maintenance Manual referred on items III and IV above, respectively, shall be maintained as part of the permanent records of the modified rotorcraft.

---------- END ----------

F-400-01C (05.06) Fl. 2 de 2 (Sheet) (of) H.02-2655-0

Required Placards

Install one of the following per section 2.7.


This document when released outside Onboard Systems is supplied for identification, engineering evaluation, or inspection purposes only and shall not be duplicated, used or disclosed to produce articles and/or subject matter covered thereby without the written permission of Onboard Systems. By accepting this document, the recipeint agrees to affix this legend to any duplication of reproduction, in whole or part of the document of information disclosed thereby made persuant to such permission.
For a description of any revisions see document number 150-066-00.
11.00 IN.
(279 MM)
1.80 IN.
(46 MM)
MAX. HOOK LOAD 3086 LBS.
CARGA MAXIMA NO GANCHO 1400 KGS.
38 POINT FONT
COLOR TO CONTRAST
WITH BACKGROUND.
MATERIAL: 3M SCOTCHCAL 3650 VINYL OR EQUIVALENT.
UNLESS OTHERWISE SPECIFIED
DIMENSIONS ARE IN INCHES
TOLERANCES ARE
±.03 LINEAR
±0.5 DEG. ANGLES
+.005 -.001 HOLES
ONBOARD SYSTEMS INTERNATIONAL
DRN 04/03/07 MH
CHK
REVISED
TITLE
MAX. HOOK LOAD PLACARD
REF 215-200-00
SCALE
SIZE
SHEET 1 OF 1
REV 0
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For a description of any revisions see document number 150-066-00.
11.00 IN.
(279 MM)
1.80 IN.
(46 MM)
MAX. HOOK LOAD 2557 LBS.
CARGA MAXIMA NO GANCHO 1160 KGS.
38 POINT FONT
COLOR TO CONTRAST
WITH BACKGROUND
MATERIAL: 3M SCOTCHCAL 3650 VINYL OR EQUIVALENT
UNLESS OTHERWISE SPECIFIED
DIMENSIONS ARE IN INCHES
TOLERANCES ARE
±.03 LINEAR
±0.5 DEG. ANGLES
+.005 -.001 HOLES
ONBOARD SYSTEMS INTERNATIONAL
DRN 04/03/07 MH
CHK
REVISED
TITLE
MAX. LOAD PLACARD 1660 LBS
REF 215-201-00
SCALE
SIZE
SHEET 1 OF 1
REV 0

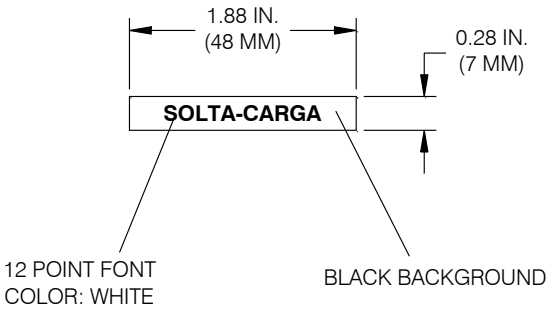
Brazilian CHST

Required Placards continued

Install the following per section 2.7.

This document when released outside Onboard Systems is supplied for identification, engineering evaluation, or inspection purposes only and shall not be duplicated, used or disclosed to produce articles and/or subject matter covered thereby without the written permission of Onboard Systems. By accepting this document, the recipeint agrees to affix this legend to any duplication of reproduction, in whole or part of the document of information disclosed thereby made persuant to such permission.

For a description of any revisions see document number 150-066-00.



MATERIAL: 3M SCOTCHCAL 3650 VINYL OR EQUIVALENT.

UNLESS OTHERWISE SPECIFIED
DIMENSIONS ARE IN INCHES
TOLERANCES ARE

±.03 LINEAR	±0.5 DEG. ANGLES	+.005 -.001 HOLES

ONBOARD SYSTEMS INTERNATIONAL

DRN	04/03/07	MH	
CHK			
REVISED			

TITLE: CARGO RELEASE PLACARD

REF: 215-202-00

SCALE | SIZE | SHEET 1 OF 1 | REV 0

EASA STC

European Aviation Safety Agency

SUPPLEMENTAL TYPE CERTIFICATE

10016937, REV. 2

This Supplemental Type Certificate is issued by EASA, acting in accordance with Regulation (EC) No. 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation and in accordance with Commission Regulation (EC) No. 1702/2003 to

ONBOARD SYSTEMS INT.
13915 NW 3rd COURT
VANCOUVER WA 98685
USA

and certifies that the change in the type design for the product listed below with the limitations and conditions specified meets the applicable Type Certification Basis and environmental protection requirements when operated within the conditions and limitations specified below:

Original Product TC Number:	**EASA.R.008**
TC Holder:	**EUROCOPTER**
Model:	**AS 350 B, AS 350 B1**
Model:	**AS 350 B2, AS 350 B3**
Model:	**AS 350 BA, AS 350 D**
Original STC Number:	**FAA STC SR01164SE**

EASA Certification Basis:
As per EASA TCDS.R.008 Issue 04, 23 November 2009.
The Certification Basis for the original product and the following additional or alternative airworthiness requirements are applicable to this certificate/ approval.
Change under Revision 2 complies with FAR 27 at amdt 27-39.
The requirements for environmental protection and the associated certificated noise and/ or emissions levels of the original product are unchanged and remain applicable to this certificate/ approval.

Description of Design Change:
External Load Swing Suspension System models 200-280-01 and -04 (with Keeperless Cargo Hooks) and model 200-280-02 and -03 (with Hydraulic Cargo Hooks)
Revision 1 - introduced cargo hook kit 200-280-03 with a collective friction knob and a new release lever to accommodate the AS350B3 style collective configuration
Revision 2 - introduced cargo hook kit 200-280-04 (with Keeperless Cargo Hook 528-029-00)

See Continuation Sheet(s)

For the European Aviation Safety Agency,

Date of issue: 21.06.2010

For **Massimo MAZZOLETTI**
Certification Manager
Rotorcraft, Balloons, Airships

Note:
The following numbers are listed on the certificate:
EASA old Project Number: EASA.IM.R.S.01324, REV. 2

SUPPLEMENTAL TYPE CERTIFICATE - 10016937, REV. 2 - ONBOARD SYSTEMS INT.

EASA STC continued

European Aviation Safety Agency

This STC is the validation of FAA STC SR01164SE issued on January 22, 2003 and last amended on April 13, 2010

Associated Technical Documentation:
Definition and installation:
Onboard Systems Master Drawing List No. 155-086-00, Rev. 27, dated October 30, 2009, approved by FAA.
Onboard Systems Owner's Manual:

No. 120-104-01 (for system model 20028001), Rev. 10 dated October 30, 2008, approved by FAA,
or
No. 120-104-02 (for system model 20028002 and -03), Revision 18 dated July 1, 2009, approved by FAA,
or
No. 120-104-03 (for system model 20028004), Revision 0 dated October 12, 2009, approved by FAA.
Inspection and maintenance:
Onboard Systems Instructions for Continued Airworthiness:

No. 123011-01 (for system model 20028001), Rev. 3 dated October 29, 2007 approved by FAA,
or
No. 123011-02 (for system model 20028002 and -03), Rev. 5 dated January 27, 2009 approved by FAA,
or
No. 123011-03 (for system model 20028004), Rev. 0 dated January 27, 2010 approved by FAA,
and

Onboard Systems Service Manual:

No. 122005-00 (for Keeperless Cargo Hook 528-023-01), Rev. 17 dated March 9, 2010 approved by FAA,
or
No. 122-015-00 (for Hydraulic Cargo Hook), Rev. 7 dated January 30, 2009 approved by FAA,
or
No. 122-017-00 (for Keeperless Cargo Hook 528-029-00), Rev. 8 dated March 10, 2010 approved by FAA.
Operation:

Onboard Systems RFMS:

No. 121-012-01 (for system model 20028001), Rev. 2 dated June 12, 2007 approved by FAA,
or
No. 121-012-02 (for system model 200-280-02 and -03) Rev. 3 dated October 25, 2007 approved by FAA,
or
No. 121-012-03 (for system model 200-280-04) Rev. 0 dated March 4, 2010 approved by FAA.

Note:
The following numbers are listed on the certificate:
EASA old Project Number: EASA.IM.R.S.01324, REV. 2

SUPPLEMENTAL TYPE CERTIFICATE - 10016937, REV. 2 - ONBOARD SYSTEMS INT.

EASA STC continued

European Aviation Safety Agency

or later revisions of the above listed documents approved by EASA in accordance with EASA ED Decision 2004/04/CF (or subsequent revisions of this decision)

Limitations:
1. EASA-approved AS350 Flight Manual and <External Load Transport: Cargo Swing> FM supplement are required.
2. This STC is approved only for the product configuration as defined in the approved design data referred to in the paragraph "Description". Compatibility with other aircraft/engine configurations shall be determined by the installer.
3. Installation and operation of the system on AS350B3 with the modification OP-3369 incorporated is not approved.

Conditions:
Prior to installation of this modification it must be determined that the interrelationship between this modification and any other previously installed modification and/ or repair will introduce no adverse effect upon the airworthiness of the product.

This Certificate shall remain valid unless otherwise surrendered or revoked.

- end -

Note:
The following numbers are listed on the certificate:
EASA old Project Number: EASA.IM.R.S.01324, REV. 2

SUPPLEMENTAL TYPE CERTIFICATE - 10016937, REV. 2 - ONBOARD SYSTEMS INT.